



HAL
open science

Impact of Geometric, Operational, and Model Uncertainties on the Non-ideal Flow Through a Supersonic ORC Turbine Cascade

Nassim Razaaly, Giacomo Persico, Pietro Marco Congedo

► **To cite this version:**

Nassim Razaaly, Giacomo Persico, Pietro Marco Congedo. Impact of Geometric, Operational, and Model Uncertainties on the Non-ideal Flow Through a Supersonic ORC Turbine Cascade. Energy, In press, 10.1016/j.energy.2018.11.100 . hal-01982234

HAL Id: hal-01982234

<https://inria.hal.science/hal-01982234v1>

Submitted on 15 Jan 2019

HAL is a multi-disciplinary open access archive for the deposit and dissemination of scientific research documents, whether they are published or not. The documents may come from teaching and research institutions in France or abroad, or from public or private research centers.

L'archive ouverte pluridisciplinaire **HAL**, est destinée au dépôt et à la diffusion de documents scientifiques de niveau recherche, publiés ou non, émanant des établissements d'enseignement et de recherche français ou étrangers, des laboratoires publics ou privés.

Impact of Geometric, Operational, and Model Uncertainties on the Non-ideal Flow Through a Supersonic ORC Turbine Cascade

Nassim RAZAALY^{a,*}, Giacomo PERSICO^b, Pietro Marco CONGEDO^a

^a*DeFI Team (Inria Saclay Ile-de-France, Ecole Polytechnique), CMAP, 1 rue d'Estienne d'Orves, 91120 Palaiseau, France.*

^b*Laboratorio di Fluidodinamica delle Macchine, Dipartimento di Energia, Politecnico di Milano, Via Lambruschini 4, I-20156 Milano, Italy.*

Abstract

Typical energy sources for Organic Rankine Cycle (ORC) power systems feature variable heat load and turbine inlet/outlet thermodynamic conditions. The use of organic compounds with heavy molecular weight introduces uncertainties in the fluid thermodynamic modeling. In addition, the peculiarities of organic fluids typically leads to supersonic turbine configurations featuring supersonic flows and shocks, which grow in relevance in the aforementioned off-design conditions; these features also depends strongly on the local blade shape, which can be influenced by the geometric tolerances of the blade manufacturing. This study presents an Uncertainty Quantification (UQ) analysis on a typical supersonic nozzle cascade for ORC applications, by considering a two-dimensional high-fidelity turbulent Computational Fluid Dynamic (CFD) model. Kriging-based techniques are used in order to take into account at a low computational cost, the combined effect of uncertainties associated to operating conditions, fluid parameters, and geometric tolerances. The geometric variability is described by a finite Karhunen-Loeve expansion representing a non-stationary Gaussian random field, entirely defined by a null mean and its autocorrelation function. Several results are illustrated about the ANOVA decomposition of several quantities of interest for different operating conditions, showing the importance of geometric uncertainties on the turbine performances.

Keywords: Geometric Variability, Uncertainty Quantification, ORC Turbine, SU2, Non-Ideal Compressible Fluid Dynamics

1. Introduction

Organic Fluids are characterized by complex molecules and moderate to large molecular weights, resulting in critical point temperature significantly lower than that of water. By virtue of these features, the implementation of organic fluids in Rankine cycles, referred to as Organic Rankine Cycle (ORC) power systems, has prompted the development of widely distributed, thermal energy conversion devices suitable to exploit heat sources featuring low temperature level such as, for example, geothermal energy, concentrated solar energy, biomass or waste heat from industrial processes. These sources also share a certain degree of variability in the heat load, which might combine with variations in the condenser temperature due to the inherent annual variability of ambient temperature; as a result, the thermodynamic condition at the inlet and, in some cases, at the outlet of the ORC turbine undergoes significant variation, resulting in a variable expansion ratio and aerodynamic loading on each of the cascades of the machine (see [1]). To improve the reliability of this technology, the robustness to variation in input conditions must be taken into account at an early stage of the development process.

On the other hand, just a few highly uncertain thermodynamic data are available for many ORC working fluids, especially the heaviest and most complex ones [2, 3, 4]. This complicates the development of high-fidelity Equations of State describing the thermodynamic behavior of these fluids, and leaves a certain margin for uncertainty in the parameters of the model equations (see [5] for a discussion). Furthermore, the need for a reduction in production costs of ORC systems may push ORC companies towards large-scale manufacturing techniques based on low-quality

*Corresponding author. Tel.: +33524574177. Mail: nassim.razaaly@inria.fr

machining and assembling technologies, which might result in significant geometric tolerances on the final product [6].

On the basis of the aforementioned considerations, three areas of uncertainties can be identified when analyzing ORC turbines. Theoretical analysis of ORC turbines is nowadays performed by resorting to Computational Fluid Dynamics (CFD); in the classical approach, CFD simulations are run considering a pre-defined geometric configurations and by assigning a set of deterministic input data and model parameters. In order to evaluate the effects of geometry and parameter variation through a CFD model, several stochastic approaches have been developed in recent years [7] [8], which treat the problem solution as a vector of random variables depending, via the governing equations, on one or more stochastic input parameters. These may affect the solution via the equations themselves, via the initial and/or boundary conditions and via the geometry of the computational domain. The results of these stochastic simulations is no longer a single deterministic solution, but rather the probability distribution function of the solution or at least its lower-order statistical moments, such as the expectancy and the variance. By virtue of this class of simulations under uncertainty, the mean system performance can be evaluated along with its variability, and this information can be taken into account for system optimization. For instance, one can think of optimizing the mean system performance while minimizing some variability parameter to get a more robust design and avoid overfitting problems. The interested reader may refer, e.g., to [6] and [9] for a discussion of optimization strategies under uncertainties in real-gas scenario.

The present work, for the first time in the open Literature on ORC power systems, investigates the application of some advanced stochastic techniques to the analysis of a typical supersonic turbine cascade for ORC applications by considering a high-fidelity turbulent solver and including a detailed characterization of the geometric uncertainties. In fact, some recent works dealt with ANOVA-based analysis applied to the simulation of non-ideal gas flows, without an accurate characterization of geometric uncertainties: i) using Euler solvers and considering uncertainties on operating conditions and thermodynamics models [10, 11, 9], ii) using turbulent solvers [12].

In particular, in this paper, a high-fidelity two-dimensional CFD model is used, coupled with the $k - \omega$ -SST model to introduce the effects of turbulence and with the Peng-Robinson-Stryjek-Vera Equation of State to treat the non-ideal thermodynamics of the organic vapour under consideration. Multiple sources of uncertainty are taken into account, related to the geometric tolerances of the blade and to the operating conditions, and also considering those of some parameters of the thermodynamic model.

This paper is organized as follows. Section 2 provides a description of the CFD model used in this paper. How uncertainties are modeled is illustrated in Section 3. Then, Section 4 is devoted to the presentation of the UQ methods used for the sensitivity analysis. Section 5 illustrates the main outcomes of this paper and a physical analysis. Finally, some conclusions and perspectives are drawn in Section 6.

2. Computational Flow Model

In this section, the flow solver and the flow model used in this work are recalled. The turbine configuration and the operating conditions of interest are also introduced.

2.1. CFD Solver

The code SU2 [13, 14] has been employed to carry out the numerical analysis presented in this work, in particular the solver version developed for non-ideal Computational Fluid-Dynamics (SU2 NI-CFD) [15]. The reliability of solver predictions was assessed in [16]. The SU2 NI-CFD solver relies on an embedded thermodynamic library which includes several Equation of States (EoS) and advanced turbulence models. In the following, a brief overview of the main features of SU2 are recalled. The three-dimensional Navier-Stokes equations for compressible flows read

$$\frac{\partial \mathbf{u}}{\partial t} + \nabla \cdot \mathbf{f}(\mathbf{u}) = \nabla \cdot \mathbf{d}(\mathbf{u}) \quad (1)$$

The vector of the conserved variables $\mathbf{u} = (\rho, \mathbf{m}, E^t)^T$ includes the mass density $\rho \in \mathbb{R}^+$, the three component vector related to momentum density $\mathbf{m} \in \mathbb{R}^3$, and the specific total energy $E^t \in \mathbb{R}$, $E^t = \rho \left(e + \frac{1}{2} \|\mathbf{v}\|^2 \right)$ (where e is the specific internal energy while $\mathbf{v} = \mathbf{m}/\rho$ is the velocity vector).

The functions $\mathbf{f}(\mathbf{u}) = [\mathbf{m}, (\mathbf{m} \otimes \mathbf{m})/\rho + P\mathbf{I}, \mathbf{m}(E^t + P)/\rho]^T$ and $\mathbf{d}(\mathbf{u}) = [\mathbf{0}, \mathbf{\Pi}, \mathbf{v}^T \cdot \mathbf{\Pi} - \mathbf{q}]^T \in \mathbb{R}^5 \times \mathbb{R}^3$ are, respectively, the advection and pressure fluxes and the viscous and thermal fluxes. In these latter functions, $P = P(\mathbf{u})$ is the pressure, \mathbf{I} is the identity matrix of dimension 3, $\mathbf{\Pi} = \mathbf{\Pi}(\mathbf{v})$ is the viscous stress tensor and \mathbf{q} is the thermal flux.

The system of equations is supplemented by constitutive relations that bound the state of the fluid to the thermodynamic and to the transport quantities. SU2 NI-CFD solves, in fact, the Reynolds-Averaged Navier-Stokes (RANS) equations, in which the effects of turbulence are introduced via the so-called turbulent or eddy viscosity and of the turbulent thermal conductivity, which are evaluated using well-established turbulence models, as reported in [14].

The SU2 suite is equipped with Non-Reflecting Boundary Conditions (NRBC) [17], designed to avoid spurious pressure oscillations due to the reflection of acoustic waves at the domain boundaries and to perform computations on a truncated domain, therefore lowering the computational burden; this is particularly relevant in the present case, which features shock waves generated in the rear suction side of the blades and propagating downstream up to the outflow boundary.

2.2. Flow Model and Turbine Configuration

In the analysis presented hereinafter, the inviscid fluxes are discretized using a second order approximate Riemann solver (ARS) of Roe upwind type [18, 19, 20] along with the slope limiter proposed by van Albada; central differences are used for the viscous terms. The combination of such schemes guarantees high resolution in the numerical solution of the system of equations. The $k - \omega$ Shear Stress Transport (SST) model [21] is used to retrieve the effects of turbulence in RANS simulations. The organic vapour of interest is considered a polytropic non-ideal gas, whose volumetric behavior is described by the Peng-Robinson-Stryjek-Vera Equation of State. Uniform thermo-physical quantities were also assigned, estimated as representative values for the entire transformation by resorting to the RefProp library.

Critical pressure	14.152 bar
Critical temperature	564.1 K
Critical density	256.82 kg.m ⁻³
γ	1.0165
Acentric factor ω	0.529
Gas constant	35.152 J/kg/K
μ	1.1517 $\times 10^{-5}$ Pa.s
k	0.03799 W/(m.K)

Table 1: Gas properties of the siloxane MDM

The configuration of interest is a well known axial-flow supersonic nozzle cascade operating with siloxane MDM, first presented in [1]. The properties of the MDM are reported in Table 1. In this work, blade-to-blade effects are investigated, considering a two-dimensional flow around the blade profiles at the midspan section of the cascade. Total Pressure P_{in}^t , total Temperature T_{in}^t , and axial flow direction are assigned at the inlet, while static pressure P_{out}^s is given at the outlet. A representation of the blade shape, alongside the computational mesh, is given in Figure 1. Following [1], we consider first a nominal (or full-load) operating condition for this nozzle cascade, characterized by an inlet thermodynamic state close to the saturation curve, and a high expansion ratio equal to ≈ 6 . As well known, ORC power systems are requested to operate at part-load for long periods during their technical life, due to changes in the thermal power made available by the heat source and in the condenser temperature; this variation implies a large change in the turbine pressure ratio, resulting in a variation of aerodynamic loading on each cascade. In [1], the implication of part-load operation for this cascades was estimated so to reduce the pressure ratio to ≈ 4 , by an increase of cascade outlet pressure. Both full-load and part-load operating conditions are considered in this study, both of them reported in Table 2.2.

