



Turbulence modelling for CFD

Remi Manceau

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Turbulence modelling for CFD

RÉMI MANCEAU

Senior CNRS Researcher



Applied Mathematics Departement

Inria-Cagire group

CNRS–University of Pau

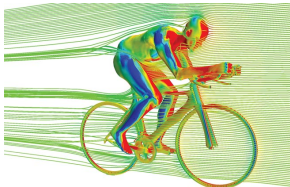
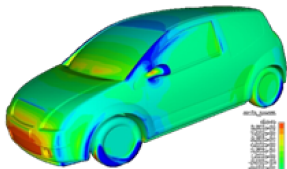
Pau, France

ENSGTI

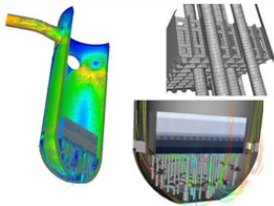
2023-2024

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<https://team.inria.fr/rmanceau/documents>



LOCAL FLOW CONDITIONS IN A PWR LOWER CORE



Outline

I. Introduction to CFD for turbulent flows

- ✓ Issues of CFD
- ✓ Computational cost linked to turbulence
- ✓ The different modeling approaches
- ✓ Global picture of CFD codes

II. Industrial standard: RANS

- ✓ Closure problem, similarities with continuum mechanics
- ✓ Physical and mathematical principles
- ✓ Eddy-viscosity models
- ✓ Reynolds-stress models
- ✓ Problems due to walls

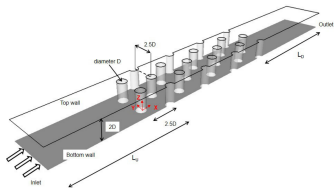
III. Heat transfer modelling

- ✓ General equations and approximations
- ✓ Modelling in the forced convection regime
- ✓ Modelling in the mixed and natural convection regimes

Introduction to computational fluid dynamics (CFD) for turbulent flows

1. Challenges of numerical simulation

1.1. What is CFD?

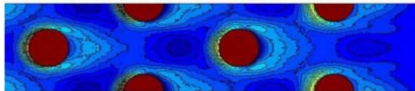


Configuration to be studied:

Here, the flow above a wall with pins
(cooling of turbine blades)

Experimental approach

Measurement of the wall
temperature
Infrared camera



Computer program to solve them

Numerical approach

Conservation equations

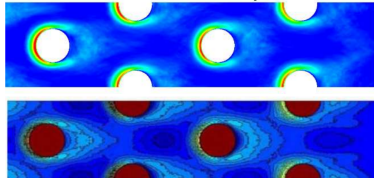
- ✓ Mass
- ✓ Momentum
- ✓ Energy

```
do iel=1,ncell
  pij = xprod(iii,jjj)
  phii1 = -cvara_ep(iel)*
    (cssgs1*xaniso(iii,jjj)+cssgs2*(aikakj-d1s3*deltij*aai))
  phii2 = - cssgr1*trprod*xaniso(iii,jjj)
    + trrij*xstrai(iii,jjj)*(cssgr2-cssgr3*sqrt(aai))
    + cssgr4*trrij*(aiksjk-d2s3*deltij*aklskl)
    + cssgr5*trrij* aikrjk
  epsij = -d2s3*cvara_ep(iel)*deltij
  w1(iel) = crom(iel)*cell_f_vol(iel)*(pij+phii1+phii2+epsij)
  w2(iel) = cell_f_vol(iel)/trrij*crom(iel)*
    cssgs1*cvara_ep(iel) + cssgr1*max(trprod,0.d0) )
end do
```

Numerical simulation



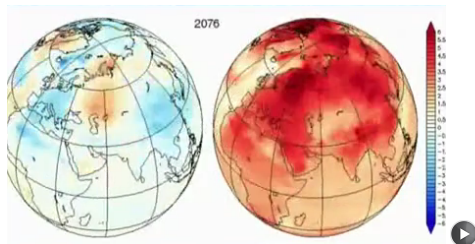
Results: wall temperature



1.2. Numerical simulation: Why?

1.2.1. Because experiments are impossible

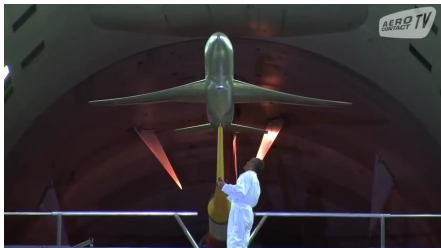
Collision Milky Way/Andromeda
4 billion years from now



Climate modelling

1.2.2. Because measurements are long/complex/costly/partial

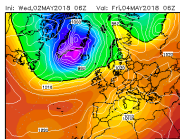
Aerodynamics of cars...



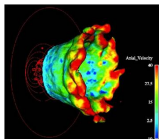
...and planes

1.3. Fields of application

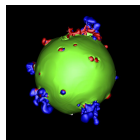
1.3.1. Fundamental and applied research



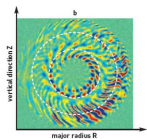
Environmental studies



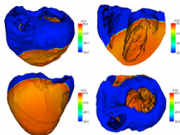
Combustion



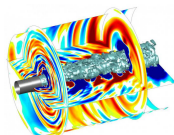
Astrophysics



Nuclear physics
among many others...

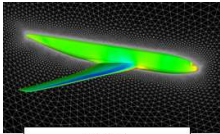


Biology

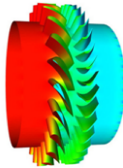


Acoustics

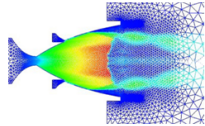
1.3.2. Examples of industrial applications (RANS)



A320G

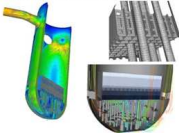


Vega2 Turbine

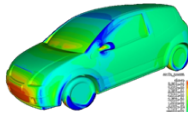


ATAC Tuiere
Vulcain

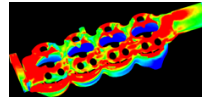
LOCAL FLOW CONDITIONS IN A PWR LOWER CORE



Nuclear power plant

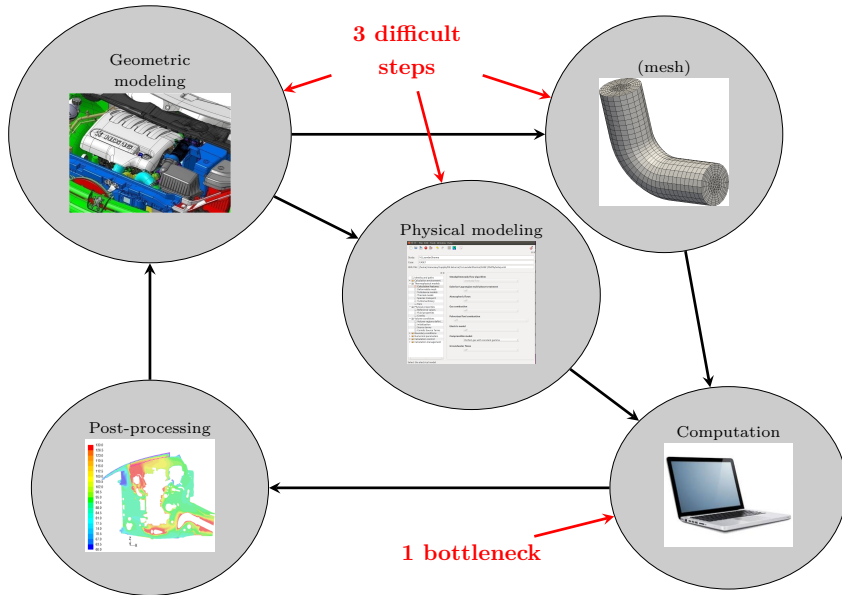


Car aerodynamics

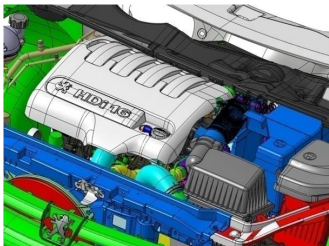


Engine cooling

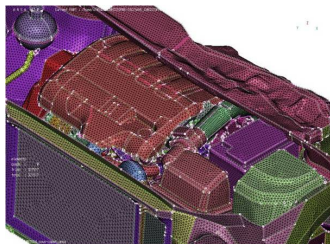
2. Components of a computation



2.1. Geometrical modeling



**Numerical definition
(CAD)**

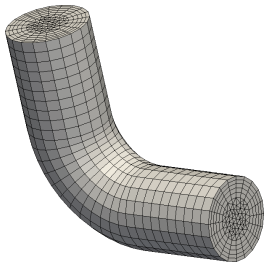


Surface mesh

- ✓ What is to be meshed is not the elements, but the volume between the elements, contrary to structural analysis.
- ✓ An external “skin” is needed
- ✓ Only useful elements must be kept
- ✓ Details must be removed (reduction of the size of the mesh)

2.2. Volume discretization (mesh)

- ✓ No analytical solution of the Navier-Stokes equations
- ✓ Example: finite volume discretization: the fluid volume is divided into small control volumes (cells) in which the variables (3 velocity components, pressure) are considered constants



N cells= N values for each variable

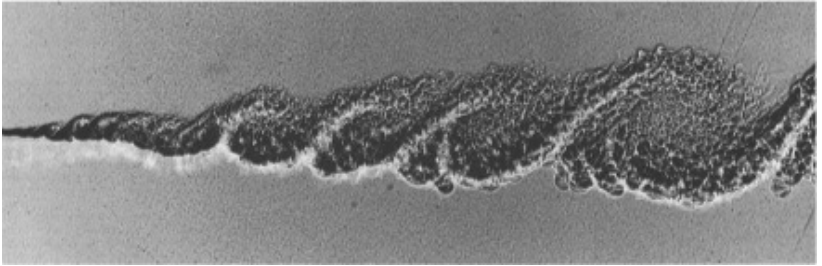
- ✓ Approximation of the equations $\Rightarrow 4 \times N$ system of equations for $4 \times N$ unknowns
 \Rightarrow can be solved by a computer
- ✓ Fine mesh (large N) \Rightarrow accurate solution (convergence toward the exact solution)

3. Turbulence and computational cost

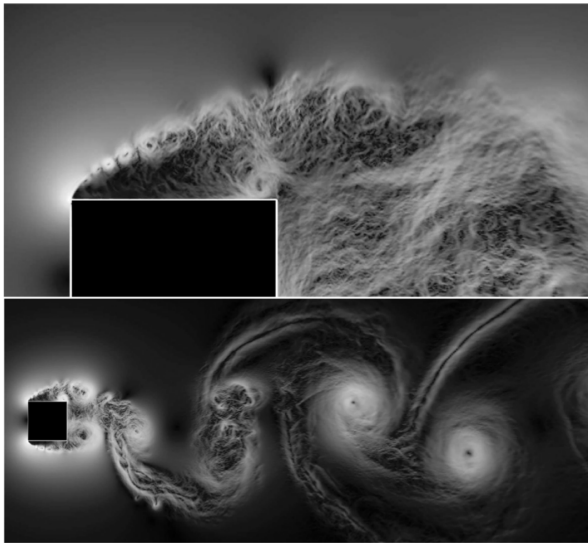
3.1. Turbulence is everywhere



3.2. Turbulent scales



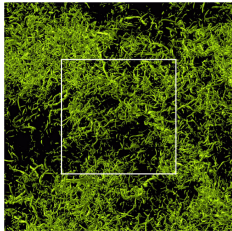
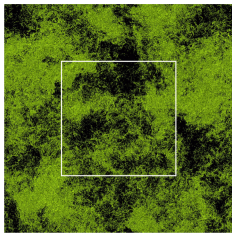
Vizualisation of a mixing layer. From Brown and Roshko (1974).