2.3. Grid Generation and Convergence

The computational meshes are generated by resorting to an advancing-front/Delaunay algorithm. To create hybrid grids suitable for viscous simulations, quadrilateral elements are first added over the solid walls, to build a boundary

Condition	P_{in}^t	T_{in}^t	P_{out}^s
Full-load	8.0 bars	543.65 K	1.33 bars
Part-Load	8.0 bars	543.65 K	2.0 bars

Table 2: Operating Conditions

layer mesh. Afterwards, the advancing front algorithm triangulates the remaining portion of the computational domain. Since the cascade configuration features a typical fish-tail shock pattern at the blade trailing edge and the onset of a main shock in the unguided turning region on the rear suction side [1], in these areas the grids were refined so to enhance the quality of the solution. A representative mesh composed by 180 kcells is reported in Figure 1, alongside two enlargements corresponding to the diverging channel and to the trailing edge.

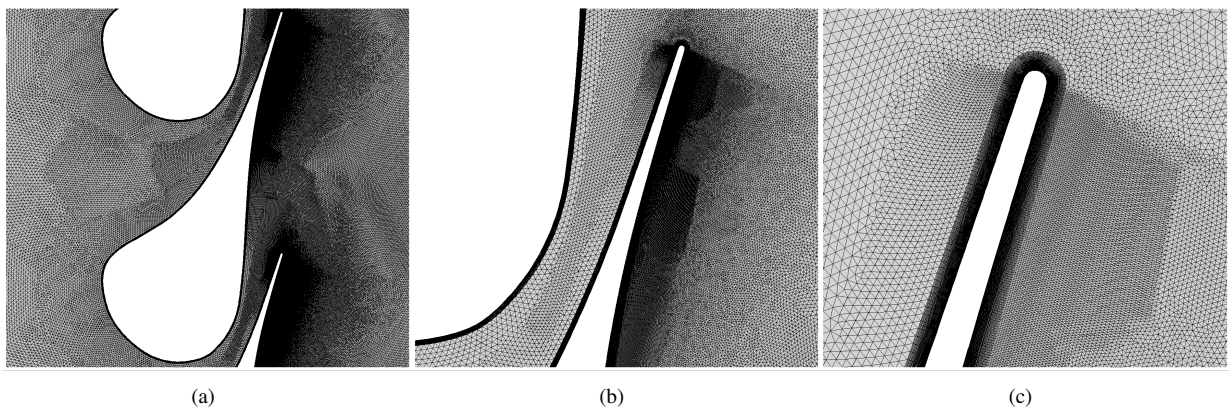


Figure 1: 180k cells mesh.

As no wall functions are used, the near-wall region of the boundary layer is solved entirely by the simulation. To ensure a proper resolution in the near-wall region, the height of the first cell of the boundary layer grid was set to equal to $\approx 2 \cdot 10^{-7}$, corresponding to a y^+ value lower than 1. The y^+ value was evaluated considering the dynamic viscosity value assumed for the whole domain, and given in Table 1; the considered values of stream velocity and density are instead the ones resulting from simulations at the exit of the cascade, the reference length being the blade chord.

A dedicated grid dependence analysis was performed at the full-load condition. Two quantities were used as metric for evaluating the mesh convergence, namely the total pressure loss $Y = \frac{P_{in}^t - P_{out}^t}{P_{in}^t - P_{out}^s}$ (where P_{out}^t denotes the mass-averaged total pressure at the outlet), and the mass flow rate per unit span \dot{m} . Figure 2 reports the relative percentage error on the performance estimators. In this study, grid independence was considered achieved with the mesh composed by 590k cells; on the basis of the quantitative results, the 180k cells mesh (depicted in detail in Figure 1) was considered as the optimal compromise between accuracy and computational cost (error of 2%) and, hence, it was employed to compute the blade performance in the UQ framework. We perform a posteriori assessment that the variability induced by the uncertainty quantification study is mainly predominant with respect to the mesh error.

2.4. SU2 Solver/CPU

The RANS simulations were performed on a cluster equipped with Intel(R) Xeon(R) CPU X5650 at 2.67GHz. An adaptive CFL is selected to be between 10 and 100. Simulations are assumed to be converged when residuals on the density are decreased by six orders of magnitude, or after 15000 iterations. Ill-converged simulations are re-run automatically with a minimal CFL of 1. Simulations run with 6 cores take about 8 hours for meshes composed by 180 kcells.

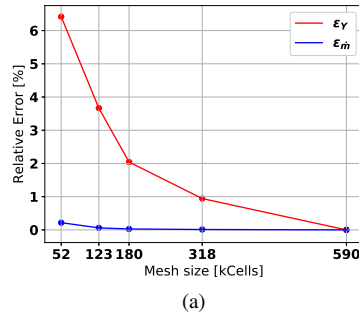


Figure 2: Grid Analysis on the baseline configuration (RANS), for the two QoI m and Y .

2.5. Cascade Flow Field at Full- and Part-Load Conditions

Calculations of the flow in the cascade for the full- and part-load conditions are now reported to properly show the most relevant flow features, in view of the UQ analysis which is the core of this study. Figure 3 reports the distribution of Mach number for the two conditions, and show that high supersonic flows (with peak Mach number values of 2) are established in the rear sides of the cascade and in the cascade-exit flow. As a result of the high Mach number of the free-streams (i.e., out from the boundary layer) on both the suction and pressure sides of the blade, classical fish-tail shock systems are generated at the blade trailing edge. The shock generated on the pressure side of the trailing edge impinges on the suction side of the adjacent blade, and it is reflected; this shock wave grows in strength when propagating downstream, as it merges with the compression occurring on the rear suction side of the blade, and due to the local concave shape of the blade. This leads to the onset of a strong shock wave in the cascade-exit flow field. A similar shock pattern can be observed for the two conditions, even though the main shock is stronger and slightly less inclined (with respect to the axial direction) in the part-load condition. The flow angle is also clearly different in the two conditions, as marked by the higher inclination of the wakes at part-load operation. This feature can be interpreted by considering that, in this latter condition, the converging-diverging nozzle becomes significantly over-expanded with respect to the outlet pressure; as a result, a post-compression takes place that, in cascade configuration, induces a flow turning in tangential direction (so to reduce the 'virtual' cross-section normal to the flow at the exit). The post-compression also explains the higher shock strength at part-load operation. The blade wakes and, especially, the shocks contribute to the loss generation, which is quantified in this study in terms of the aforementioned total pressure loss Y . Y and the mass flow rate per unit span m are the Quantities of Interest (QoI) considered in that work.

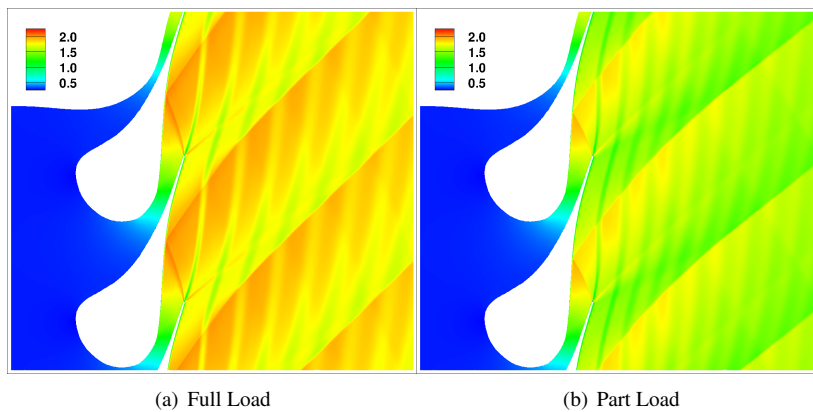


Figure 3: Mach contours: Nominal Conditions

3. Source and Modeling of Uncertainties

In this work, three classes of uncertainties are taken into account. At first, the geometric variability along the blade due to manufacturing tolerances was considered, discussed in subsection 3.1 and modeled through the realization of a random vector of practical dimension ($N_\sigma = 7$ here), stemming from a infinite dimension gaussian process. Moreover, uncertainties in operating conditions as well as in two parameters of the thermodynamic model were also considered and are their quantification is discussed in section 3.2.

3.1. Modeling Geometric Variability

Various approaches have been introduced to model the geometric variations due to blade turbine manufacturing. Based on measurement data, Principal Component Analysis (PCA) can be used to build a probabilistic model of variability from the empirical mean and covariance of surface deviations at different locations on the blade [22, 23, 24]. Following [25, 26], we assume that the geometric variability in manufactured turbine blades can be accurately described as a *non-stationary Gaussian Random Field* $e(s, \omega)$, ω being a coordinate in the sample space Ω , and $(\Omega, \mathcal{F}, \mathbb{P})$ a complete probability space. The arclength $s \in [0, 1]$ parametrizes the location on the blade surface, starting at the trailing edge ($s = 0$), going around the leading edge ($s = \frac{1}{2}$), and continuing back to the trailing edge on the opposite side of the blade ($s = 1$). This approach is somehow similar to the PCA [22], except that in the present work, mean and covariance are provided in closed form rather than estimated from data.

Random Field. The Random field $e(s, \omega)$ represents the error between the manufactured surface and the nominal (perfect) one in the normal direction at the point parametrized by s . It is fully defined by its mean $\bar{e}(s)$ (null here) and autocovariance function $C(s, t)$: it captures the correlation between manufacturing errors at locations s and t along the blade surface, and describes the smoothness and correlation length of the random field. It is written as [26]:

$$C(s, t) = \sigma(s) \sigma(t) \rho(s, t) \quad (2)$$

where $\sigma(s)$ is the standard deviation of the random field at location s , considered constant here: $\sigma(s) = \sigma_0$. σ_0 quantifies the level of manufacturing variability. The non-stationary autocorrelation function ρ is defined by [26]:

$$\rho(s, t) = \exp\left(-\frac{|s - t|^2}{L^2(s, t)}\right) \quad (3)$$

where the *square* correlation length $L^2(s, t)$ is defined by:

$$L^2(s, t) = L(s)L(t) \\ L(s) = L_0 + (L_{LE} - L_0) \exp\left(-\frac{|s - \frac{1}{2}|^2}{\omega^2}\right) \quad (4)$$

The values $L_0 = 0.1$, $\omega = 0.1$ [26] and $L_{LE} = 1.0 \times 10^{-2}$, all normalized by the blade half-arclength were used.

In that study, trailing edge is modelled as a circular arc, thus, the impact of manufacturing variability at the trailing edge is not addressed.