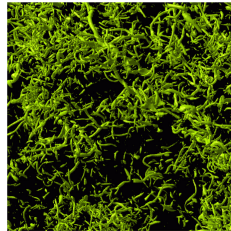
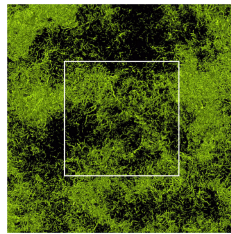
Flow around a square cylinder at $Re = 22000$ (DNS, Trias *et al.*, 2015)

Visualization of the magnitude of the pressure gradient

Movie at: <https://www.youtube.com/watch?v=c8zKWaxohng>



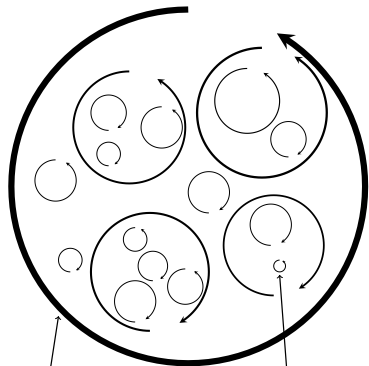
Close-up (scale 4)



Close-up (scale 8)

DNS of homogeneous turbulence at $Re_\lambda = 732$.

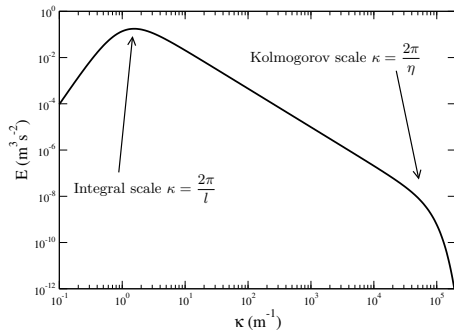
From Yokokawa *et al.* (2002).



Integral scale l

Kolmogorov scale η

Physical space

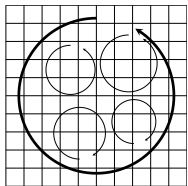
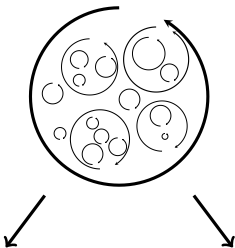


Fourier space

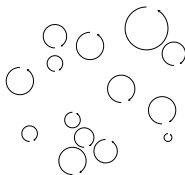
(naive pictures)

3.3. Influence of the mesh

- ✓ Mesh not sufficiently fine \Rightarrow the smallest eddies are not resolved



Resolved eddies



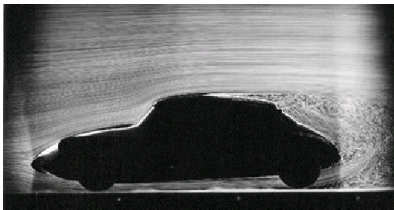
Unresolved eddies

3.3.1. Free turbulence (no walls)

- ✓ Ratio of the largest vortices to the smallest vortices: $\frac{l}{\eta} \propto Re_t^{3/4}$

$Re_t = \frac{ul}{\nu}$ where u and l are the characteristic scales of the largest eddies

- ✓ Example: wake of a car
(100 km h⁻¹)



$$Re_t \simeq 150000 \quad \Rightarrow \quad \frac{l}{\eta} \simeq 7500 \quad \Rightarrow \quad \eta \simeq 100 \mu\text{m}$$

- ✓ η determines the grid step for a direct numerical simulation of turbulence (DNS)
- ✓ The number of grid cells is proportional to $Re_t^{9/4}$

3.3.2. Wall turbulence (boundary layer)

- ✓ Ratio of the largest vortices to the smallest vortices $\propto Re_\tau$

$Re_\tau = \frac{\delta u_\tau}{\nu}$ where δ is the boundary layer thickness and u_τ the friction velocity

- ✓ Number of grid cells $\propto Re_\tau^{11/4}$ (less than 3, due to the expansion in the direction normal to the wall)

\leadsto Example: boundary layer of a car (100 km h^{-1})

Smallest vortices $\simeq 15 \mu\text{m}$

3.4. Evaluation of the computational cost

- ✓ Number of floating point operations per iteration:

Many numerical methods have a computational cost (number of operations) proportional to $N \ln N$ (where N is the number of cells)

- ✓ Time step linked to the grid step for stability and/or accuracy reasons:

Courant-Friedrich-Levy (CFL) number: $\text{CFL} = \frac{U \Delta t}{\Delta x} \simeq 1$

\Rightarrow time-step Δt proportional to $Re^{-3/4}$

- ✓ For a good convergence of the statistics, the “fluid particles” must travel through the domain at least 10 times.
- ✓ Example: car (100 km h^{-1})

▷ Number of grid cells $\simeq 10^{18}$

▷ Number of time steps $\simeq 10^6$

▷ Memory requirement $\simeq 10^{20}$ bytes

▷ Computational cost $\simeq 10^{27}$ floating point operations

3.5. Available computer power

3.5.1. Performance of processors

✓ Measured in Flops: FLoating-Point Operations Per Second

= clock frequency \times number of operations per clock cycle

This unit is a measure of the available brute force

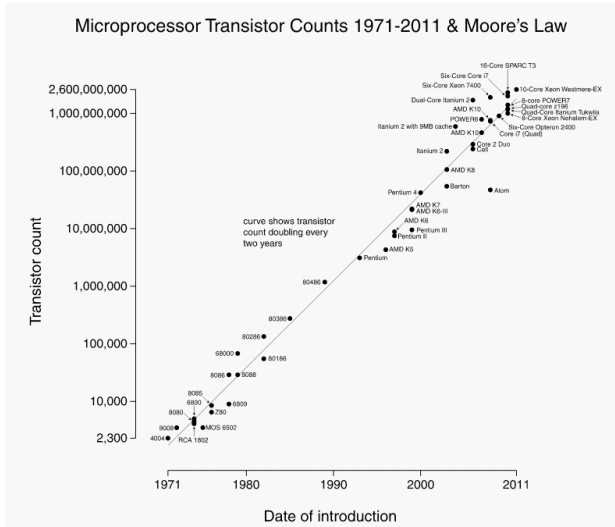
✓ This theoretical performance is... theoretical

\leadsto Peak performance = number of operation the processor(s) (CPU) are able to reach

\leadsto Actual performance = performance really achieved during a computation

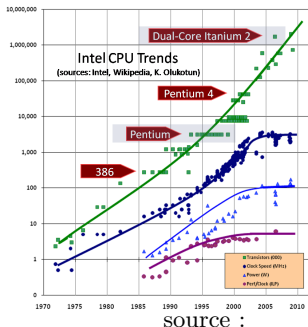
✓ Can be much lower, in particular for parallel computer

- ✓ **Moore's law** : Moore (1975) : The number of transistors of a processor double every second year (10 years=factor of 32)



source: <http://www.tablette-tactile.net>

- ✓ Supposed to be valid until about 2015 (“quantum wall”)
- ✓ However, the clock frequency increase slowed down after 2003 and in particular after 2006: “thermal wall” = the heat produced cannot be removed anymore

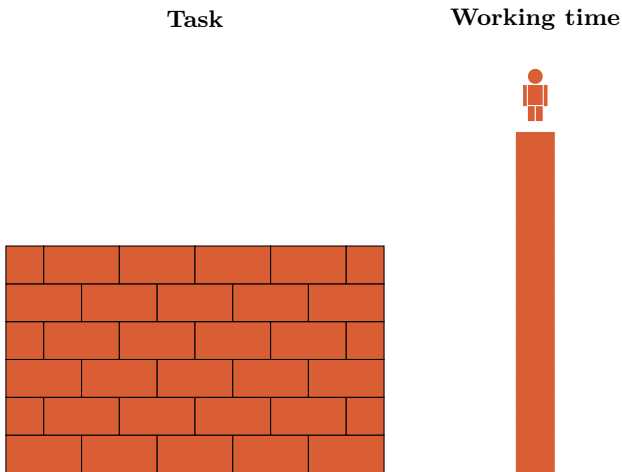


source : <http://www.gotw.ca/publications>

⇒ Evolution toward multi-core processors for personal computers

3.5.2. Supercomputers

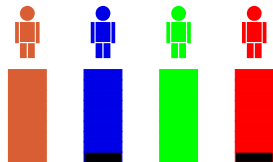
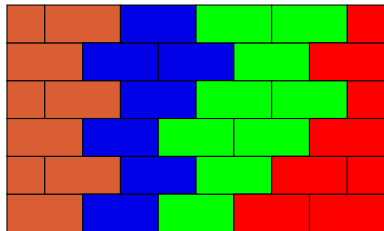
- ✓ Analogy: building of a wall (adapted from V. Perrier)



How to reduce the working time (computational time)?

✓ Task decomposition

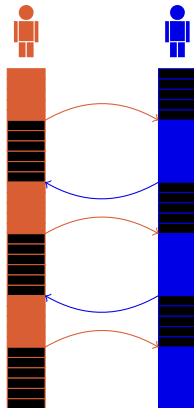
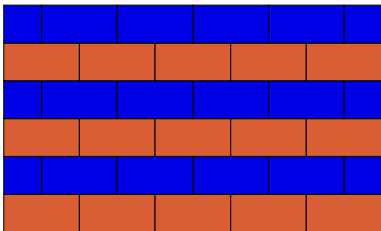
~ Principle



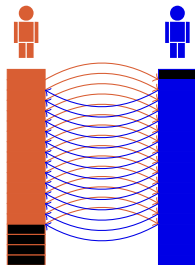
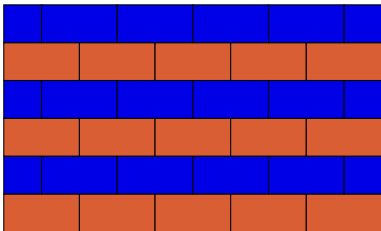
Does not work: take care of the scheduling of the tasks!

✓ Optimization of the computing time \Rightarrow find a high-performance scheduling

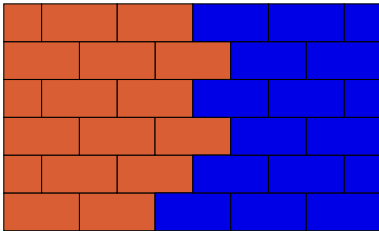
\Rightarrow HPC=High-Performance Computing



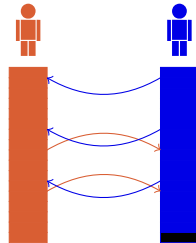
Too much waiting time



Too many communications (slow)



Excellent!



✓ Vectorial machines

↪ Designed for computing the same operation onto a vector (vectorial processor) rather than a real (scalar processor)

↪ Very efficient treatment of loops:

```
DO I=1,10000
```

```
    X(I)=Y(I)*Z(I)
```

```
END DO
```

Scalar processor: sequential treatment of the iterations

Vectorial processors: N data treated simultaneously

↪ Historically: the Cray family

Cray-2: first supercomputer above the GFlops in 1985

↪ Many others, among which the Playstation 2

↪ Their cost is such that they are replaced today by parallel machines

✓ Parallel machines

Association of several processors or cores (from 2 to millions)

Laptop



- ✓ 6 cores
- ✓ 122 GFlops (Giga = 10^9)
- ✓ 16 GB

Cluster
Univ. Pau



- ✓ 1088 cores
- ✓ 73 TFlops (Tera = 10^{12})
- ✓ 4.6 TB

Jean Zay
IDRIS
Paris

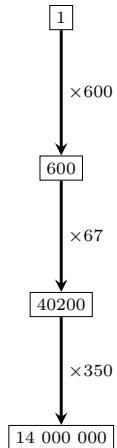


- ✓ 61 120 cores
- ✓ 4.9 Pflops (Peta = 10^{15})
- ✓ 293 TB

Frontier
USA



- ✓ 8 730 112 cores
- ✓ 1.7 Eflops (Exa = 10^{18})
- ✓ 9.2 PB

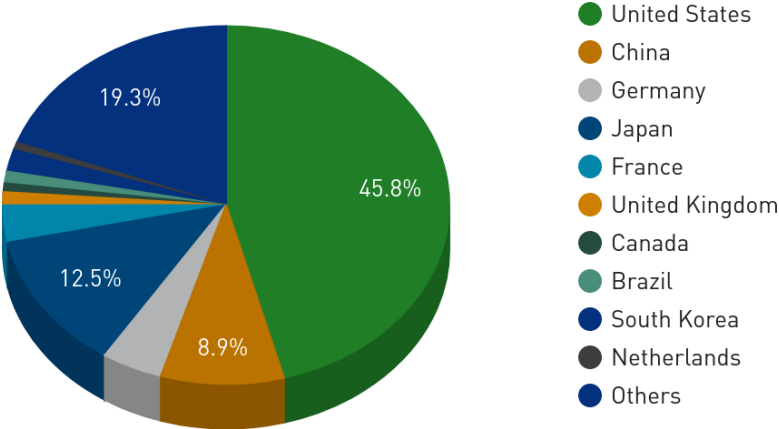


3.5.3. World ranking (Nov 2023)

Rank	System	Cores	Rmax (PFlop/s)	Rpeak (PFlop/s)	Power (kW)
1	Frontier - HPE Cray EX235a, AMD Optimized 3rd Generation EPYC 64C 20Hz, AMD Instinct MI250X, Slingshot-11, HPE DOE/SC/Oak Ridge National Laboratory United States	8,699,904	1,194.00	1,679.82	22,703
2	Supercomputer Fugaku - Supercomputer Fugaku, A64FX 68C 2.2GHz, Tofu interconnect D, Fujitsu RIKEN Center for Computational Science Japan	7,630,848	442.01	537.21	29,899
3	LUMI - HPE Cray EX235a, AMD Optimized 3rd Generation EPYC 64C 20Hz, AMD Instinct MI250X, Slingshot-11, HPE EuroHPC/CSC Finland	2,220,288	309.10	428.70	6,016
4	Leonardo - Bull/Sequana XH2000, Xeon Platinum 8358 32C 2.6GHz, NVIDIA A100 5XM4 64 GB, Quad-rail NVIDIA HDR100 Infiniband, Atos EuroHPC/CINECA Italy	1,824,768	238.70	304.47	7,404
5	Summit - IBM Power System AC922, IBM POWER9 22C 3.07GHz, NVIDIA Volta GV100, Dual-rail Mellanox EDR Infiniband, IBM DOE/SC/Oak Ridge National Laboratory United States	2,414,592	148.60	200.79	10,096
6	Sierra - IBM Power System AC922, IBM POWER9 22C 3.1GHz, NVIDIA Volta GV100, Dual-rail Mellanox EDR Infiniband, IBM / NVIDIA / Mellanox DOE/NNLS/LNL United States	1,572,480	94.64	125.71	7,438
7	Sunway TaihuLight - Sunway MPP, Sunway SW26010 260C 1.45GHz, Sunway, NRCPC National Supercomputing Center in Wuxi China	10,649,600	93.01	125.44	15,371
8	Perlmutter - HPE Cray EX235n, AMD EPYC 7763 44C 2.45GHz, NVIDIA A100 5XM4 40 GB, Slingshot-10, HPE DOE/SC/LBNL/NERSC United States	761,856	70.87	93.75	2,589
9	Selene - NVIDIA DGX A100, AMD EPYC 7742 64C 2.25GHz, NVIDIA A100, Mellanox HDR Infiniband, Nvidia NVIDIA Corporation United States	555,520	63.46	79.22	2,646
10	Tianhe-2A - TH-IVB-FEP Cluster, Intel Xeon E5-2692v2 12C 2.2GHz, TH Express-2, Matrix-2000, NUDT National Super Computer Center in Guangzhou China	4,981,760	61.44	100.68	18,482

Top 10 of world rankings,
military excluded (source :
<http://www.top500.org>)

Countries Performance Share

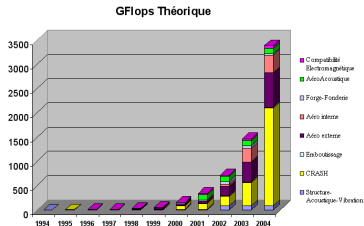


Country performance share in the top 500

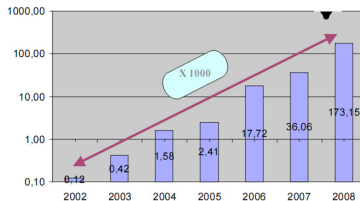
Rank	System	Cores	Rmax (PFlop/s)	Rpeak (PFlop/s)	Power (KW)
12	Adastra - HPE Cray EX235a, AMD Optimized 3rd Generation EPYC 64C 20GHz, AMD Instinct MI250X, Slingshot-11, HPE Grand Equipement National de Calcul Intensif - Centre Informatique National de l'Enseignement Supérieur (GENCI-CINES) France	319,072	46.10	61.61	921
22	CEA-HF - BullSequana XH2000, AMD EPYC 7763 64C 2.45GHz, Alos BXI V2, Atos Commissariat à l'Energie Atomique (CEA) France	810,240	23.24	31.76	4,959
39	PANGEA III - IBM Power System AC922, IBM POWER9 18C 3.45GHz, Dual-rail Mellanox EDR Infiniband, NVIDIA Volta GV100, IBM Total Exploration Production France	291,024	17.86	25.03	1,367
54	Tera-1000-2 - Bull Sequana X1000, Intel Xeon Phi 7250 68C 1.4GHz, Bull BXI 1.2, Atos Commissariat à l'Energie Atomique (CEA) France	561,408	11.97	23.40	3,178
78	Taranis - Bull Sequana XH2000 , AMD EPYC 7742 64C 2.25GHz, Mellanox InfiniBand HDR100, Atos Meteo France France	294,912	8.19	10.32	1,672
87	Belenos - Bull Sequana XH2000 , AMD EPYC 7742 64C 2.25GHz, Mellanox HDR100, Atos Meteo France France	294,912	7.68	10.47	1,655
92	JOLIOT-CURIE ROME - Bull Sequana XH2000 , AMD Rome 7H12 64C 2.6GHz, Mellanox HDR100, Atos CEA/TGCC-GENCI France	197,120	6.99	12.04	1,436
119	Pangea - SGI ICE X, Xeon Xeon E5-2670/ E5-2680v3 12C 2.5GHz, Infiniband FDR, HPE Total Exploration Production France	220,800	5.28	6.71	4,150
135	Jean Zay - HPE SGI 8400, Xeon Gold 6248 20C 2.5GHz, NVIDIA Tesla V100 5XM2, Intel Omni-Path, HPE CNRS/DRIS-GENCI France	93,960	4.48	7.35	
139	CRONOS - BullSequana X, Xeon Platinum 8260 24C 2.4GHz, Infiniband HDR, Atos EDF France	81,600	4.30	7.14	1,226

Top 10 in France

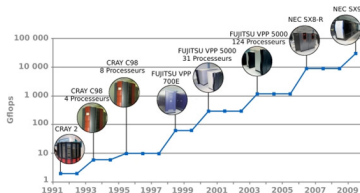
3.5.4. Evolution of the available computing power



At PSA

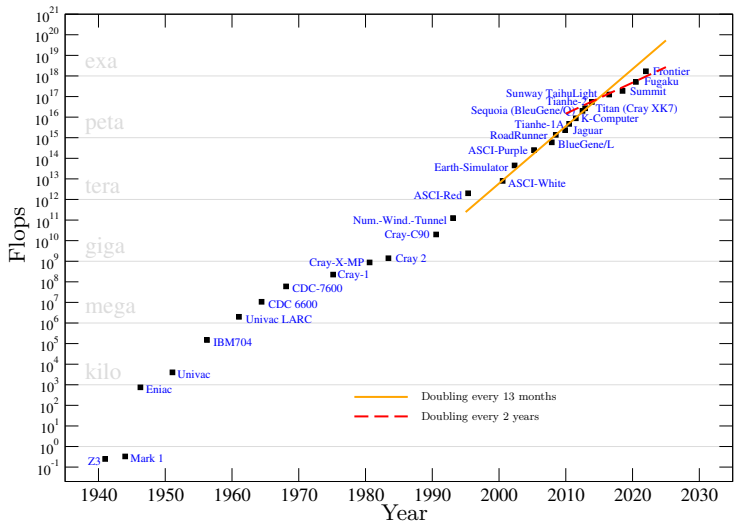


At EDF



At Météo France

Rapid increase in the early 2000s
Examples in the french industry



Evolution of the most powerfull computer from 1940
(adapted from T.B. Gatski, private communication)

- ✓ The memory necessary to compute the flow around a car at 100 km h^{-1} represents 11000 times the largest supercomputer today \Rightarrow The computation is far from possible
- ✓ If the computation were possible, it would take 18 years!
- ✓ Extrapolation: if the increase of the computing power continues at the same pace as today, DNS will be possible in car/aerospace engineering in 2080 (evaluation from Spalart, Boeing)
- ✓ Using available computer power in the car industry today, it would be possible to perform DNS of cars at only 1 km h^{-1}
- ✓ Using 100% of the largest super-computer:
DNS of a car at 3 km h^{-1}
Memory limitation.

✓ Is such computing power desirable?



Frontier (rank 1)

Power: 21 MW

(Electric consumption of a town of $\simeq 27000$ inhabitants)

- ✓ The ecological footprint in terms of CO₂ emitted by one CPU hour in France has been estimated at 4.68 g (Berthoud *et al.*, 2020)

- ✓ For example, Perez Arroyo's (heroic) calculation (2021) of a fan-compressor-combustion chamber assembly of an aircraft engine in LES (with wall functions) required 31.6 million CPU hours on the Joliot-Curie national supercomputer, which corresponds to about 150 tons of CO₂, i.e.:
 - ↪ 1.2 million km for a car

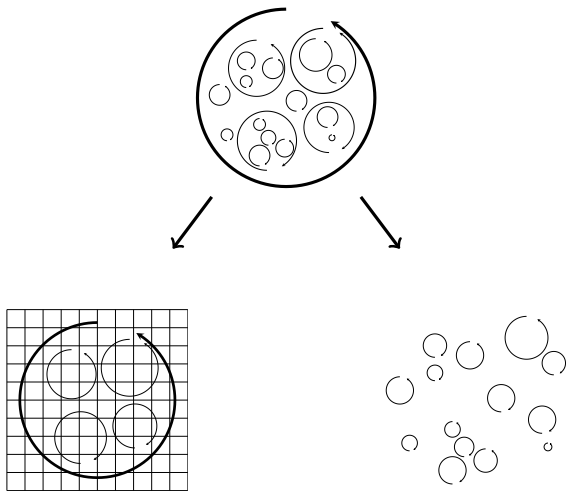
 - ↪ 3000 km of flight of an A320

 - ↪ The average consumption of a French person during fifteen years

- ✓ The 18-year calculation cited above would result in 6.7 million tons of CO₂ (emissions of about 700,000 French people during one year)