Discretization/Simulation. The Karhunen-Loeve (KL) expansion is used to represent the random field: it is based on a spectral decomposition of the autocovariance functions. The random field can be represented exactly by [27, 28, 29]:

$$e(s, \omega) = \bar{e}(s) + \sum_{i=1}^{\infty} \sqrt{\lambda_i} \Phi_i(s) Z_i(\omega), \quad (5)$$

where the λ_i are arranged in descending order. Z_i are standard uncorrelated normal random variables; $\lambda_i \in \mathbb{R}^+$ and $\Phi_i : [0, 1] \rightarrow \mathbb{R}$ are respectively the eigenvalues and orthonormal eigenfunctions of the autocovariance function C , also referred to as the kernel function in this context, obtained by solving the homogeneous Fredholm integral equation of the second kind:

$$\int_{[0,1]} C(s, t)\Phi_i(t)dt = \lambda_i\Phi_i(s), \forall s \in [0, 1]. \quad (6)$$

This integral equation is solved numerically on the discretized blade surface using the Nyström method [27, 30], with N_g Gauss quadrature points ($N_g = 500$ here) to approximate the integral. Discretized Eigenmodes are consequently evaluated solving a discrete eigenproblem, using for instance a Singular Value Decomposition (SVD). It results in a finite expansion further truncated to give a stochastic dimension of N_{KL} , determined by the decay of the eigenvalues. The normalized cumulative energy from the first p modes is defined by

$$E(p) = \frac{\sum_{i=1}^p \lambda_i}{\sum_{i=1}^{N_g} \lambda_i}. \quad (7)$$

It is used to select the number of modes retained in the KL expansion:

$$N_{KL} = \arg \min\{k \in \mathbb{N}^* \text{ s.t. } E(k) < \epsilon\}, \quad (8)$$

where ϵ is a user defined parameter driving the accuracy of the expansion. $\epsilon = 99\%$ is chosen here.

Figure 4 illustrates the eigenmodes decay and selection. By analyzing the effects of the modes, it is evident that the odd modes alter, with different 'wavenumber', the blade thickness: mode 1 appears as a simple omothetic scaling of the blade, mode 3 increases the thickness in the trailing edge region and makes the blade more slender in the leading edge, mode 5 makes the blade more slender in the central part of the blade and increases the thickness both at leading edge and trailing edge, and so on for mode 7. Conversely, the even modes act mainly as blade rotation: mode 2 appears indeed as a rigid blade rotation centered close to the leading edge, mode 4 and mode 6 are more complex deformations in which the leading edge, the trailing edge and the central part of the blade 'deflect' (by a different amount) with respect to the original blade.

Figure 5 shows samples of geometric perturbation with the proposed method.

Three levels of geometric variability are considered to study its impact on the QOI:

- No geometric variability: $\sigma_0 = 0$.
- Moderate geometric variability: $\sigma_0 = 3 \times 10^{-5}m$.
- Large geometric variability: $\sigma_0 = 6 \times 10^{-5}m$.

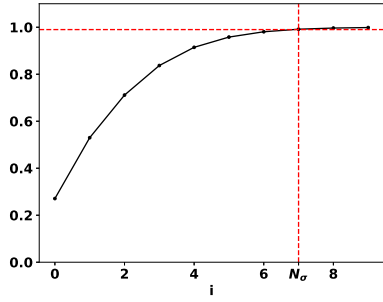
In the following, the unity of σ_0 will be skipped for a sake of brevity.

3.1.1. Mesh Deformation

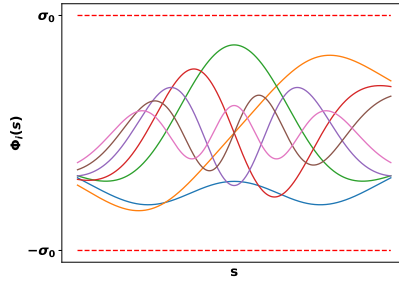
The introduction of geometric uncertainties involves issues related to the computational mesh for the flow simulation. It requires either to modify the initial mesh, or to resort to an automated mesh tool; the former approach was used here.

We rely on the use of a grid deformation in-house tool in order to accurately deform the grid for an assigned boundary displacement. The present approach follows the work of [31], successfully applied in [32], to achieve a highly flexible and robust deformation tool for unstructured meshes, based on the interpolation of boundary nodes displacements to the whole mesh with Radial Basis Function (RBF's). A linear system of equations only involving boundary nodes has to be solved, and no grid connectivity information is needed.

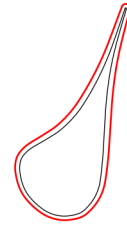
In this subsection, the following notations are. d denotes the dimension space (here $d = 2$), $\mathbf{x} \in \mathbb{R}^d$ a node in the mesh, $\mathbf{x}_{new} \in \mathbb{R}^d$ its new location, n_b the number of boundary nodes, $\{\mathbf{x}_{b_j}\}_{j \in \llbracket 1, n_b \rrbracket}$ the set of nodes at the boundary, p a



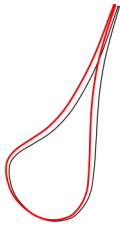
(a) Cumulative Energy $E(p)$ as a function of number of modes (Eq. 7). $\epsilon = 99\% \implies N_\sigma = 7$



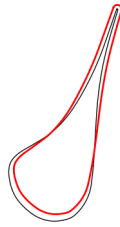
(b) Eigenmodes $\Phi_i(s)$ retained



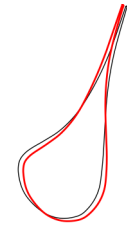
(c) Mode 1



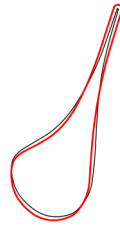
(d) Mode 2



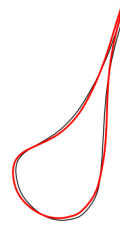
(e) Mode 3



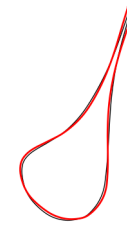
(f) Mode 4



(g) Mode 5



(h) Mode 6



(i) Mode 7

Figure 4: Eigenmodes Illustration, with scale=50, $\sigma_0 = 6 \times 10^{-5}$.

polynomial.

The interpolation function $s : \mathbb{R}^d \rightarrow \mathbb{R}^d$, describing the displacement in the whole domain, can be written as a sum of basis functions:

$$s(\mathbf{x}) = \sum_{j=1}^{n_b} \alpha_j \Phi(\|\mathbf{x} - \mathbf{x}_{b_j}\|) + p(\mathbf{x}), \quad (9)$$

where Φ is a given RBF. Coefficients of the linear polynomial p and α_j are determined by the interpolation

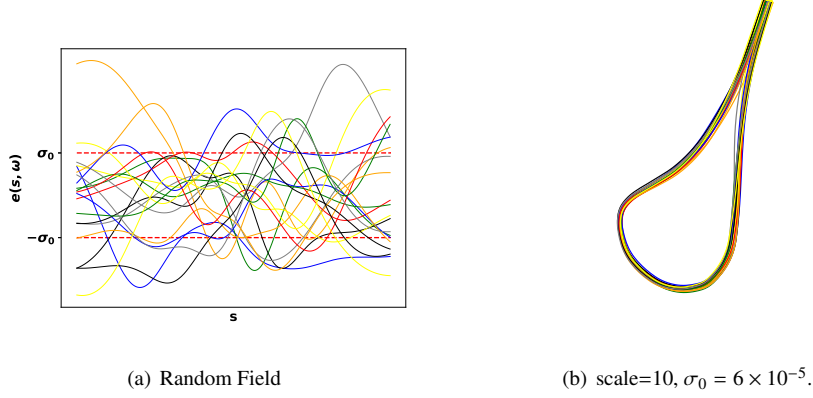


Figure 5: Geometric Variability Modeling: Samples.

conditions:

$$s(\mathbf{x}_{b_j}) = \mathbf{d}_{b_j} \quad (10)$$

$$\sum_{j=1}^{n_b} \alpha_j q(\mathbf{x}_{b_j}) = 0 \quad (11)$$

\mathbf{d}_{b_j} is the imposed displacement of the boundary node \mathbf{x}_{b_j} . Equation 11 has to be satisfied for all polynomials q with a degree less or equal than that of polynomial p . In our case, the displacement \mathbf{d}_{b_j} is nullified for all boundary nodes, except the ones belonging to the blade. Independently for each spatial direction, the coefficients of the polynomial p and α_j are recovered solving a $(n_b + 4) \times (n_b + 4)$ symmetric positive definite linear system, using a Cholesky decomposition for instance.

The new position \mathbf{x}_{new} of a node in the interior domain initially located in \mathbf{x} is then directly derived:

$$\mathbf{x}_{new} = \mathbf{x} + s(\mathbf{x}) \quad (12)$$

Each point is moved individually involving that no mesh-connectivity information is needed. The RBF retained in this work is the so-called *Volume* function, with global support, defined by $\Phi(r) = r$.

3.2. Operating Conditions and Thermodynamic Model Uncertainties

Uncertainties associated to the thermodynamic conditions are considered in this work, namely on P_{in}^t, T_{in}^t and P_{out}^s , for both full-load and part-load turbine operation. The values of the uncertainties are reported in Tables 3 and 4. The uncertainties were estimated considering typical measurement errors in the monitoring devices of power systems; in particular, $\pm 5kPa$ of uncertainty in the pressure measurements and $\pm 0.35K$ of uncertainty in temperature measurements are assumed.

Two coefficients associated to the thermodynamic model of the fluid are also considered uncertain, namely the acentric factor ω and the heat capacity ratio γ of the fluid (which is considered polytropic in this study). A specific quantification of the variability of these parameters not being available, a uniform uncertainty of 5% is assumed for ω and a uniform range ± 0.004 ensuring thermodynamical consistency for γ . This is consistent with respect to other studies performed in literature over the coefficients of the thermodynamic models (see for example [4, 2, 9]).

Tables 3 and 4 summarize the scenarios corresponding respectively to full- and part-load, each considering three different geometric uncertainty levels.

Globally, the random vector is then denoted as $\boldsymbol{\theta} = (P_{in}^t, T_{in}^t, P_{out}^s, \gamma, \omega, \xi_1, \dots, \xi_{N_\sigma})$, or $\boldsymbol{\theta} = (P_{in}^t, T_{in}^t, P_{out}^s, \gamma, \omega)$ if geometric variations are not taken into account.

Geometric Variations	No $\sigma_0 = 0$	Moderate $\sigma_0 = 3 \times 10^{-5}$	Large $\sigma_0 = 6 \times 10^{-5}$
P_{out}^s [bars]	$\mathcal{U}[1.28, 1.38]$	$\mathcal{U}[1.28, 1.38]$	$\mathcal{U}[1.28, 1.38]$
P_{in}^t [bars]	$\mathcal{U}[7.95, 8.05]$	$\mathcal{U}[7.95, 8.05]$	$\mathcal{U}[7.95, 8.05]$
T_{in}^t [K]	$\mathcal{U}[543.3, 544.0]$	$\mathcal{U}[543.3, 544.0]$	$\mathcal{U}[543.3, 544.0]$
γ [-]	$\mathcal{U}[1.012, 1.020]$	$\mathcal{U}[1.012, 1.020]$	$\mathcal{U}[1.012, 1.020]$
ω [-]	$\mathcal{U}[0.518, 0.540]$	$\mathcal{U}[0.518, 0.540]$	$\mathcal{U}[0.518, 0.540]$
ξ	-	$\mathcal{N}(0, \mathbf{I}_7)$	$\mathcal{N}(0, \mathbf{I}_7)$
Dimension θ	5	12	12

Table 3: Full Load: Uncertain Input, for different geometric variability levels

Geometric Variations	No $\sigma_0 = 0$	Moderate $\sigma_0 = 3 \times 10^{-5}$	Large $\sigma_0 = 6 \times 10^{-5}$
P_{out}^s [bars]	$\mathcal{U}[1.95, 2.05]$	$\mathcal{U}[1.95, 2.05]$	$\mathcal{U}[1.95, 2.05]$
P_{in}^t [bars]	$\mathcal{U}[7.95, 8.05]$	$\mathcal{U}[7.95, 8.05]$	$\mathcal{U}[7.95, 8.05]$
T_{in}^t [K]	$\mathcal{U}[543.3, 544.0]$	$\mathcal{U}[543.3, 544.0]$	$\mathcal{U}[543.3, 544.0]$
γ [-]	$\mathcal{U}[1.012, 1.020]$	$\mathcal{U}[1.012, 1.020]$	$\mathcal{U}[1.012, 1.020]$
ω [-]	$\mathcal{U}[0.518, 0.540]$	$\mathcal{U}[0.518, 0.540]$	$\mathcal{U}[0.518, 0.540]$
ξ	-	$\mathcal{N}(0, \mathbf{I}_7)$	$\mathcal{N}(0, \mathbf{I}_7)$
Dimension θ	5	12	12

Table 4: Part Load: Uncertain Input, for different geometric variability levels

4. UQ methods

This section introduces the techniques employed for the quantification of the uncertainties on the quantities of interest.