4. Consequences for the computation of turbulent flows

- ✓ Obtaining the complete range of turbulent scales is not possible \Rightarrow **modeling approaches**: some scales are not resolved



Resolved eddies

Unresolved eddies

4.1. Different modeling approaches

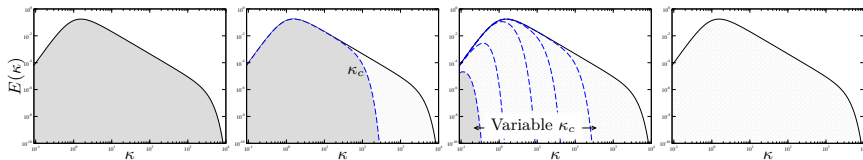
- ✓ **Large-eddy simulation (LES)**: the energetic scales are resolved, but not the dissipative scales
 - ↪ Reliable method for fine enough grids
 - ↪ Problem: close to the wall, cost \simeq cost DNS
- ✓ **Reynolds-averaged Navier-Stokes (RANS)** approach: only statistics are computed (average, second moments)
 - ↪ Affordable cost \Rightarrow Standard in the industry (commercial codes)
 - ↪ Problem: not always reliable; only statistics (unsteady information missing)
- ✓ **Hybrid LES/RANS approaches**:
 - ↪ Somewhere in between
 - ↪ Many different approaches: URANS, SDM, OES, VLES, DES, LNS, PANS, XLES, PITM, TPITM, FSM, SBES, *etc.*
 - ↪ In particular : LES in some regions/RANS in other regions (wall)



Resolved scales



Modelled scales



DNS

LES

Hybrid RANS/LES

RANS



DNS

Avail.: 2080



LES

Avail.: 2070
Cost wall region \simeq
DNS
Avail. : 2045 with wall
functions



Hybrid

Avail.: 2000
Rapid development

Variable according to the region



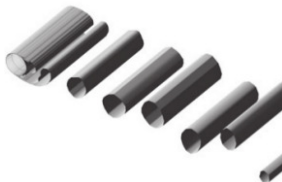
RANS

Avail.: 1985
Industrial standard

Example: flow around a circular cylinder



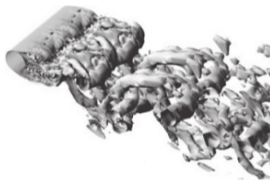
RANS



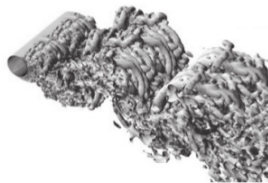
2D URANS



3D URANS



Hybrid (DES)
coarse grid



Hybrid (DES)
fine grid \simeq LES

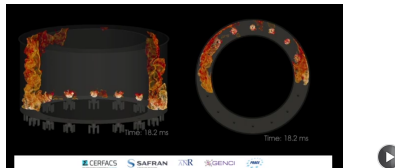
From Spalart (2009), $Re = 50000$.

	RANS	Hybrid	LES	DNS
Availability (external aerodynamics)	1985	2000	2070 (2045 if wall functions)	2080
Applications	Standard method in the industry	Applied research	Applied research	Research
Active research since	1877 (theory) 1965 (computations)	1997	1963	1987
Maturity	Mature	Immature	Mature	Mature

4.2. Use in the industry

- ✓ LES: is used in applied research for internal aerodynamics (engines, combustion chambers, power plants), *etc.*) and can be envisaged in the near future for sub-domains in external aerodynamics (local LES around a rear-view mirror, for instance).

Combustion chambers: starts to be used for *design* instead of RANS (niche application).



LES of ignition in a combustion chamber

- ✓ Some hybrid methods are already available in most of the codes, but their low level of maturity confine them to applied research for the moment.
- ✓ The standard for industrial design is the RANS method.

4.3. Formalism

✓ An operator $\overline{\cdot}$ is applied to the equations of motion (Navier-Stokes equations)

\leadsto RANS: operator = statistical average

\leadsto LES: operator = low-pass convolution filter

✓ Physical variables (velocity \mathbf{u}^* , pressure p^*) are decomposed into

$$\mathbf{u}^* = \underbrace{\overline{\mathbf{u}^*}}_{\text{Resolved field}} + \underbrace{\mathbf{u}}_{\text{Non-resolved field}}$$

- ✓ Navier-Stokes equations:

$$\frac{\partial u_i^*}{\partial t} = \mathcal{F}(u_i^*, p^*)$$

- ✓ Decomposition + Application of the operator \Rightarrow

$$\frac{\partial \overline{u_i^*}}{\partial t} = \mathcal{F}(\overline{u_i^*}, \overline{p^*}) - \frac{\partial \tau_{ij}}{\partial x_j}$$

- ✓ Due to the nonlinearity of the equations:

$$\tau_{ij} = \overline{u_i^* u_j^*} - \overline{u_i^*} \overline{u_j^*}$$

- ✓ Represents the influence of the non-resolved field on the resolved field (stress)

\leadsto RANS: τ_{ij} = Reynolds-stress tensor

\leadsto LES: τ_{ij} = subfilter-stress (ou subgrid-stress) tensor

$\Rightarrow \tau_{ij}$ **requires modeling**

5. Global picture of the CFD codes

✓ Codes can be classified into three main categories:

~> “Home made” codes

~> Collaborative codes

~> Commercial codes

5.1. “Home made” codes

- ✓ Developed in research institutions by teams or even individuals
- ✓ Often very specific: weakly flexible but very efficient (confined to simple geometries, to a specific physical phenomenon, a particular method, *etc.*)
- ✓ There are many of them

5.2. Collaborative codes (French centered)

- ✓ Some research centers (CEA, ONERA, CERFACS, IFPEN, *etc.*) or companies (Airbus, Dassault, EDF, *etc.*) develop and use collaborative codes (several groups or several research centers, many users)
- ✓ This development implies a significant financial effort
- ✓ But is beneficial for:
 - ↪ developing the internal skills and knowledge
 - ↪ developing specific applications
 - ↪ avoiding the “black-box” effect of commercial codes
- ✓ Examples:
 - ↪ TrioCFD (CEA)
 - ↪ elsA (ONERA-Airbus-SAFRAN)
 - ↪ CEDRE (ONERA)
 - ↪ Code_Saturne (EDF)
 - ↪ AVBP (CERFACS, IFPEN)

5.3. Commercial codes

- ✓ Often very generalist (“able to address all the problems”)
- ✓ Licenses, in general expensive, must be bought (*very* expensive for parallel computing)
- ✓ They are black-boxes (no access to the source code)
- ✓ They can be modified using *user subroutines*
- ✓ Examples:
 - ~> Fluent
 - ~> StarCD
 - ~> StarCCM+
 - ~> CFX
 - ~> Powerflow
 - ~> Xflow
 - ~> *etc.*

5.4. Open-source initiatives

- ✓ The sources (the code, not only the executable files) of some codes are publicly available, freely distributed under an open-source license
 - ↪ Code_Saturne by EDF
 - ↪ OpenFoam
 - ↪ Incompac3d by Pprime/Imperial College
- ✓ This strategy aims at:
 - ↪ Enabling a development at a lower cost, by a community of users: Linux type strategy
 - ↪ Enabling a community of users, exchanges, large-scale validation
 - ↪ Sell associate services (advice, studies, *etc.*) : Google type strategy
 - ↪ Demonstrate the skills and knowledge of the company (EDF case)

5.5. Distinctive features of the codes

- ✓ Beside the numerical methods, the type of meshes, *etc.*, codes have their own features.
- ✓ They often propose both RANS and LES models
- ✓ Many have now hybrid models (DES, SAS)
- ✓ There is a rise in Lattice-Boltzmann methods (Powerflow, Xflow, ProLB)

6. Conclusion of this general introduction

- ✓ In CFD, the limiting factor is the computing power, because of turbulence
- ✓ One has to model turbulence
- ✓ For industrial applications (design), the standard method is RANS modeling
- ✓ Many parameters have an influence on the results:
 - ↪ The mesh
 - ↪ Boundary conditions/size of the domain
 - ↪ Discretization schemes
- ✓ But the most influential choice is the turbulence model: it is very important to understand the underlying hypotheses and the limitations

- ✓ The only rigorous way to ensure the quality of the results would be to test, for every case, the influence of all these parameters \Rightarrow perform a large number of computations
- ✓ In practice, this is impossible
- ✓ Know-how must be built-up
- ✓ Oftentimes engineers follow a fixed methodology: recommended model, mesh type, domain size, numerical schemes, *etc.*
- ✓ The experience gained increases the knowledge of the range of application of computations (repeatability, influential parameters, recurrent modeling problems, experiment/computation discrepancies, *etc.*)
- ✓ This know-how must be sustained:
 - \leadsto models evolve
 - \leadsto computing power increases very fast

Statistical turbulence modeling

(RANS)

1. Introduction

- ✓ Aim of turbulence modeling: to replace the Navier-Stokes equations by a model.
 - ↪ The numerical resolution of the equations (the model) must be as cheap as possible: overnight computations are necessary for parametric studies and much shorter for optimization.
 - ↪ The model must be predictive: only the parameters of the flow must be necessary \Rightarrow no a priori knowledge of the solution.
 - ↪ The model must represent at best the physics of the flow.

~> The model must give the useful quantities:

- ▷ At least global quantities: aerodynamic forces (drag, lift, etc.), heat transfer between fluids and solids, mixing of a pollutant, *etc.*;
- ▷ very often: variations of these quantities with parameters (velocities, temperature differences, shape parameters), in order to optimize a system;
- ▷ but also: separation location of boundary layers, pressure field at the wall, flow structure, shocks location, acoustic sources, *etc.*;
- ▷ and more and more: response to a control strategy (blowing, suction, MEMS, *etc.*).

⇒ Turbulence modeling, is the science that consists in building these models, for the purpose of their use by engineers.

This science is seeking an impossible goal: provide a simple, cheap, robust model, for all situations.

2. Books

✓ Turbulence and modeling:

- ↪ Chassaing, P. – *Turbulence en mécanique des fluides. Analyse du phénomène en vue de sa modélisation à l'usage de l'ingénieur* – Toulouse, France, Cépaduès-Éditions, 2000, *Collection Polytech*.
- ↪ Pope, S. – *Turbulent Flows* – Cambridge University Press, Cambridge, UK, 2000.
- ↪ Davidson, P. A. – *Turbulence: An Introduction for Scientists and Engineers* – OUP Oxford, 2004.
- ↪ Durbin, P. A. and Pettersson Reif, B. A. – *Statistical Theory and Modeling for Turbulent Flows* – John Wiley & Sons, Ltd, Chichester, UK, 2001.
- ↪ Hanjalić, K. and Launder, B. – *Modelling Turbulence in Engineering and the Environment* – Cambridge University Press
- ↪ Deville, M.O. and Gatski, T.B. – *Mathematical Modeling for Complex Fluids and Flows* – Springer, 2012.
- ↪ Bailly, Ch., Comte-Bellot, G. – *Turbulence* – CNRS éditions, Paris, 2003.
- ↪ Viollet, P.-L. , Chabard, J.-P. , Esposito, P. and Laurence, D. – *Mécanique des fluides appliquée* – Presses de l'École nationale des ponts et chaussées, Paris, 1998.
- ↪ Sagaut, P., Deck, S. and Terracol, M. – *Multiscale and multiresolution approaches in turbulence*. Imperial College Press, London, 2006.

✓ Numerical methods:

- ↪ Ferziger, J. H. and Perić, M. – *Computational methods for fluid dynamics* – Springer, 1996.

3. Reminder: the Navier–Stokes equations

- ✓ Notation: in RANS modeling, instantaneous values are generally denoted by f^* , mean values by F (Reynolds average) and fluctuating values by f

$$\Rightarrow f^* = F + f$$

- ✓ Einstein's notation is used: implicit summation over indices repeated in a term.

In simpler words:

$$\frac{\partial \phi_i}{\partial t} + u_k \frac{\partial \phi_i}{\partial x_k} = 0$$

is a compact expression standing for

$$\frac{\partial \phi_1}{\partial t} + u_1 \frac{\partial \phi_1}{\partial x_1} + u_2 \frac{\partial \phi_1}{\partial x_2} + u_3 \frac{\partial \phi_1}{\partial x_3} = 0$$

$$\frac{\partial \phi_2}{\partial t} + u_1 \frac{\partial \phi_2}{\partial x_1} + u_2 \frac{\partial \phi_2}{\partial x_2} + u_3 \frac{\partial \phi_2}{\partial x_3} = 0$$

$$\frac{\partial \phi_3}{\partial t} + u_1 \frac{\partial \phi_3}{\partial x_1} + u_2 \frac{\partial \phi_3}{\partial x_2} + u_3 \frac{\partial \phi_3}{\partial x_3} = 0$$

There is no summation over i since it is not repeated *in a single term*.

This two-term equation actually represents 3 equations containing 4 terms. It is compact!

Be careful: $\frac{\partial^2 \phi}{\partial x_k^2}$ (no repetition) is different from $\frac{\partial^2 \phi}{\partial x_k \partial x_k}$

- ✓ Although the basic equations used in fluid mechanics are well known, it is worth recalling them here since the way they are derived is very similar to what will be made below in turbulence modeling. The derivation can be summarized in three steps:
- ~> writing of the conservation equations;
 - ~> closure of the system using constitutive relations, which introduce *state variables* that describe the macroscopic state of the fluid at a given point and a given time;
 - ~> writing of equations of state, which describe the evolution of these state variables.

Conservation equations:

$$\begin{aligned} \checkmark \text{ Mass conservation (continuity): } & \frac{\partial \rho^*}{\partial t} + \frac{\partial \rho^* u_i^*}{\partial x_i} = 0 \\ \checkmark \text{ Momentum conservation: } & \frac{\partial \rho^* u_i^*}{\partial t} + \frac{\partial \rho^* u_i^* u_j^*}{\partial x_j} = \frac{\partial \sigma_{ij}^*}{\partial x_j} + \rho^* g_i \\ \checkmark \text{ Energy conservation: } & \frac{\partial \rho^* e^*}{\partial t} + \frac{\partial \rho^* u_i^* e^*}{\partial x_i} = \sigma_{ij}^* s_{ij}^* - \frac{\partial}{\partial x_i} \gamma_i \end{aligned}$$

✓ These equations describe the behavior of the fluid at macroscopic scale, whatever the type of fluid. However, they involve 14 unknowns:

↪ density ρ^* ;

↪ the three components of velocity u_i^* ;

↪ the six independent components of the stress tensor σ_{ij}^* (symmetric tensor) ;

↪ internal energy e^* ;

↪ and the three components of the heat flux γ_i ;

for 5 equations only.

✓ In order to close the system, constitutive equations must be introduced, which describe the properties of the fluid (and thus are dependent on the type of fluid).

Constitutive relations:

A fluid is *Newtonian* if it can be described by the following linear laws giving:

- ✓ the stress tensor σ_{ij}^* as a function of the strain tensor s_{ij}^* (Newton's law):

$$\sigma_{ij}^* = \left(-p^* + \lambda \frac{\partial u_l^*}{\partial x_l} \right) \delta_{ij} + 2\mu s_{ij}^* \quad (1)$$

where $s_{ij}^* = \frac{1}{2} \left(\frac{\partial u_i^*}{\partial x_j} + \frac{\partial u_j^*}{\partial x_i} \right)$, and δ_{ij} is the identity tensor (Kronecker's symbol),

- ✓ and the heat flux γ_i as a function of the temperature gradient (Fourier's law):

$$\gamma_i = -k \frac{\partial T^*}{\partial x_i} \quad (2)$$

- ✓ We have provided the 9 equations which were missing.
 - ✓ The properties of the fluid (dynamic viscosity μ and volume viscosity λ , conductivity k) are known.
 - ✓ But two state variables (pressure p^* and temperature T^*) have been introduced which describe the state of the fluid at a given point and a given time (i.e., they describe at the macroscopic scale the properties of the molecular agitation).
- ⇒ We need additional relations, *equations of state* to obtain the evolution of these state variables.

Equations of state:

- ✓ For a perfect gas, we can write:

$$p^* = \rho^* r T^* \quad (3)$$

and

$$de^* = C_v dT^* \quad (4)$$

- ✓ The first relation is replaced by the incompressibility constraint

$$\frac{\partial u_i^*}{\partial x_i} = 0 \quad (5)$$

in the case of an incompressible flow (thus introducing a famous paradox: the equation of state does not involve pressure).

- ✓ The solutions of this system are very complex, and contain all the physical phenomena observed in a Newtonian fluid flow: vortices (in particular turbulence), shocks, acoustic waves, transformation of kinetic energy into internal heat, *etc.*
- ✓ This system can sometimes be simplified, but one must first answer these questions:
 - Which assumptions make these simplifications possible?
 - Is it sufficient to address the problem?
- ✓ Example: it is often assumed that μ , λ , k and C_v are independent of temperature.
Warning: this is not valid for large temperature differences.
- ✓ Example: incompressible flow ($\operatorname{div} u_i^* = 0$).
Incompressible fluids do not exist! However, assuming that the effects of compressibility are negligible is justified in numerous cases (low Mach numbers).
Warning: this excludes acoustic waves and shocks from the solution.

✓ **Low-Mach-number approximation:**

↪ For low Mach numbers, it can be assumed that density does not depend on pressure, but only on temperature

$$\rho^* = f(\cancel{P^*}, T^*)$$

↪ For a perfect gas, we have

$$p^* = \rho^* r T^*$$

such that

$$d\rho^* = \frac{\partial \rho^*}{\partial \cancel{P^*}} \bigg|_{T^*} dP^* + \frac{\partial \rho^*}{\partial T^*} \bigg|_{P^*} dT^* = -\frac{\rho^*}{T^*} dT^*$$

and

$$\rho^* = \rho_0^* \frac{T_0^*}{T^*}$$

↪ The flow is thus considered incompressible, but density varies as a function of the inverse of the temperature: the fluid is dilatable.

↪ The equations of motion can be derived using asymptotic expansions at the limit of small Mach numbers.

- ✓ A very common approximation, when the temperature differences are weak, is the **Boussinesq approximation**: the density variations can be neglected in the Navier–Stokes equations ($\rho^* = \rho_0$), except in the buoyancy term $\rho^* g_i$ (buoyant force).
- ✓ Usual additional simplifications:
 - $\leadsto \mu, k$ and C_v are assumed to be independent of temperature
 - \leadsto A Taylor series expansion at first order is made of the density around the reference temperature T_0^* :

$$\rho^* = \rho_0^* - \beta \rho_0^* (T^* - T_0^*)$$

$$\text{with } \beta = -\frac{1}{\rho_0^*} \frac{\partial \rho^*}{\partial T} \text{ (Isobar compressibility coefficient)}$$

such that the buoyancy term reads: $\rho^* g_i = \rho_0^* g_i - \rho_0^* g_i \beta (T^* - T_0^*)$

- ✓ When the temperature variations are weak and the buoyancy effects are negligible (forced convection), the density ρ^* can be considered as a constant (simply denoted by ρ).

The Navier–Stokes equation then reduce to

$$\frac{\partial u_i^*}{\partial x_i} = 0 \quad (6)$$

$$\frac{\partial u_i^*}{\partial t} + u_j^* \frac{\partial u_i^*}{\partial x_j} = -\frac{1}{\rho} \frac{\partial p^*}{\partial x_i} + g_i + \nu \frac{\partial^2 u_i^*}{\partial x_j \partial x_j} \quad (7)$$

$$\frac{\partial T^*}{\partial t} + u_i^* \frac{\partial T^*}{\partial x_i} = \alpha \frac{\partial^2 T^*}{\partial x_i \partial x_i} + \frac{2\mu}{\rho C_v} s_{ij}^* s_{ij}^* \quad (8)$$

✓ Remarks:

- ↪ the volume viscosity λ has disappeared;
- ↪ the kinematic viscosity $\nu = \mu/\rho$ and the thermal diffusivity $\alpha = k/\rho C_v$ naturally appear;
- ↪ the dynamics does not depend on the thermal field \Rightarrow the dynamics can be resolved, Eqs. (6) et (7), and then the velocity field used for solving the temperature equation \Rightarrow the temperature is then a *passive scalar* ;
- ↪ if the flow is isothermal ($T^* = \text{constant}$), equation (8) disappears.

✓ If the gravitational field is oriented along $-z$:

$$g_i = \begin{bmatrix} 0 \\ 0 \\ -g \end{bmatrix}$$

one can define: $p^{**} = p^* + \rho g z$

$$\Rightarrow -\frac{1}{\rho} \frac{\partial p^*}{\partial x_i} + g_i = -\frac{1}{\rho} \frac{\partial p^{**}}{\partial x_i}$$

✓ It is seen that **when the buoyancy forces can be neglected**, the gravity has no influence on the velocity field: gravity only influences the pressure.

✓ Most of the time, in CFD codes, when the buoyancy forces can be neglected, p^{**} is used, and then g_i disappears from the equations.

Warning: this does not mean that the gravity is neglected, but rather that the hydrostatic pressure is included in p^{**} .

\Rightarrow The computation will not provide the pressure, but p^{**} !

p^* can be obtained from $p^{**} - \rho g z$.

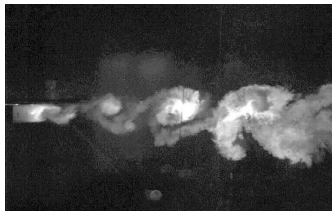
3.1. The Reynolds decomposition

- ✓ In RANS modeling, the first modeling consist in assuming that the instantaneous quantities are random variables:
 - ⇒ a random behavior of the flow is assumed, while the Navier–Stokes equations are obviously deterministic.

This is the chaotic character of the system that makes this assumption possible.

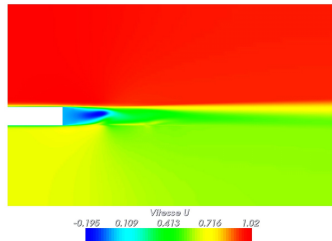
⇒ The deterministic character of turbulence is not accounted for:
in particular, the presence of coherent structures (large-scale eddies, with, very often, a pseudo-periodic character) can play a significant role in the flow (see the figure below).

⇒ The flow is thus presumed fully turbulent: the problem of transition to turbulence is outside of the scope of the present course.



Mixing layer downstream a thick plate

Experiments by Perret and Delville (Institute Pprime, university of Poitiers/CNRS/ENSMA)



RANS solution ($k-\varepsilon$ low-Reynolds number model).

- ✓ The flow is decomposed into mean and fluctuating parts by using the Reynolds average:

in RANS modeling, this average is denoted by $\overline{\quad}$.