4.1. UQ Methods for scalar QoI

Let us introduce the generic model $\mathbf{Y}(\theta) = g(\theta)$, where g is a scalar response function representing the QoI associated to a physical model depending of a vector θ . This is a set of d independent random variables, where each θ_i features a Probability Density Function (PDF) f_{θ_i} . In the following, we provide a basic description about the methods for evaluating the statistics, and the Sobol Indices (ANOVA).

Note that in practice, the generic model \mathbf{Y} is replaced by a Kriging-based surrogate model [33, 34] $\tilde{\mathbf{Y}}$. An anisotropic Matern $\frac{5}{2}$ kernel is chosen for the covariance function, along with a null trend. The hyperparameters are tuned maximizing the likelihood. The latter is built based on Latin Hypercube Sampling (LHS) evaluations.

4.1.1. UQ Surrogated-based: Scalar Statistics Evaluation

In this study, the following statistics of $\mathbf{Y}(\theta)$ are considered: mean $\mu_Y = \mathbb{E}_\theta[\mathbf{Y}]$, standard deviation $\sigma_Y = \mathbb{V}_\theta^{\frac{1}{2}}(\mathbf{Y})$ and α -quantile = q_α^Y . We recall below their definition. The expectation of $f(\theta)$, a.k.a. its mean, is defined as:

$$\mathbb{E}_\theta[\mathbf{Y}] = \int \mathbf{Y}(\theta) f_\theta(\theta) d\theta. \quad (13)$$

where f_θ designs the PDF of θ , defined as $\prod_{i=1}^d f_{\theta_i}$ here, due to the independence assumption made above. The so-called variance of $f(\theta)$ is defined as:

$$\mathbb{V}_\theta[\mathbf{Y}] = \mathbb{E}_\theta[\mathbf{Y}^2] - \mathbb{E}_\theta[\mathbf{Y}]^2. \quad (14)$$

For $\alpha \in [0, 1]$, the α -quantile, is defined by:

$$q_\alpha^Y = \inf\{q \in \mathbb{R} \text{ s.t. } \mathbb{P}_\theta(\mathbf{Y} < q) \leq \alpha\}. \quad (15)$$

Otherwise speaking, it measures the probability that \mathbf{Y} , due to the variability of $\boldsymbol{\theta}$, is lower than q_α^Y is α , *i.e.* $\mathbb{P}_\theta(\mathbf{Y} < q_\alpha^Y) = \alpha$.

The Coefficient Of Variation (CoV) defined by $\text{CoV} = \frac{\sigma_Y}{\mu_Y}$ is also used to quantify the normalized variability of \mathbf{Y} .

Corresponding empirical estimators based on Monte-Carlo technique are computed with the following algorithm:

- Build a DOE (LHS) and Compute a CFD computation for each sample, yielding: $\{\boldsymbol{\theta}_j, \mathbf{Y}(\boldsymbol{\theta}_j)\}_{j \in [1, N_{LHS}]}$
- Build a surrogate from the DOE: $\tilde{\mathbf{Y}}(\boldsymbol{\theta})$
- Sample a large MC set: $\{\boldsymbol{\theta}_k\}_{k \in [1, N_{MC}]}$
- Compute MC empirical statistics $\hat{\mu}_{\tilde{\mathbf{Y}}}$, $\hat{\sigma}_{\tilde{\mathbf{Y}}}^2$ and $\hat{q}_\alpha^{\tilde{\mathbf{Y}}}$ based on the surrogate $\tilde{\mathbf{Y}}$ using Equations 16, considering the set $\{\tilde{\mathbf{Y}}(\boldsymbol{\theta}_k)\}_k$ sorted:

$$\begin{aligned} \hat{\mu}_{\tilde{\mathbf{Y}}} &= \frac{1}{N_{MC}} \sum_{k=1}^{N_{MC}} \tilde{\mathbf{Y}}(\boldsymbol{\theta}_k) \\ \hat{\sigma}_{\tilde{\mathbf{Y}}}^2 &= \frac{1}{N_{MC}} \sum_{k=1}^{N_{MC}} (\tilde{\mathbf{Y}}(\boldsymbol{\theta}_k) - \hat{\mu}_{\tilde{\mathbf{Y}}})^2 \\ \hat{q}_\alpha^{\tilde{\mathbf{Y}}} &= \tilde{\mathbf{Y}}(\boldsymbol{\theta}_{[\alpha N_{MC}]}) \end{aligned} \quad (16)$$

4.1.2. ANOVA: Sobol Indices

Variance-cased sensitivity indices are common tools in the analysis of complex physical phenomena. More precisely, we are interested in the total Sensitivity Index (SI) [35, 36].

The variance decomposition of the response can be written as follows:

$$\mathbb{V}(\mathbf{Y}) = \sum_{u \subseteq U} \sigma_u^2(\boldsymbol{\theta}_u), \quad (17)$$

where $U = (1, 2, \dots, d)$ is the set of random variables indexes and σ_u^2 is the variance introduced by interactions of random variables $\boldsymbol{\theta}_u \subseteq \boldsymbol{\theta}_U$. The associated sensitivity measure of $\boldsymbol{\theta}_u$, named Sobol SI, is written as the correlation ratio:

$$S_u = \frac{\mathbb{V}(\mathbb{E}[\mathbf{Y}|\boldsymbol{\theta}_u])}{\mathbb{V}(\mathbf{Y})} = \frac{\sum_{v \leq u} \sigma_v^2}{\mathbb{V}(\mathbf{Y})}. \quad (18)$$

For a single variable θ_i , the first order Sobol SI S_i is given by:

$$S_i = \frac{\mathbb{V}(\mathbb{E}[\mathbf{Y}|\theta_i])}{\mathbb{V}(\mathbf{Y})}. \quad (19)$$

It quantifies the contribution to $\mathbb{V}(\mathbf{Y})$ of the main effect of θ_i varying alone, but averaged over variations in other input parameters. Note that $\sum_i^d S_i = 1$.

The total order Sobol SI S_{T_i} is defined by:

$$S_i^T = \frac{\mathbb{E}[\mathbb{V}(\mathbf{Y}|\theta_{(i)})]}{\mathbb{V}(\mathbf{Y})} = 1 - \frac{\mathbb{V}(\mathbb{E}[\mathbf{Y}|\theta_{(i)}])}{\mathbb{V}(\mathbf{Y})}, \quad (20)$$

where the $\theta_{(i)}$ notation indicates the set of all variables except θ_i . It measures the contribution to $\mathbb{V}(\mathbf{Y})$ of θ_i caused by its interactions with all other input variables, with $\sum_i^d S_i^T \geq 1$.

The Sobol SI are estimated using a Monte-Carlo algorithm [36].

4.2. UQ methods for High Dimensional QoI

We are interested in evaluating efficiently statistics (mean and variance) where the quantity of interest is a vector, possibly with a large dimension (say 10^5), for example to determine the distribution of statistical quantities over the computational domain. As a consequence, the method presented in Subsubsection 4.1 is not tractable, since it would require to build a large number of surrogate models.

Let us consider the generic random variable $\mathbf{G}(\mathbf{X})$ where $\mathbf{G} : \mathbb{R}^d \rightarrow \mathbb{R}^n$ is a multidimensional function depending on a random vector \mathbf{X} characterized by its PDF $f_{\mathbf{X}}$. In the following, we describe a stochastic method combining a PCA (Principal Component Analysis) decomposition to reduce the dimension space n , surrogate modeling and MC sampling. Below, a summary of the method:

1. Set an *initial DOE*: Data generation using LHS
2. *PCA*: compute the eigenmodes, and select the most energetic ones.
3. *Build a surrogate* of each scalar coefficient in the reduced basis.
4. *MC sampling* on the scalar coefficients.
5. Recover the *statistics of G*

1) *Initial DOE*. N_{LHS} samples using LHS are generated, then stored in the mean subtracted data matrix $\mathcal{Y} = [\mathbf{G}(\mathbf{x}_i)^{(j)} - \boldsymbol{\mu}_G^{(j)}]_{ij} \in \mathbb{R}^{N_{LHS} \times n}$. $\mathbf{G}(\mathbf{x}_i) \in \mathbb{R}^n$ denotes the i -th sample. $\boldsymbol{\mu}_G \in \mathbb{R}^n$ denotes the data empirical mean, defined as $\boldsymbol{\mu}_G^{(j)} = \frac{1}{N_{LHS}} \sum_{k=1}^{N_{LHS}} \mathbf{G}(\mathbf{x}_k)^{(j)}$. Mean subtraction is an integral part of the solution towards finding a principal component basis minimizing the data approximation mean square error [37].

2) *PCA and modes selection*. The semi-definite positive empirical covariance matrix is built:

$$\mathbf{C} = \frac{\mathcal{Y}^T \mathcal{Y}}{N_{LHS} - 1} \quad (21)$$

$\mathbf{C} \in \mathbb{R}^{n \times n}$ is then decomposed into an orthogonal basis of eigenvectors $\mathbf{V} = (\mathbf{v}_1, \dots, \mathbf{v}_n)$, involving $\mathbf{v}_i^T \mathbf{v}_j = \delta_{ij}$, as

$$\mathbf{C} = \mathbf{V} \boldsymbol{\Lambda} \mathbf{V}^T \quad (22)$$

$\boldsymbol{\Lambda} = \text{diag}(\lambda_1, \dots, \lambda_n)$ is a diagonal matrix where the positive eigenvalues are sorted in decreasing order. Similarly to Equation 8, the l most energetic modes are retained, with $\epsilon = 99.99\%$.

Note that in the cases that we consider, $N_{LHS} < n$, so the covariance matrix \mathbf{C} has a rank N_{LHS} at maximum so $l \leq N_{LHS}$. The l eigenvectors $\{\mathbf{v}_k\}_{k \in \llbracket 1, l \rrbracket}$ with $\mathbf{v}_k \in \mathbb{R}^n$ are set as basis vectors, the vector solution $\mathbf{G}(\mathbf{x})$ at an unknown location \mathbf{x} is sought in the form

$$\mathbf{G}(\mathbf{x}) - \boldsymbol{\mu}_G = \sum_{i=1}^l h_i(\mathbf{x}) \mathbf{v}_i \quad (23)$$

where $h_i(\mathbf{x})$ are the scalar coefficients in the reduced basis, which need to be approximated by a surrogate.

3) *Build a surrogate.* For each $i \in \llbracket 1, l \rrbracket$, the DOE $\{\mathbf{x}_k, \mathbf{v}_i^T(\mathbf{G}(\mathbf{x}_k) - \boldsymbol{\mu}_G)\}_{k \in \llbracket 1, N_{LHS} \rrbracket}$ is used to build the surrogate \tilde{h}_i for h_i . A surrogate for \mathbf{G} consequently reads

$$\tilde{\mathbf{G}}(\mathbf{x}) = \sum_{i=1}^l \tilde{h}_i(\mathbf{x}) \mathbf{v}_i + \boldsymbol{\mu}_G \quad (24)$$

4-5) *MC sampling and statistics of \mathbf{G} .* The decomposition in the reduced basis (Equation 23) is used to approximate the random vector. From the linearity of the operator \mathbb{E}_X , we obtain:

$$\mathbb{E}_X[\mathbf{G}(\mathbf{X})] = \boldsymbol{\mu}_G + \sum_{i=1}^l \mu_{h_i} \mathbf{v}_i \quad (25)$$

where $\mu_{h_i} = \mathbb{E}_X[h_i(\mathbf{X})]$ denotes the mean of the coefficient h_i . Please note that this quantities differs from the mean $\boldsymbol{\mu}_G$ only for the inherent error of using a limited number of modes, hence it is of the order of $1 - \epsilon$.

The variance of $\mathbf{G}(\mathbf{X})$ is derived by component $j \in \llbracket 1, n \rrbracket$, since they are uncorrelated w.r.t. the random vector \mathbf{X} :

$$\begin{aligned} \mathbb{V}_X[\mathbf{G}(\mathbf{X})^{(j)}] &= \mathbb{V}_X\left[\sum_{i=1}^l h_i(\mathbf{X}) \mathbf{v}_i^{(j)}\right] \\ &= \sum_{i=1}^l \mathbf{v}_i^{(j)} \mathbf{v}_i^{(j)} \sigma_{h_i}^2 + 2 \sum_{i=1}^l \sum_{k=i+1}^l \mathbf{v}_i^{(j)} \mathbf{v}_k^{(j)} \rho_{ik} \end{aligned} \quad (26)$$

where $\sigma_{h_i}^2 = \mathbb{V}_X[h_i(\mathbf{X})]$ and $\rho_{ik} = \mathbb{E}_X[h_i(\mathbf{X})h_k(\mathbf{X})] - \mu_{h_i}\mu_{h_k}$ denote respectively the variance of h_i and the covariance between h_i and h_k . This derivation (Equation 26) takes into account that the coefficients h_i can be correlated.

μ_{h_i} , $\sigma_{h_i}^2$, ρ_{ik} are evaluated numerically by means of MC on the metamodels \tilde{h}_i so the mean (resp. variance) of $\mathbf{G}(\mathbf{X})$ are easily reconstructed using Equation 25 (resp. Equation 26).

Computational aspects. Note that in practice, the eigen-decomposition (Λ , \mathbf{V}) is not obtained by evaluating the covariance matrix C and then perform the diagonalization as suggested by Equation 22. Indeed, $C \in \mathbb{R}^{n \times n}$ can be very large and ill-conditioned since $\text{cond}(C) = \text{cond}(\mathcal{Y})^2$.

A Singular Value Decomposition (SVD) on the mean subtracted data matrix \mathcal{Y} is preferred, yielding

$$\mathcal{Y} = \mathbf{U} \boldsymbol{\Sigma} \mathbf{V}^T \quad (27)$$

$\mathbf{U} \in \mathbb{R}^{N_{LHS} \times N_{LHS}}$ is an orthonormal matrix with the property $\mathbf{U}^T \mathbf{U} = \mathbf{I}_{N_{LHS}}$. $\boldsymbol{\Sigma} \in \mathbb{R}^{N_{LHS} \times n}$ is a *diagonal matrix* of eigenvalues $\{\sigma_k\}_k$. Equation 27 is justified by

$$\begin{aligned} C &= \frac{(\mathbf{U} \boldsymbol{\Sigma} \mathbf{V}^T)^T \mathbf{U} \boldsymbol{\Sigma} \mathbf{V}^T}{N_{LHS} - 1} \\ &= \mathbf{V} \frac{\boldsymbol{\Sigma}^T \boldsymbol{\Sigma}}{N_{LHS} - 1} \mathbf{V}^T \end{aligned} \quad (28)$$

Equation 28 also shows the underlying relation between eigenvalues of matrices C and \mathcal{Y} :

$$\lambda_i = \frac{\sigma_i^2}{N_{LHS} - 1} \quad (29)$$

4.3. Stochastic Convergence

An analysis of the convergence of statistics is performed considering full load and $\sigma_0 = 6 \times 10^{-5}$ in order to justify the number of samples necessary to conduct the UQ analysis based on the tools presented above. Three sets of LHS DOE of respective size 100, 200 and 400 are considered. Those sizes are chosen on a basis of similar studies conducted with a inviscid model, not presented here for a sake of brevity.

Based on the three LHS sets, Figure 6 plots the PDFs of \dot{m} and Y , while Figure 7 summarizes their total Sobol SI of input parameters. PDF's and total Sobol SI are in very good agreement for the different LHS sets; however, a slight error in the PDF of Y is observed when using 100 samples, hence all the UQ studies were conducted based on a LHS set of size $N_{LHS} = 200$.

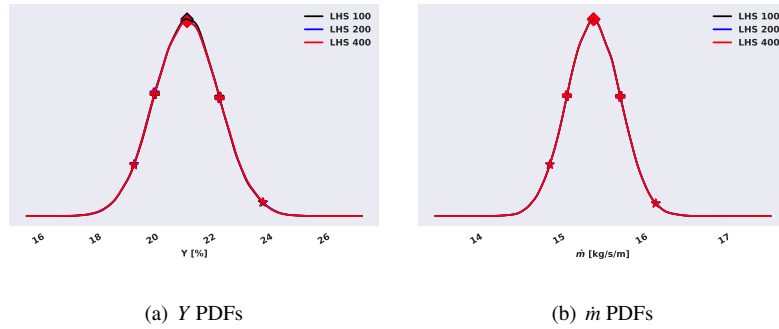


Figure 6: PDF Convergence analysis, based on LHS sets of size 100, 200 and 400. Stars indicate 5% and 95% quantiles, 'plus' indicate $\mu \pm \sigma$ and diamonds indicate μ .

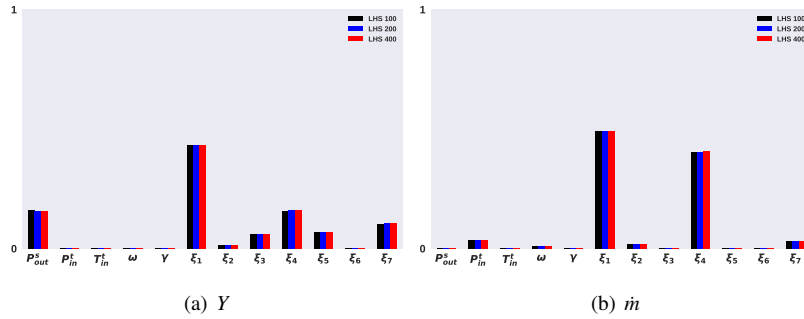


Figure 7: Total Sobol SI Convergence analysis, based on LHS sets of size 100, 200 and 400.

5. Results

This section presents a comprehensive uncertainty quantification on the basis of the stochastic framework formulated above. Based on a LHS set of size $N_{LHS} = 200$ for each case, a comprehensive UQ analysis is performed. In Subsection 5.1, a sensitivity analysis (ANOVA) is carried out. In Subsection 5.2, marginal and joint PDF's of the QoI are studied, their statistics in Subsection 5.3. In Subsection 5.4, the impact of the uncertain input on the flow field is studied.

5.1. Sensitivity Analysis

A Sensitivity analysis was performed for the QOI Y and \dot{m} , and is presented in Figures 8 and 9 respectively, for both full- and part-load. First considering the influence of thermodynamic conditions in absence of geometric variability, the uncertainty in the outlet static pressure has the highest impact on the loss coefficient, especially in full-load condition; this is because the uncertainty in the outlet static pressure is relatively much higher than that on the inlet total pressure, and hence it dominates on the variability of pressure ratio; the variability of pressure ratio, in turns, induces a variation of shock strength and, hence, aerodynamic losses. In part-load condition, the inlet total pressure grows in relevance on Y ; as a matter of fact, for this off-design condition the difference between the inlet and the outlet pressures reduces, while the uncertainty remains the same, so both the uncertainties contribute in a measurable way to the variability of pressure ratio. The outlet static pressure does not have any impact on the flow rate, as expected since the cascade is choked in both full- and part-load conditions. Conversely, the uncertainty in inlet total pressure dominates the variability of flow rate in both the conditions, as the inlet total pressure determines the value of fluid density; this latter, in turn, is only marginally influenced by the inlet total temperature in the conditions of interest.

The uncertainties in the two parameters of the thermodynamic model have a minimal impact on the variability of Y . Conversely, the ratio of specific heat capacities has a relevant impact on the flow rate, as it influences the relationship

between the mass flow rate and the expansion ratio in choked-flow conditions. The acentric factor does not have a measurable effects on both the QoIs. The relevance of operational and model uncertainties is drastically reduced when manufacturing tolerances are plugged into the flow model, even though the mutual hierarchy of the formers remains the same, and the considerations reported above still apply. If geometric variability is large, manufacturing tolerances dominate the variability of the QoIs. In particular, it is observed that mode 1 and mode 4 are responsible for most of the variability of flow rate, and also have the highest impact on the variability of loss coefficient. An analysis of the geometric modes reveals that Mode 1 induces a net variation of geometric throat and also mode 4 has a large impact on the throat, as it is associated to a rotation which alters significantly both the leading edge and the rear side of the blade, which define the geometric throat of the cascade. These considerations justify the relevance of mode 1 and 4 on the flow rate; however, in a supersonic configuration a variation of the geometric throat without a corresponding change in expansion ratio makes the nozzle to operate in off-design conditions, leading to the onset of stronger shocks and, hence, modifying the cascade performance. This may explain why mode 1 and mode 4 also dominate the variability of Y . This latter, however, is influenced by all the modes; as a matter of fact, the development of boundary layer and the onset of the leading shock depend on the detailed shape of the blade, both in the diverging part of the bladed duct and in the region of un-guided turning, and all the modes alter the blade in the rear part. As a further consideration, it is well known in cascade aerodynamics that trailing edge thickness has a relevant impact on the wake width and, hence, on the viscous loss; this contributes to explain the relative relevance of all the odd modes on Y , as all of them alter the trailing edge thickness; this is particularly visible in full-load condition, for that viscous effects contribute more than shocks to the aerodynamic loss, as at full-load the shocks have lower strength.