Mean velocities: $U_i = \overline{u_i^*}$

Fluctuating velocities: $u_i = u_i^* - U_i$

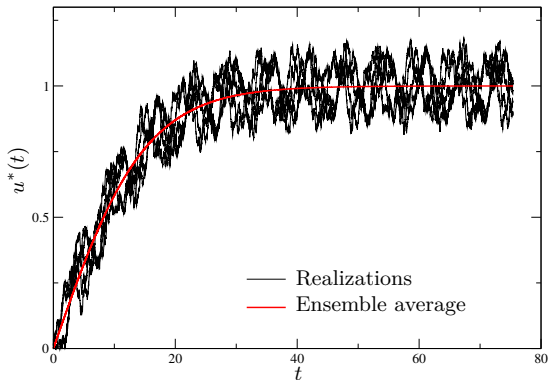
Mean pressure: $P = \overline{p^{**}}$ (the hydrostatic pressure is thus included)

Fluctuating pressure: $p = p^{**} - P$

- ✓ The Reynolds average of the turbulent (random) variable f^* is its statistical average, equivalent to the ensemble average (law of large numbers), defined by:

$$\overline{f^*}(\mathbf{x}, t) = \lim_{N \rightarrow \infty} \left(\frac{1}{N} \sum_{n=1}^N f_n^*(\mathbf{x}, t) \right) \quad (9)$$

i.e., evaluated at each point and each time by performing a large number of experiments.



3.2. The closure problem

- ✓ The introduction of the Reynolds decomposition leads to the averaged Navier–Stokes equations:

$$\frac{\partial U_i}{\partial x_i} = 0 \quad (10)$$

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial P}{\partial x_i} + \nu \frac{\partial^2 U_i}{\partial x_j \partial x_j} - \frac{\partial \overline{u_i u_j}}{\partial x_j} \quad (11)$$

- ✓ There are 4 equations for 10 unknowns: $P, U, V, W, \overline{u^2}, \overline{v^2}, \overline{w^2}, \overline{uv}, \overline{uw}, \overline{vw}$.
 \Rightarrow the system is open.

- ✓ Writing the transport equations for the Reynolds stress tensor $\overline{u_i u_j}$:

$$\begin{aligned}
 \frac{\partial \overline{u_i u_j}}{\partial t} + \underbrace{U_k \frac{\partial \overline{u_i u_j}}{\partial x_k}}_{C_{ij}} &= \underbrace{\nu \frac{\partial^2 \overline{u_i u_j}}{\partial x_k \partial x_k}}_{D_{ij}^\nu} - \underbrace{\frac{\partial \overline{u_i u_j u_k}}{\partial x_k}}_{D_{ij}^T} - \underbrace{\frac{1}{\rho} \overline{u_i \frac{\partial p}{\partial x_j}} - \frac{1}{\rho} \overline{u_j \frac{\partial p}{\partial x_i}}}_{\phi_{ij}^*} \\
 &\quad - \underbrace{\overline{u_i u_k} \frac{\partial U_j}{\partial x_k} - \overline{u_j u_k} \frac{\partial U_i}{\partial x_k}}_{P_{ij}} - \underbrace{2\nu \frac{\partial \overline{u_i}}{\partial x_k} \frac{\partial \overline{u_j}}{\partial x_k}}_{\varepsilon_{ij}} \quad (12)
 \end{aligned}$$

leads to 6 new equations...

... but unfortunately to 34 new unknowns: $\overline{u_i u_j u_k}$, $\overline{u_i \frac{\partial p}{\partial x_j}}$, $\frac{\partial \overline{u_i}}{\partial x_k} \frac{\partial \overline{u_j}}{\partial x_k}$

\Rightarrow the problem is still open.

- ✓ One could write equations for these unknowns, but the process always produces more unknowns than equations.
- ✓ One must decide when to stop.

✓ **First moment closure (or eddy-viscosity modeling):**

↪ The 4 equations (10) and (11) are solved.

⇒ the first moments are obtained: U, V, W, P

↪ It is necessary to “invent” a relation between the variables that are not resolved as functions of the variables that are resolved:

this relation providing the second moments (the $\overline{u_i u_j}$'s) as functions of the first moments is called a *first order model* or, more widely, an *eddy-viscosity model*.

↪ This is the equivalent of a *constitutive relation* for a material: for instance, for a Newtonian, incompressible fluid, the stress tensor is related to the strain tensor by

$$\sigma_{ij}^* = -p^* \delta_{ij} + 2\mu s_{ij}^*$$

This is a model for the fluid, the model of Newton. This model closes of the system (the Navier–Stokes equations).

∼→ An eddy-viscosity model is the equivalent of the model of Newton for the “turbulent fluid”: this is a relation between the Reynolds stress tensor (which exactly play the role of a stress in (11) to the mean pressure and the mean strain tensor.

Moreover, the simplest models are based on the Boussinesq relation, similar to the Newton relation:

$$-\rho \overline{u_i u_j} = -\frac{2}{3} \rho k \delta_{ij} + 2\mu_t S_{ij} \quad \text{where } S_{ij} = \frac{1}{2} \left(\frac{\partial U_i}{\partial x_j} + \frac{\partial U_j}{\partial x_i} \right)$$

where $k = \frac{1}{2} \overline{u_i u_i}$ is the turbulent energy.

✓ **Second moment closure, SMC (or Reynolds stress models, RSM):**

~> The 4 equations (10) and (11) are solved.

⇒ to obtain the first moments: U, V, W, P

~> The 6 transport equations for the Reynolds stress tensor (12) are also solved

⇒ to obtain the second moments: $\overline{u_i u_j}$

~> It is necessary to “invent” relations giving the variables that are not solved as functions of the variables that are solved:

such relations giving the unknowns $\left(\overline{u_i u_j u_k}, \overline{u_i \frac{\partial p}{\partial x_j}}, \overline{\frac{\partial u_i}{\partial x_k} \frac{\partial u_j}{\partial x_k}} \right)$ as functions of the first and second moments form a *second moment closure*.

~> These are also equivalent to *constitutive relations*, but very complex ones.

Example: the Daly–Harlow model for the triple correlations is

$$\overline{u_i u_j u_k} = -C_s \frac{k}{\varepsilon} \overline{u_k u_l} \frac{\partial \overline{u_i u_j}}{\partial x_l}$$

- ↪ A Reynolds stress model solves the transport equations for the Reynolds stresses, while eddy-viscosity models evaluate them from the mean velocities and pressure by a simple algebraic relation: Reynolds stress models contain obviously “more physics”.
- ↪ In particular, the production terms does not require modeling, since they only involve moments of order less or equal to two:

$$P_{ij} = - \overline{u_i u_k} \frac{\partial U_j}{\partial x_k} - \overline{u_j u_k} \frac{\partial U_i}{\partial x_k}$$

This terms have a fundamental influence.

✓ The question everybody wants to ask: what modeling level do we need?

This question does unfortunately not have a simple answer, for different reasons:

- ~> It depends on the effort one is ready to put in the study: Reynolds stress models are not much more expansive in terms of CPU cost, but slightly more difficult to correctly handle.
- ~> It depends on the objectives: prediction of global quantities (drag, lift, global heat transfer) or of more precise details (separation location, flow structure, turbulent scales, anisotropies, *etc.*).

✓ But it is not sufficient: everything is dependent on the type of flow!

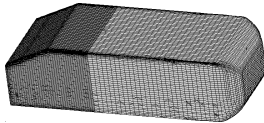
~> For attached flows (airfoil at low angle of attack), a simple model can be sufficient to obtain the drag (mixing length model).

~> For more complex flows, a more sophisticated model can be necessary to obtain the drag!

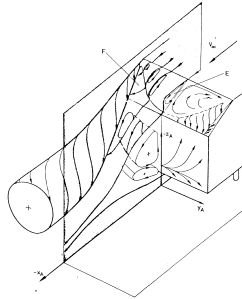
~> As a general rule, the understanding of the physical mechanisms that play a role in a flow and a good knowledge of the potential of the models to reproduce these mechanisms is necessary to find the relevant adequacy.

~> A “universal” model, that is able to predict everything in any situation *does not exist*.

✓ Example: the Ahmed body (simplified car)



From Manceau (2003)



From Ahmed *et al.* (1984)

~ The flow structure is very complex.

~ The pressure at the surface of the body is very dependent on the separated regions.

⇒ A very refined model is necessary, even to simply obtain the drag (actually, nowadays, none of the available RANS models give satisfactory results!)

3.3. A quick historical review of the different types of models

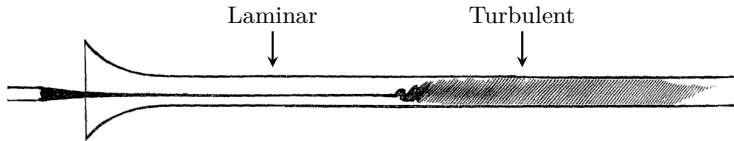
3.3.1. Before the advent of computers: the pioneers

1877 Boussinesq invents the concept of eddy viscosity.

- ~ Simple phenomenological reasoning: the “turbulent material” behaves like a Newtonian fluid, i.e., the mean stress is proportional to the mean strain
- ~ This is the first **turbulence model**.
- ~ The notion of *eddy-viscosity* is introduced.

✓ What is the reasoning of Boussinesq? By observing river flows, he understood that:

~> Turbulence strongly enhances mixing



Reynolds (1883) experiment: flow in a pipe, transport of dye

~> Turbulence is at the origin of a resistance to the flow stronger than in a laminar flow

✓ Modern interpretation: Navier–Stokes equations

$$\frac{\partial u_i^*}{\partial t} + u_j^* \frac{\partial u_i^*}{\partial x_j} = \frac{1}{\rho} \frac{\partial \sigma_{ij}^*}{\partial x_j} = -\frac{1}{\rho} \frac{\partial p^*}{\partial x_i} + \nu \frac{\partial^2 u_i^*}{\partial x_j \partial x_j}$$
$$\frac{\partial \rho C_v T^*}{\partial t} + u_i^* \frac{\partial \rho C_v T^*}{\partial x_i} = \frac{\partial \gamma_i}{\partial x_i} + \sigma_{ij}^* s_{ij}^* = \alpha \frac{\partial^2 \rho C_v T^*}{\partial x_i \partial x_i} + 2\rho \nu s_{ij}^* s_{ij}^*$$

Modelling the stress tensor σ_{ij}^* introducing a molecular viscosity leads to the appearance of two effects:

↪ A diffusion in the momentum equation $\nu \frac{\partial^2 u_i^*}{\partial x_j \partial x_j}$

↪ A dissipation of energy into heat $2\rho \nu s_{ij}^* s_{ij}^*$

✓ The idea of Boussinesq can be expressed as:

~> The enhancement of mixing by turbulence can be modelled as a mean diffusion effect

~> The resistance to the flow can be modelled as a mean dissipation effect

⇒ The flow behaves as if a strong additional viscosity was introduced

✓ Mean momentum equation

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial P}{\partial x_i} + \nu \frac{\partial^2 U_i}{\partial x_j \partial x_j} - \frac{\partial \overline{u_i u_j}}{\partial x_j}$$

$\leadsto -\rho \overline{u_i u_j}$ plays exactly the same role as σ_{ij}^* \Rightarrow it is denoted as the *Reynolds stress tensor*

\leadsto It is thus modelled by introducing a turbulent viscosity: ν is replaced by $\nu + \nu_t$
where $\nu_t \gg \nu$

$$\nu \frac{\partial^2 U_i}{\partial x_j \partial x_j} - \frac{\partial \overline{u_i u_j}}{\partial x_j} = \nu \frac{\partial^2 U_i}{\partial x_j \partial x_j} + \frac{\partial}{\partial x_j} \left(\nu_t \frac{\partial U_i}{\partial x_j} \right) = \frac{\partial}{\partial x_j} \left((\nu + \nu_t) \frac{\partial U_i}{\partial x_j} \right)$$

∼ We know nowadays that the model writes

$$\overline{u_i u_j} = -2\nu_t S_{ij} + \frac{2}{3}k\delta_{ij} \quad \text{where } S_{ij} = \frac{1}{2} \left(\frac{\partial U_i}{\partial x_j} + \frac{\partial U_j}{\partial x_i} \right)$$

This is the famous Boussinesq relation

∼ Note that the half-trace of the tensor is correct:

$$\frac{1}{2}\overline{u_i u_i} = -\nu_t S_{ii} + \frac{1}{3}k\delta_{ii} = k \quad \text{since } S_{ii} = 0$$

where $k = \frac{1}{2}\overline{u_i u_i}$ is the turbulent energy

↪ The introduction of the turbulent viscosity indeed leads to:

- ▷ A strong diffusion in the mean momentum equation: $\frac{\partial}{\partial x_j} \left((\nu + \nu_t) \frac{\partial U_i}{\partial x_j} \right)$
- ▷ A strong dissipation of the mean mechanical energy: $2\rho(\nu + \nu_t)S_{ij}S_{ij}$ (energy is not directly dissipated into heat, but first transformed into turbulent energy, as will be seen later)

- ~> Comparing this relation with the constitutive relation for Newtonian fluids, it can be seen how this model is similar to the Newton model.
- ~> The eddy-viscosity and the turbulent energy play roles equivalent to those of the molecular viscosity and the pressure, respectively.
- ~> However, this model does not provide a relation to evaluate ν_t and k , which strongly vary in the flow and from one flow to another.
- ~> This type of models is called an **algebraic model**, or a **zero-equation model**.
n-equation model = model involving n additional differential equations
(compared to the averaged Navier–Stokes equations, equations (10) and (11)).