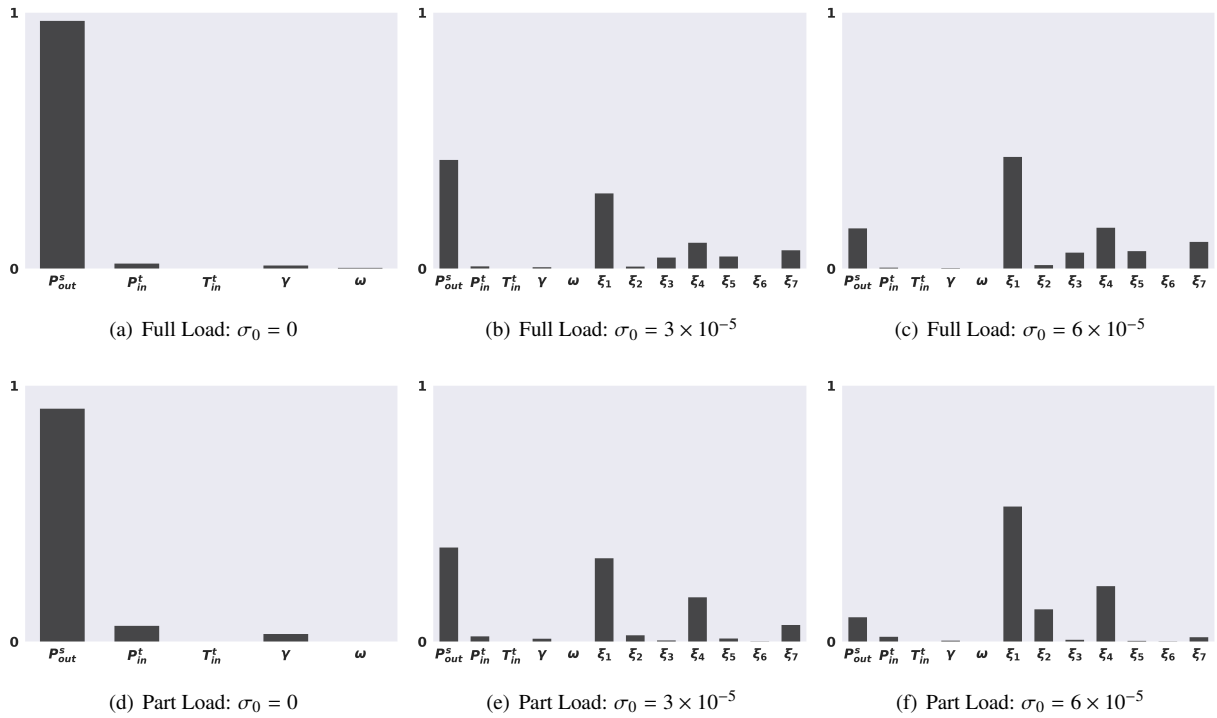


Figure 8: Sensitivity Analysis Y : Total Sobol SI

5.2. PDF Comparisons

The PDFs of Y and \dot{m} are provided in Figure 10 for the three different geometric variations, in both full-load and part-load cases. In an alternative two-dimensional representation, the joint PDFs are also provided in Figure 11.

In absence of geometric variability, the PDFs of Y have a complex non-gaussian shape with different mean values and support; in particular, the support is narrow enough to make the full-load conditions always outperforming the

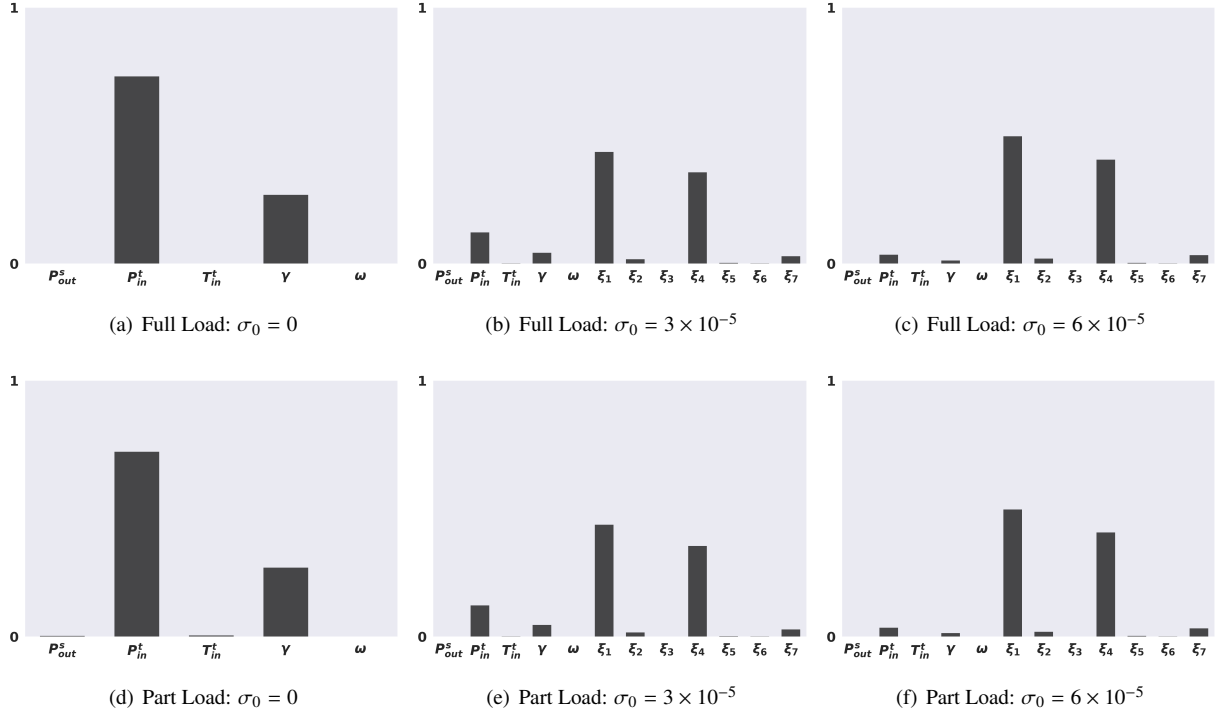


Figure 9: Sensitivity Analysis m : Total Sobol SI.

part-load one. Conversely, the PDFs of m are of gaussian type with very narrow support; this result had been already observed and discussed in detail in [38], and is motivated by the combination of inlet pressure and temperature uncertainties. As the cascade is choked in the two conditions, and the uncertainties in the inlet conditions are the same, the PDFs appear identical.

The influence of geometric tolerances is, again, very significant from both the qualitative and quantitative point of view. At first, the introduction of geometric variability makes the PDFs of Y to become of gaussian type. This is probably because the geometric variability itself is modeled as gaussian process; however, this indicate the dominant effect of manufacturing tolerances, when they are plugged into the uncertainty framework. On the quantitative ground, the geometric variability clearly enlarges the support of the PDFs in all the conditions and for both the QoIs. In particular, the variability in loss coefficient increases from about 2% in absence of tolerances to about 4-6% for the largest tolerance. Despite this, the mean of the PDFs undergoes a slight change (only visible for part-load condition) which does not alter the ranking between full-load and part-load conditions in terms of mean performances; however, when considering the actual realizations of the flow process in presence of the largest geometric variability, a significant overlapping between the PDFs in the two conditions exists (between 21.5% and 24.5%) making the two operating conditions equivalent from the performance perspective for a relatively large percentage of realizations.

The impact of manufacturing tolerances on the flow rate is identical for the two conditions and simply enlarges by a factor ≈ 5 the support of the PDF.

5.3. Statistics

On the basis of the PDFs reported in the previous section, the most relevant statistics (mean, variance, 5-quantile, and 95-quantile) were computed and are reported in Table 5. They show, in more quantitative terms, the features already observed when commenting the PDFs in terms of mean, variance, and support (which is roughly given by the difference between the two quantiles). The interference between the PDFs of Y at large geometric variability is confirmed by the 5-quantiles and 95-quantiles, thus confirming that it is not just a matter of tail probabilities. The analysis also confirms that the quantitative equivalence between the flow rate in the full- and part-load conditions

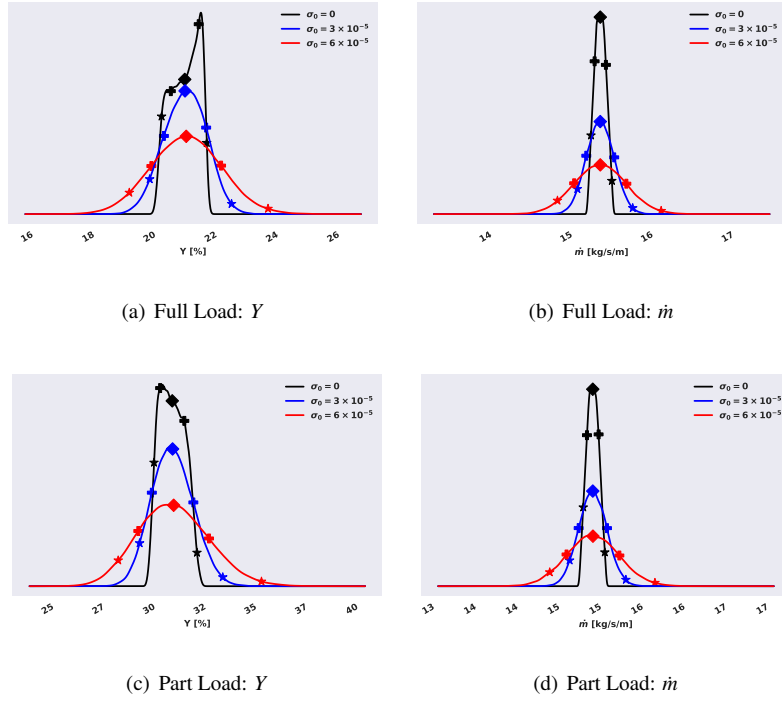


Figure 10: PDF of Y and \dot{m} at full and partial load: comparison between no/moderate/high geometric variations.

does not hold only for the mean properties but it is valid for the whole stochastic framework. This result, which is physically consistent, might be considered as a further proof of the validity of the proposed statistic methodology based on surrogates.

		$\sigma_0 = 0$	$\sigma_0 = 3 \times 10^{-5}$	$\sigma_0 = 6 \times 10^{-5}$
Full Load (μ, σ, q_5, q_{95})	Y [%]	(20.91, 0.45, 20.15, 21.54)	(20.91, 0.69, 19.75, 22.01)	(20.94, 1.14, 19.09, 22.83)
	\dot{m} [kg/s/m]	(15.33, 0.07, 15.22, 15.44)	(15.33, 0.17, 15.04, 15.61)	(15.33, 0.32, 14.80, 15.86)
Part Load (μ, σ, q_5, q_{95})	Y [%]	(30.77, 0.60, 29.87, 31.75)	(30.79, 1.02, 29.16, 32.53)	(30.84, 1.74, 28.10, 33.83)
	\dot{m} [kg/s/m]	(15.38, 0.07, 15.26, 15.49)	(15.38, 0.17, 15.10, 15.66)	(15.38, 0.32, 14.86, 15.90)

Table 5: Scalar Statistics (mean, standard deviation, 5% and 95% quantiles) of the QoI Y and \dot{m} , for the three geometric noise levels.

5.4. Physical Analysis

In this section, QoI of high dimension variability is conducted using the UQ tools described in Subsection 4.2 to perform a physical analysis of the flow. The distributions of Mach number over the entire flow domain and its variability are now discussed, by resorting to the mean and the Coefficient of Variance.

The QoI here shown refer to Mach number at *node location* of the initial mesh, the latter allowed to be deformed conserving the initial nodes connectivity. Prescribing the QoI to assigned spatial location would require a cumbersome interpolation/extrapolation, adding numerical inaccuracies of the QoI, in the vicinity of the blade wall, due geometric blade perturbations.

The distributions of mean and CoV of the Mach number on the computational domain is shown Figures 12 and 13, corresponding respectively to the full- and part-load conditions. The mean Mach number distributions are actually very similar each other and to the deterministic ones. This is probably because the uncertainties are relatively small and are symmetric with respect to the deterministic values; moreover, the manufacturing tolerances are also assumed

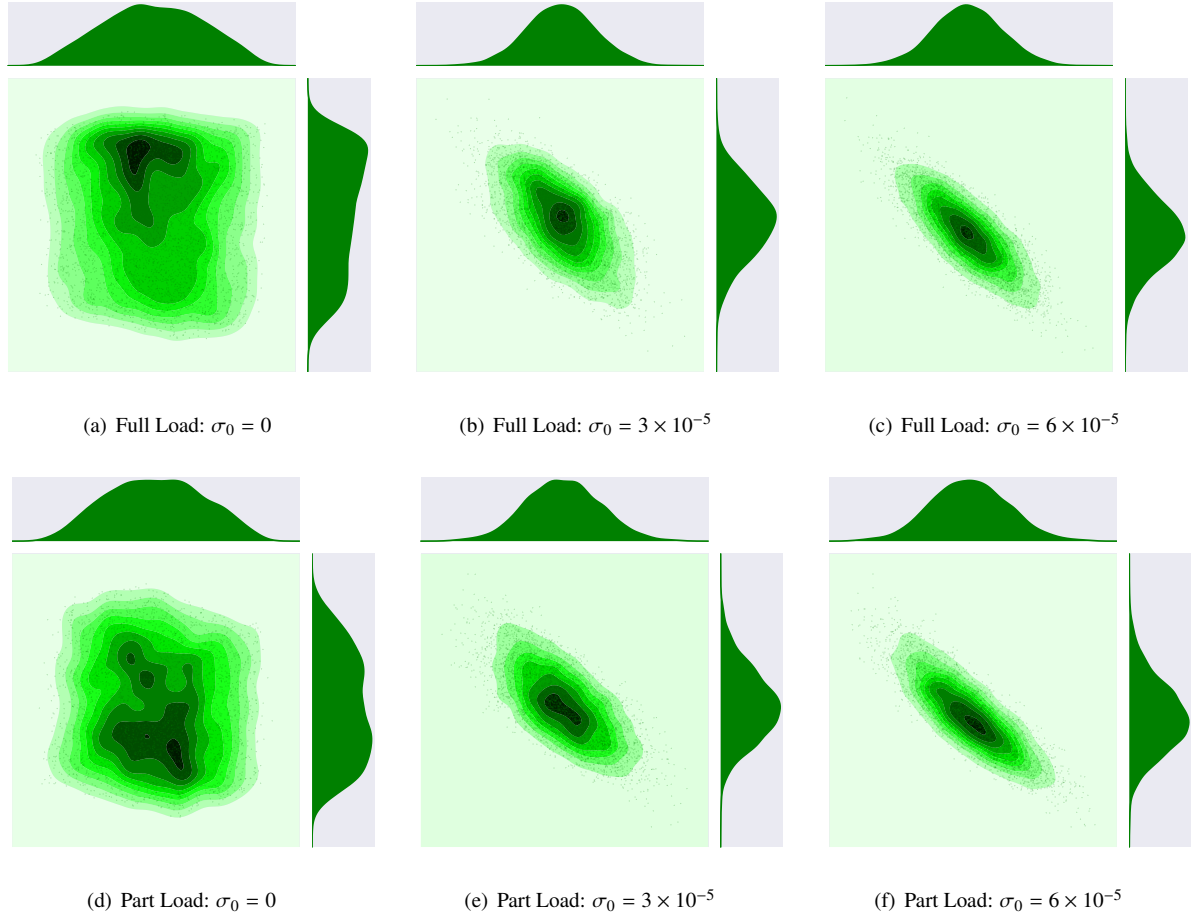


Figure 11: Joint PDF (\dot{m}, Y)

to be gaussian processes whose mean correspond to the original blade shape. However, the same uncertainties produce significant perturbations, that grow significantly when activating the geometric variability. CoV values slightly exceeding 2% are found, with the highest values concentrated in the wakes and in the shock fronts.

In case of no geometric variability, almost null CoV is found in the bladed channel up to the shock on rear suction side of the blade; after that, high CoV is observed in the whole rear suction side up to the trailing edge, where local peaks of CoV occur in the wake and in the fishtail shock system. Moving downstream, the wake width increases due to turbulent and viscous diffusion, resulting in a cascade of wide regions of moderate CoV; in correspondence to the main shock front, the CoV remains locally high and increases moving downstream, as the leading shock grows in strength. As already commented, the main shock grows in magnitude moving downstream as the compression waves generated in the rear suction side, and induced by the local concave shape of the blade, coalesce in a shock front at a certain distance from the blade, enforcing the weak reflected shock coming from the trailing edge of the adjacent blade. The variability in operating conditions (in outlet pressure, in particular) and in the specific heat capacity ratio influence the shock strength justifies the local high values of CoV.

When geometric variability is plugged into the uncertainty framework, the character of the CoV distribution does not change but some evident quantitative effects appear; similar features take place for the two operating conditions. At first, progressively higher CoV values are found upstream of the blade when the manufacturing tolerance increases; the CoV reduces in the accelerating region of the bladed nozzle, and especially in the supersonic divergent, where the mean Mach number grows significantly. This suggests that manufacturing tolerances induce a nearly uniform variance in this region, so that the CoV distribution is dominated by that of the mean. On the rear suction side of

the blade and in the cascade-exit flow field high CoV values are observed in the wakes and in the shock regions, as already observed in absence of geometric variability; however, wider areas of high CoV appear, and, at a certain distance from the blade, moderate-to-high CoV cover the entire flow field. The larger extension of high CoV regions downstream of the cascade can be explained by considering that the modes alter significantly the blade shape in the rear part. In particular, the compression wave generated on the concave part of the rear suction side is perturbed by the local variability in the blade shape, altering both the shock strength and the shock inclination; this contributes to explain the wider high CoV region observed across the shock in presence of manufacturing tolerances. Beside that, the wake appears even more sensitive than the shock to geometric uncertainty; this is probably motivated by the fact that the geometric variability, and in particular the odd modes, alter the trailing edge thickness, which has a direct impact on the wake width.

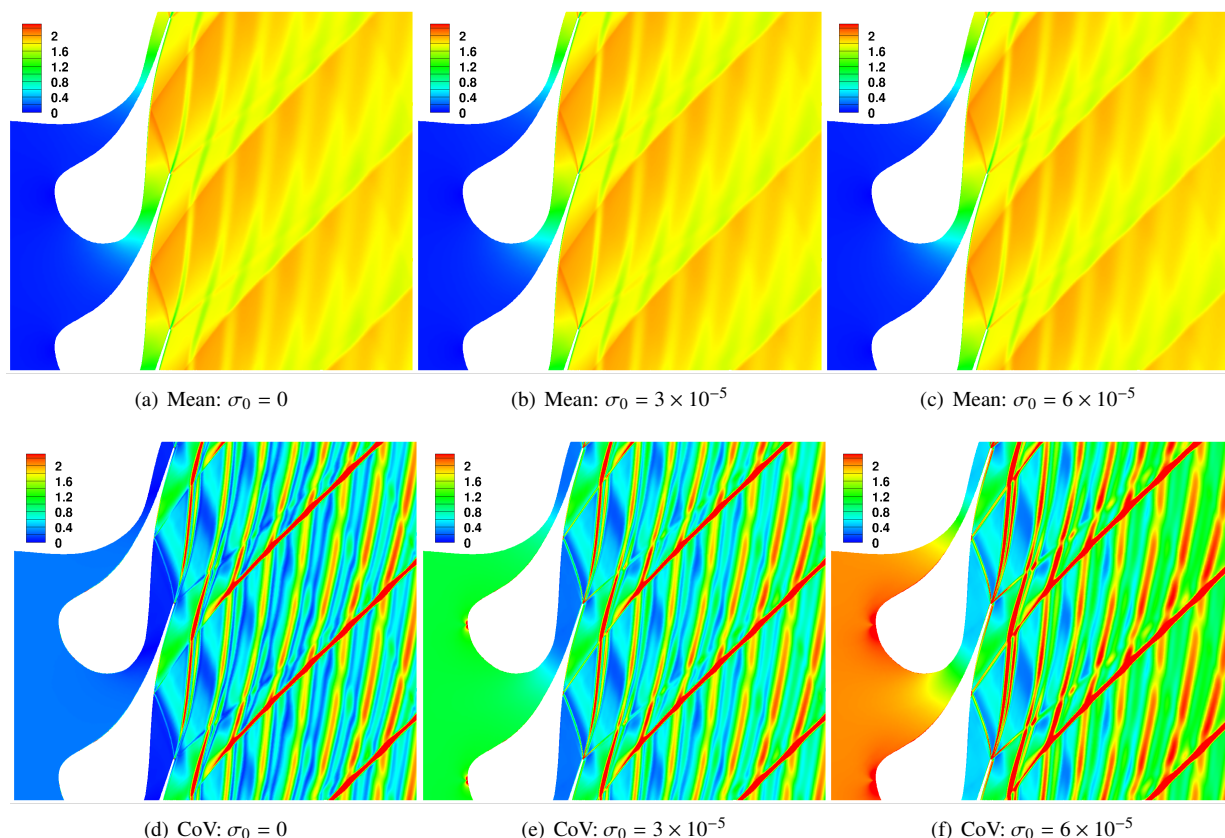


Figure 12: Mean and CoV[%] Mach contours for full load.

6. Conclusion

This paper has proposed an uncertainty quantification analysis applied to the simulation of a turbulent non-ideal flow within a supersonic turbine cascade for Organic Rankine Cycle applications. In particular, a Kriging-based method has been coupled with a Computational Fluid Dynamic solver permitting to consider multiple sources of uncertainties associated with operating conditions, fluid parameters and an accurate representation of geometric tolerances. Two main scenarios are considered, *i.e.* a full-load (a high expansion ratio equal to ≈ 6) and a part-load (pressure ratio to ≈ 4) operating conditions.

The primary outcome of the paper is an accurate statistical study of two quantities of interest, namely the total pressure loss and the mass flow rate. In particular, the sensitivity analysis, the probability density functions, and the statistical moments are computed.

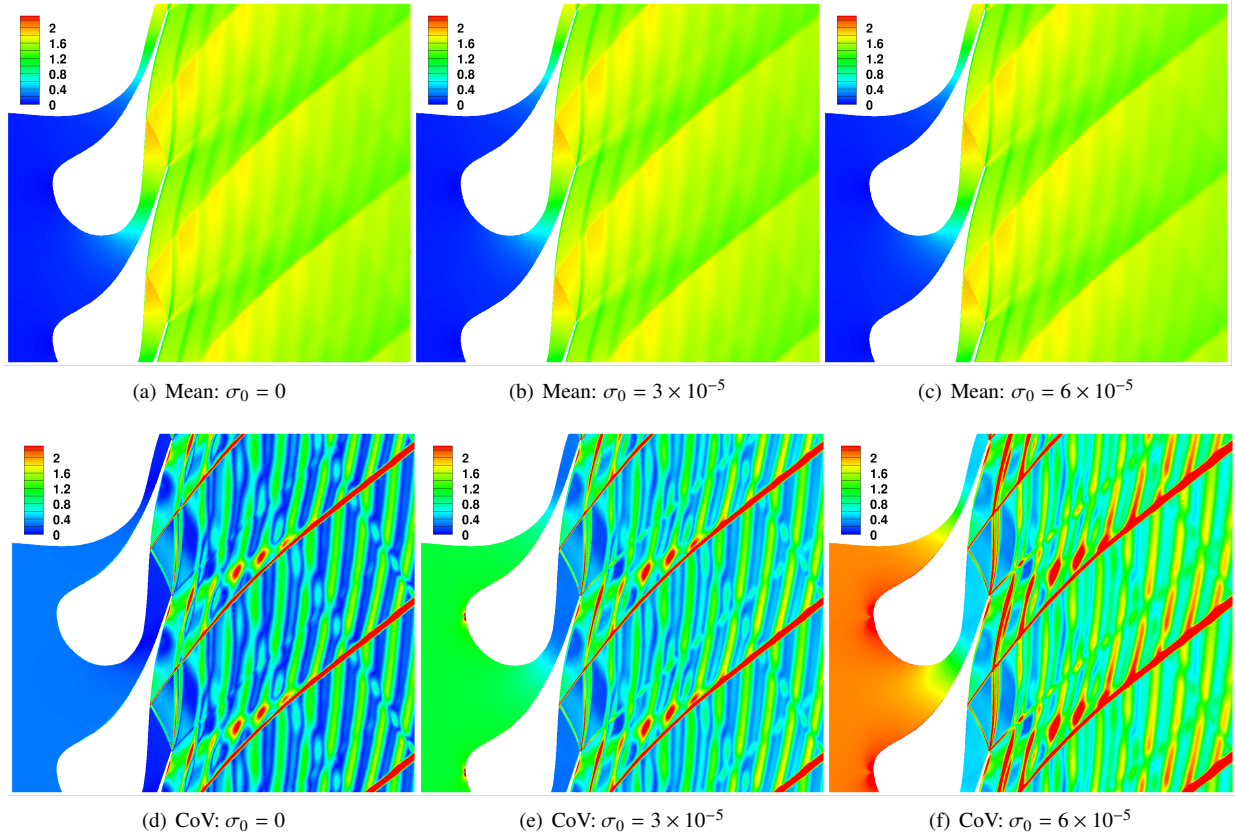


Figure 13: Mean and CoV[%] Mach contours for part load.