Digression:

The introduction of the Boussinesq relation in the mean velocity equations

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial P}{\partial x_i} + \nu \frac{\partial^2 U_i}{\partial x_j \partial x_j} - \frac{\partial \overline{u_i u_j}}{\partial x_j}$$

leads to

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial}{\partial x_i} \left[P + \frac{2}{3} \rho k \right] + \frac{\partial}{\partial x_j} \left[(\nu + \nu_t) \left(\frac{\partial U_i}{\partial x_j} + \frac{\partial U_j}{\partial x_i} \right) \right]$$

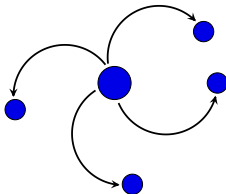
Defining: $P^* = P + \frac{2}{3} \rho k$

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial P^*}{\partial x_i} + \frac{\partial}{\partial x_j} \left[(\nu + \nu_t) \left(\frac{\partial U_i}{\partial x_j} + \frac{\partial U_j}{\partial x_i} \right) \right] \quad (13)$$

\Rightarrow It is seen that, similar to gravity in the Navier–Stokes equations, the turbulent energy in the averaged equations do not modify the velocities, but only the pressure \Rightarrow **the knowledge of ν_t is sufficient to evaluate the mean velocities.**

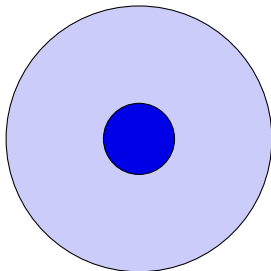
Warning: in many CFD codes, it is indeed equation (13) that is solved
 $\Rightarrow P$ is not provided by the computation, but rather P^* .

- ✓ The mixing is mainly due to the large eddies (integral scale) $\Rightarrow \nu_t$ is linked to their characteristics
- ✓ Indeed, if we consider, in order to simplify, the mixing of a passive scalar (temperature, concentration, *etc.*):
 - \leadsto For instance, at some time, let us introduce a drop of dye (instantaneous concentration= ϕ^*) in a turbulent flow with zero mean velocity



- \leadsto Turbulent mixing is due to the advection of fluid particles by the fluctuating velocity (the eddies)
- \leadsto The dye is mixed by convection at the instantaneous level of description.

- ✓ What is observed on average? (i.e., repeating the same experiment a large number of times)



- ✓ The dye spot gradually spreads and the concentration decreases, due to mixing.
- ⇒ Everything, on average, is similar to a standard diffusion process.

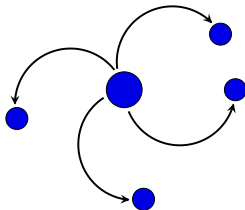
⇒ Modelling the average effect of turbulent mixing by an additional diffusion (turbulent diffusion) is legitimate.

↪ This averaged effect of convection by turbulent agitation (turbulent transport) is then called *turbulent diffusion*.

↪ Again, the similarity with continuum mechanics is clear: what is called *molecular diffusion* is the effect, at macroscopic scale, of the mixing due to molecular agitation.

- ✓ The diffusive model must be calibrated in order to reproduce the average effect of turbulent mixing.

→ Equation of pure convection: $\frac{d\phi^*}{dt} = \frac{\partial\phi^*}{\partial t} + u_k^* \frac{\partial\phi^*}{\partial x_k} = 0$



- Let us denote τ the time necessary for the large-scale eddies, of size ℓ , to transport the dye over the distance ℓ .

→ Order of magnitudes: $\frac{\phi^*}{\tau} \simeq u \frac{\phi^*}{\ell} \quad \Rightarrow \quad \tau \simeq \frac{\ell}{u}$

where u and ℓ are the integral scales (velocity and length), i.e., the characteristic scales of the largest eddies.

↪ Decomposition: $\phi^* = \Phi + \phi$ and $u_k^* = u_k$ (mean velocity = 0)

$$\leadsto \frac{\partial \phi^*}{\partial t} + u_k^* \frac{\partial \phi^*}{\partial x_k} = \frac{\partial \Phi}{\partial t} + \frac{\partial \phi}{\partial t} + u_k \frac{\partial \Phi}{\partial x_k} + u_k \frac{\partial \phi}{\partial x_k} = 0$$

↪ On average:

$$\triangleright \overline{\frac{\partial \Phi}{\partial t} + \frac{\partial \phi}{\partial t} + u_k \frac{\partial \Phi}{\partial x_k} + u_k \frac{\partial \phi}{\partial x_k}} = 0$$

$$\triangleright \overline{\frac{\partial \Phi}{\partial t}} + \overline{\frac{\partial \phi}{\partial t}} + \overline{u_k \frac{\partial \Phi}{\partial x_k}} + \overline{u_k \frac{\partial \phi}{\partial x_k}} = 0 \text{ (linearity)}$$

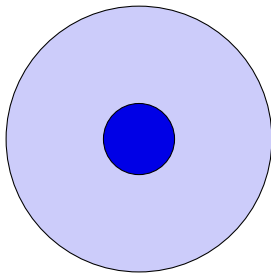
$$\triangleright \overline{\frac{\partial \Phi}{\partial t}} + \overline{\frac{\partial \phi}{\partial t}} + \overline{\frac{\partial u_k \Phi}{\partial x_k}} + \overline{\frac{\partial u_k \phi}{\partial x_k}} = 0$$

because $\frac{\partial u_k \phi}{\partial x_k} = u_k \frac{\partial \phi}{\partial x_k} + \phi \frac{\partial u_k}{\partial x_k}$ and $\frac{\partial u_k}{\partial x_k} = 0$ (incompressibility)

$$\triangleright \overline{\frac{\partial \Phi}{\partial t}} + \overline{\frac{\partial \phi}{\partial t}} + \overline{\frac{\partial u_k \Phi}{\partial x_k}} + \overline{\frac{\partial u_k \phi}{\partial x_k}} = 0 \text{ (commutation)}$$

$$\triangleright \boxed{\overline{\frac{\partial \Phi}{\partial t}} + \overline{\frac{\partial u_k \phi}{\partial x_k}}} = 0 \text{ because } \overline{\Phi} = \Phi, \overline{\phi} = 0, \overline{u_k \Phi} = \overline{u_k} \Phi = 0$$

- ↪ It is seen that the averaged effect of turbulent convection is reflected by the presence of the second moment $\overline{u_k \phi}$.
- ↪ Since the eddies are not solved, we try to represent the *mean* mixing effect by a diffusion:



- ↪ On average, we want the diffusion model to spread the dye spot over the same distance ℓ during the same time τ as the turbulent mixing does.

~> Equation of pure diffusion:

$$\triangleright \frac{\partial \Phi}{\partial t} = \frac{\partial}{\partial x_k} \left(\alpha_t \frac{\partial \Phi}{\partial x_k} \right)$$

$$\triangleright \text{Order of magnitudes: } \frac{\Phi}{\tau} \simeq \alpha_t \frac{\Phi}{\ell^2} \Rightarrow \tau \simeq \frac{\ell^2}{\alpha_t}$$

\Rightarrow It can be seen that, for the diffusion model to correctly represent the mean mixing effect, we must have

$$\alpha_t \simeq \ell u$$

~> Remark: in this case of a *passive* scalar, the turbulent diffusivity only depends on the characteristics of the turbulent velocity field.

✓ Applying these arguments to the momentum (per unit volume) ρu_i^* rather than the scalar ϕ^* , the same conclusions are drawn:

↪ The mixing of momentum is due to the mean effect of the convection of momentum by the fluctuating velocity:

$$\overline{u_j \frac{\partial \rho u_i}{\partial x_j}} = \frac{\partial \overline{\rho u_i u_j}}{\partial x_j}$$

↪ It is the term involved in the mean momentum equation (Reynolds-stress tensor).

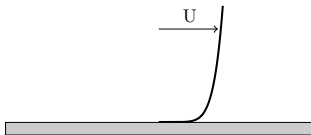
↪ We model this mean effect of fluctuating convection by a diffusion:

$$\frac{\partial}{\partial x_k} \left(\nu_t \frac{\partial U_i}{\partial x_k} \right)$$

↪ The turbulent viscosity must also satisfy: $\nu_t \simeq \ell u$

1925 Prandtl introduces the concept of mixing length to evaluate ν_t .

- ✓ We have seen that $\nu_t \simeq \ell u$, but how to evaluate ℓ and u ?
- ✓ In a boundary layer above a flat plate (idealized model for the boundary layer above an airfoil), one wants to reproduce the streamwise velocity U .



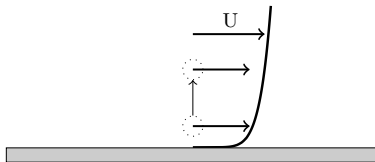
- ✓ Since variations along x are very slow compared to those along y , the only significant component of the Reynolds stress in the U -equation is \overline{uv} . Keeping the dominant terms only, the U -equation reduces to

$$U \frac{\partial U}{\partial x} + V \frac{\partial U}{\partial y} = -\frac{1}{\rho} \frac{\partial P}{\partial x} + \nu \frac{\partial^2 U}{\partial y^2} - \frac{\partial \overline{uv}}{\partial y}$$

- ✓ Boussinesq's model yields $\overline{uv} = -\nu_t \frac{\partial U}{\partial y}$. How can one evaluate ν_t ?

✓ Displaced particle argument:

→ Let us imagine a fluid particle located at y above the wall, that follows the mean flow ($u^* = U(y)$)



→ We move it vertically upwards: in other words, we give it a velocity $v^* > 0$. Its vertical fluctuating velocity (difference between the instantaneous velocity and the mean velocity) is then $v = v^* - V = v^* > 0$

→ It is assumed that it remains unchanged for a while and then is mixed with the ambient fluid after some distance ℓ_m called the *mixing length*.

In other words, its momentum is preserved, and after a distance ℓ_m , the particle exchanges its momentum with the ambient fluid.