Concerning the sensitivity analysis, results confirm the findings of previous work; the uncertainty in the outlet static pressure has the highest impact on the loss coefficient and is predominant with respect to the thermodynamic model uncertainties; the inlet total conditions, instead, dominate on the mass flow rate. With respect to the present state-of-the art, a novel result has been observed, *i.e.* the impact of operational and model uncertainties is drastically reduced when manufacturing tolerances are considered. The modal representation of manufacturing uncertainties and the related statistical analysis allow to highlight the most relevant modes, suggesting that specific areas of the blade have the highest impact on the overall uncertainty. The support of the probability density functions of the quantities of interest enlarges significantly when the manufacturing tolerances are plugged into the uncertainty analysis, and the functions take a regular gaussian shape (differently from what observed if only operative and model uncertainties are considered). On a distributed basis, the flow released by the cascade exhibits relevant sensitivity to all sources of uncertainty, but again its statistical moments grow significantly in presence of manufacturing tolerances; as the dominating modes alter the blade shape in the rear suction side and the trailing edge thickness, the strength and inclination of the main rear shock undergoes significant variability, so as the width and defect of the blade wake.

The behaviour characterized in this work could be a very relevant statement for design. Conclusions drawn will constitute the basis for future studies about robust optimization of supersonic turbine cascades, *e.g.* minimizing one of the statistics of the Quantity of Interest (such as its mean or quantile). Moreover, considering the potential interest in using low-fidelity computations, we aim to include multi-fidelity approaches for controlling the loss of accuracy generated by the use of a low-fidelity model in the optimization loop.

References

- [1] Colonna, P., Harinck, J., Rebay, S., Guardone, A.. Real-gas effects in organic rankine cycle turbine nozzles. *Journal of Propulsion and Power* 2008;24(2):282–294.
- [2] Cinnella, P., Congedo, P.M., Pediroda, V., Parussini, L.. Quantification of thermodynamic uncertainties in real gas flows. *International Journal of Engineering Systems Modelling and Simulation* 2010;2(1):12–24.
- [3] Congedo, P., Colonna, P., Corre, C., Witteveen, J., Iaccarino, G.. Backward uncertainty propagation method in flow problems: Application to the prediction of rarefaction shock waves. *Computer Methods in Applied Mechanics and Engineering* 2012;213:314–326.
- [4] Cinnella, P., Congedo, P., Pediroda, V., Parussini, L.. Sensitivity analysis of dense gas flow simulations to thermodynamic uncertainties. *Physics of Fluids* 2011;23:116101.
- [5] Congedo, P.M., Corre, C., Cinnella, P.. Numerical investigation of dense-gas effects in turbomachinery. *Computers & Fluids* 2011;(June). URL: <http://linkinghub.elsevier.com/retrieve/pii/S0045793011002003>. doi:10.1016/j.compfluid.2011.06.012.
- [6] Congedo, P., Corre, C., Martinez, J.M.. Shape optimization of an airfoil in a BZT flow with multiple-source uncertainties. *Computer Methods in Applied Mechanics and Engineering* 2011;200(1-4):216–232. URL: <http://linkinghub.elsevier.com/retrieve/pii/S0045782510002392>. doi:10.1016/j.cma.2010.08.006.
- [7] Le Maitre, O., Knio, O.. *Stochastic Spectral Methods for Uncertainty Quantification with Applications to Computational Fluid Dynamics*. Series on Scientific Computation; Springer; 2010.
- [8] Soize, C.. *Uncertainty Quantification: An Accelerated Course with Advanced Applications in Computational Engineering*. Springer; 2017.
- [9] Congedo, P., Geraci, G., Abgrall, R., Pediroda, V., Parussini, L.. Tsi metamodels-based multi-objective robust optimization. *Engineering Computations (Swansea, Wales)* 2013;30(8):1032–1053.
- [10] Cinnella, P., Hercus, S.. Robust optimization of dense gas flows under uncertain operating conditions. *Computers & Fluids* 2010;39(10):1893 – 1908. URL: <http://www.sciencedirect.com/science/article/pii/S0045793010001519>. doi:<https://doi.org/10.1016/j.compfluid.2010.06.020>.
- [11] Geraci, G., Congedo, P., Abgrall, R., Iaccarino, G.. High-order statistics in global sensitivity analysis: Decomposition and model reduction. *Computer Methods in Applied Mechanics and Engineering* 2016;301:80 – 115. URL: <http://www.sciencedirect.com/science/article/pii/S0045782515004284>. doi:<http://dx.doi.org/10.1016/j.cma.2015.12.022>.
- [12] Bufti, E., Cinnella, P.. Efficient uncertainty quantification of turbulent flows through supersonic orc nozzle blades. *Energy Procedia* 2015;82:186 – 193. URL: <http://www.sciencedirect.com/science/article/pii/S1876610215026673>. doi:<https://doi.org/10.1016/j.egypro.2015.12.018>; 70th Conference of the Italian Thermal Machines Engineering Association, ATI2015.
- [13] Palacios, F., Colonna, M.F., Aranake, A.C., Campos, A., Copeland, S.R., Economon, T.D., et al. Stanford University Unstructured (SU2): An open-source integrated computational environment for multi-physics simulation and design. In: 51st AIAA Aerospace Sciences Meeting and Exhibit. 2013.
- [14] Economon, T.D., Mudigere, D., Bansal, G., Heinecke, A., Palacios, F., Park, J., et al. Performance optimizations for scalable implicit {RANS} calculations with {SU2}. *Computers & Fluids* 2016;129:146 – 158. URL: <http://www.sciencedirect.com/science/article/pii/S0045793016300214>. doi:<http://dx.doi.org/10.1016/j.compfluid.2016.02.003>.
- [15] Vitale, S., al., . Extension of the SU2 Open Source CFD code to the simulation of turpinibulent flows of fluids modelled with complex thermophysical laws. AIAA Aviation 2015;22nd AIAA Computational Fluid Dynamics Conference.
- [16] Gori, G., Zocca, M., Cammi, G., Spinelli, A., Guardone, A.. Experimental assessment of the open-source su2 cfd suite for orc applications. *Energy Procedia* 2017;129(Supplement C):256 – 263. doi:<https://doi.org/10.1016/j.egypro.2017.09.151>.
- [17] Giles, M.B.. Nonreflecting boundary conditions for euler equation calculations. *AIAA journal* 1990;28(12):2050–2058.
- [18] Roe, P.L.. Approximate riemann solvers, parameter vectors, and difference schemes. *J Comput Phys* 1981;43(2):357–372. URL: <http://linkinghub.elsevier.com/retrieve/pii/0021999181901285>.
- [19] Vinokur, M., Montagné, J.L.. Generalized flux-vector splitting and roe average for an equilibrium real gas. *J Comput Phys* 1990;89:276.
- [20] Guardone, A., Vigevano, L.. Roe linearization for the van der Waals gas. *J Comput Phys* 2002;175:50–78.
- [21] Menter, F.. Zonal two equation kw turbulence models for aerodynamic flows. In: 23rd fluid dynamics, plasmadynamics, and lasers conference. 1993, p. 2906.
- [22] Garzon, V.E.. Probabilistic aerothermal design of compressor airfoils. Ph.D. thesis; Massachusetts Institute of Technology; 2003.
- [23] Häcker, J.. Statistical analysis of manufacturing deviations and classification methods for probabilistic aerothermal design of turbine blades. Master's degree thesis 2000.
- [24] Lange, A., Voigt, M., Vogeler, K., Johann, E.. Principal component analysis on 3d scanned compressor blades for probabilistic cfd simulation. In: 53rd AIAA/ASME/ASCE/AHS/ASC Structures, Structural Dynamics and Materials Conference 20th AIAA/ASME/AHS Adaptive Structures Conference 14th AIAA. 2012, p. 1762.
- [25] Dow, E.A., Wang, Q.. The implications of tolerance optimization on compressor blade design. *Journal of Turbomachinery* 2015;137(10):101008.
- [26] Dow, E.A., Wang, Q.. Optimal design and tolerancing of compressor blades subject to manufacturing variability. In: 16th AIAA Non-Deterministic Approaches Conference. 2014, p. 1008.
- [27] Betz, W., Papaioannou, I., Straub, D.. Numerical methods for the discretization of random fields by means of the karhunen–loève expansion. *Computer Methods in Applied Mechanics and Engineering* 2014;271:109–129.
- [28] Karhunen, K.. *Über lineare Methoden in der Wahrscheinlichkeitsrechnung*; vol. 37. Sana; 1947.
- [29] Loeve, M.. *Functions aleatoires du second ordre. Processus stochastique et mouvement Brownien* 1948;:366–420.
- [30] Nyström, E.J.. *Über die praktische auflösung von integralgleichungen mit anwendungen auf randwertaufgaben*. *Acta Mathematica* 1930;54(1):185–204.
- [31] De Boer, A., Van der Schoot, M., Bijl, H.. Mesh deformation based on radial basis function interpolation. *Computers & structures* 2007;85(11-14):784–795.

- [32] Pini, M., Persico, G., Pasquale, D., Rebay, S.. Adjoint method for shape optimization in real-gas flow applications. *ASME Journal of Engineering for Gas Turbines and Power* 2015;137(3).
- [33] C. E. Rasmussen, C.W.. *Gaussian Processes for Machine Learning*. the MIT Press; 2006.
- [34] Dubourg, V.. *Adaptive surrogate models for reliability analysis and reliability-based design optimization*. PhD Thesis; 2011.
- [35] Saltelli, A., Tarantola, S., Campolongo, F.. *Sensitivity Analysis in Practice*. Wiley; 2004.
- [36] Sobol, I.M.. Global sensitivity indices for nonlinear mathematical models and their monte carlo estimates. *Mathematics and computers in simulation* 2001;55(1-3):271–280.
- [37] Miranda, A.A., Le Borgne, Y.A., Bontempi, G.. New routes from minimal approximation error to principal components. *Neural Processing Letters* 2008;27(3):197–207.
- [38] Romei, A., Persico, G., Congedo, P.M.. Assessment of deterministic shape optimizations within a stochastic framework for supersonic organic rankine cycle nozzle cascades. *Journal of Engineering for Gas Turbines and Power* 2018;Under Review.