↪ Just before the moment when it is mixed, its instantaneous velocity is then unchanged: $u^* = U(y)$

↪ Its fluctuating velocity (with respect to the local mean velocity) is then:

$$u = u^* - U(y + \ell_m) = U(y) - U(y + \ell_m) \simeq -\ell_m \frac{\partial U}{\partial y} < 0$$

↪ u is negative when the particle moves upwards ($v > 0$), because $\frac{\partial U}{\partial y} > 0$. It can be easily seen that $u > 0$ when it moves downwards ($v < 0$). Therefore, $uv < 0$, and, on average, $\overline{uv} < 0$ (more accurately, \overline{uv} is always of a sign opposite to that of $\frac{\partial U}{\partial y}$).

↪ We have seen above that the turbulent viscosity is $\nu_t = \ell u$, where ℓ is the size of the largest eddies. It can be observed experimentally that $\ell_m \simeq \ell \simeq \kappa y$, where κ is called Karman's constant. Therefore

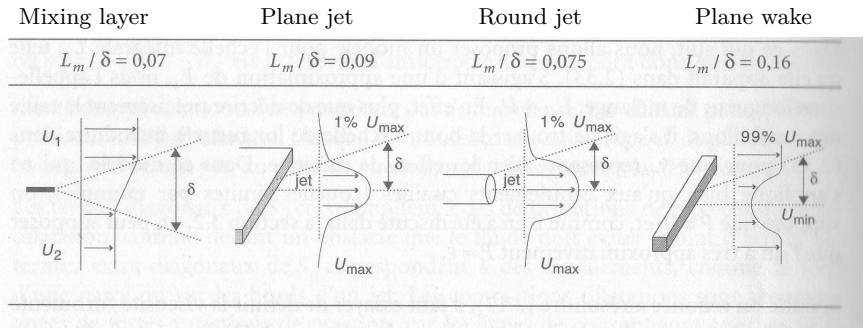
$$\boxed{\nu_t = \ell_m^2 \left| \frac{\partial U}{\partial y} \right|} \quad \Rightarrow \quad \boxed{\overline{uv} = -\ell_m^2 \left| \frac{\partial U}{\partial y} \right| \frac{\partial U}{\partial y}} \quad (14)$$

↪ These arguments are similar to those of the kinetic theory of gases. The mixing length is analog to the mean free path.

- ✓ To be valid whatever the orientation of the shear, this relation must be generalized as

$$\nu_t = \ell_m^2 \sqrt{2S_{ij}S_{ij}}$$

- ✓ This idea has proven very useful not only for boundary layers but also for free shear flows (jets, mixing layers, wakes), for which the mixing length can be related to the thickness of the shear layer.



From Viollet *et al.* (1998)

- ✓ This is also a **first moment, algebraic model (zero equation)**.
- ✓ This approximation is very useful as a “rule of thumb”.
- ✓ Under more elaborate forms, it is still used in aeronautics for *attached* flows (see below Cebeci and Smith, Baldwin and Lomax).

1942 Independently, Kolmogorov and Prandtl proposed to relate the turbulent
1945 velocity scale to the turbulent kinetic energy:

$$u \propto \sqrt{k}$$

\leadsto Introducing a coefficient C_μ , we have

$$\nu_t = C_\mu \sqrt{k} \ell$$

relation known as the *Prandtl–Kolmogorov relation*.

- ↪ Two equations of state are thus necessary to evaluate these variables.
- ↪ Prandtl proposes to solve a transport equation for k , based on the exact equation, in which the unknown terms are modeled.
- ⇒ This is also an eddy-viscosity model. But this is the first **1-equation model**. It is still necessary to prescribe the length scale l .
- ↪ Kolmogorov additionally proposes to evaluate the length scale by

$$\ell = \frac{\sqrt{k}}{\omega}$$

where ω is a characteristic frequency for the large eddies.

He proposes to solve a transport equation for ω

⇒ this is still an eddy-viscosity model. But this is the first **2-equation model**.

It is based on the scales k and ω .

1945 Chou writes the transport equations for the Reynolds stresses and shows that the pressure term can be decomposed into a rapid and a slow part

$$\phi_{ij}^* = -\frac{1}{\rho} \overline{u_i \frac{\partial p}{\partial x_j}} - \frac{1}{\rho} \overline{u_j \frac{\partial p}{\partial x_i}} = \underbrace{B_{ij}}_{\phi_{ij}^1} + \underbrace{(A_{ijml} + A_{jiml}) \frac{\partial U_l}{\partial x_m}}_{\phi_{ij}^2}$$

\Rightarrow this is the first step towards Reynolds stress models.

1951 Rotta proposes the first model of the Reynolds-stress transport equations.

~> In particular, he proposes the following model for the slow term, using a simple “return to isotropy” model:

$$\phi_{ij}^l = -C_1 \varepsilon a_{ij}$$

where $a_{ij} = \frac{\overline{u_i u_j}}{k} - \frac{2}{3} \delta_{ij}$ is the anisotropy tensor.

~> Based on the physical role of this term observed in the return-to-isotropy experiment of an initially anisotropic turbulence:

When the strain disappears, experimental observations show that the turbulence returns to an isotropic state ($\overline{u_i u_j} = \frac{2}{3} k \delta_{ij}$, i.e., $a_{ij} = 0$), under the effect of the only non-zero source terms (slow term and dissipation term).

3.3.2. After the advent of computers: the heroic age of CFD

1967 Cebeci and Smith

- ↪ Algebraic model (of mixing length type), improved later on by Baldwin and Lomax (1978)
- ↪ The variation of the mixing length with the distance to the wall is prescribed.
- ↪ These models have been and are still used intensively in aeronautics. They are not valid for separated flows.

1972 Jones and Launder

- ↪ Using previous work of Davidov (1961), Harlow and Nakayama (1967) and Hanjalić (1970): two-equation model using ε as the second variable (k - ε model).
- ↪ The final form of the model with the set of coefficients will be given by Launder and Spalding (1974).
- ↪ This is the famous **standard k - ε model**.

⇒ It is the birth of modern RANS modeling.

1975 Launder, Reece and Rodi (1975)

↪ most well-known Reynolds-stress model (known as the LRR model, or the Rotta+IP model).

(making use of the previous work of Donaldson (1968), Daly and Harlow (1970), Hanjalić and Launder (1972) and Naot, Shavit and Wolfshtein (1973))

↪ This is a 7-equation model ($\overline{u_i u_j} + \varepsilon$).

⇒ The foundations are laid: the rest of the story consists in a continuous improvement of these approaches.

4. General principles for modeling

- ✓ Some principles can be applied in modeling
(=development of turbulence models).

The use of such principles has very significant advantages:

- ~> They provide a mathematical framework for modeling.
 - ~> They give physical constraints that enable the modeler to make relevant choices.
 - ~> They thus provide a method.
- ✓ However:
 - ~> They are not sufficient for ensuring that the models will “work properly”: that is the reason why modeling is still an active field of research.
 - ~> They are not necessarily compatible with each other!
 - ~> Some of them can be considered more important than others, but it is impossible to rank them: different modelers would give different rankings.

Here is a list of the most common principles, that will be detailed in the following of this chapter:

1. Closure
2. Dimensional homogeneity
3. Completeness
4. Objectivity
5. Realizability
6. Universality
7. Consistency with the boundary layer theory
8. Numerical robustness

4.1. Closure principle

✓ We have already applied this principle previously:

\leadsto this principle simply requires that a model must consist in a closed system of equations

\Rightarrow there must be as many equations as unknowns.

4.2. Dimensional homogeneity

- ✓ This principle means that the model used for a given quantity must be of the same dimension as the quantity itself.
- ✓ Example:

$$\nu_t = \ell_m^2 \sqrt{2S_{ij}S_{ij}}$$

Both sides of this relation are indeed of the dimension L^2T^{-1} .

- ✓ This simple example is obvious, but when formalizing a bit more, this basic principle can be easily forgotten.

- ✓ Example: in an article published in 1997 in *Journal of Fluid Mechanics*, L. Wang proposes the following constitutive relation for the Reynolds stress $\mathbf{R} = [\overline{u_i u_j}]$:

$$\rho \mathbf{R} = \mathbf{F}(\mu, \rho, \mathbf{U}, \nabla \mathbf{U})$$

- ∼ The objectivity principle implies that \mathbf{U} cannot be involved in this relation.
- ∼ The fact that the Reynolds number is high implies that μ is negligible and cannot be involved either.
- ⇒ These remarks lead to $\rho \mathbf{R} = \mathbf{F}(\rho, \nabla \mathbf{U})$

- ↪ Now, there is no possible combination of ρ and $\nabla \mathbf{U} = \left[\frac{\partial U_k}{\partial x_l} \right]$ that is of the correct dimension!
- ⇒ The constitutive relation introduced by Wang is not valid.
- ⇒ It is necessary to introduce additional scales in this type of relation: the turbulent energy k , its dissipation rate ε , the turbulent time scale k/ε , *etc.* are used.
- ↪ These scales play exactly the same role as the state variables (P, T, U, S) in thermodynamics: they characterize of the state of the turbulence at a given point and a given time.

✓ Moreover, dimensional analysis provides information about the shape of the relation.

↪ Coming back to the previous example: let us introduce the state variables k and ε in the constitutive relation:

$$\rho \mathbf{R} = \mathbf{F}(\rho, \nabla \mathbf{U}, k, \varepsilon) \quad (15)$$

↪ One of the fundamental principles of physics consists in the independence of the relation of the system of units.

↪ Let us choose the following system of units:

$$L = \ell \quad ; \quad T = \tau \quad ; \quad M = \rho \ell^3$$

where ℓ and τ are respectively the integral scale and the turnover time of the largest eddies. They are linked to k and ε by $\ell = k^{3/2}/\varepsilon$ and $\tau = k/\varepsilon$.

↪ In this system, the relation must remain the same:

$$\rho^* \mathbf{R}^* = \mathbf{F}(\rho^*, \nabla^* \mathbf{U}^*, k^*, \varepsilon^*)$$

where the * indicate that the variables are expressed in the new system of units.

↪ This yields

$$\rho^* = \frac{\rho}{ML^{-3}} = 1 \quad ; \quad R^* = \frac{R}{L^2 T^{-2}} = \frac{R}{k} \quad ; \quad \nabla^* U^* = \frac{\nabla U}{T^{-1}} = \frac{k \nabla U}{\varepsilon} \quad ;$$

$$k^* = \frac{k}{L^2 T^{-2}} = 1 \quad ; \quad \varepsilon^* = \frac{\varepsilon}{L^2 T^{-3}} = 1$$

which gives

$$\frac{\mathbf{R}}{k} = \mathbf{F}\left(1, \frac{k \nabla U}{\varepsilon}, 1, 1\right)$$

↪ Let us define the function

$$\mathbf{G}(\mathbf{X}) = \mathbf{F}(1, \mathbf{X}, 1, 1)$$

↪ It is seen that relation (15) is reduced to a relation between two non-dimensional quantities

$$\frac{\mathbf{R}}{k} = \mathbf{G}\left(\frac{k \nabla U}{\varepsilon}\right)$$

(this is Vaschy's proof of Vaschy-Buckingham's Π theorem).

⇒ Dimensional analysis thus constraints the form of the constitutive relations.

4.3. Completeness

A model is complete if it does not require any a priori knowledge of the flow. In other words: the formulation of the model must be the same for every flow.

✓ Example: the mixing length model is not complete.

~> Indeed, we have seen that in order to close the system we had to prescribe the mixing length as a function of the location in the flow.

~> The mixing length is dependent on the local characteristics of the flow: it is necessary to have information about the flow.

- ✓ Similarly, the one-equation model of Prandtl giving the turbulent viscosity as $\nu_t = C_\mu \sqrt{k} \ell$ is not complete: it requires to prescribe the length scale ℓ .
- ✓ Historically, the first complete model is the Kolmogorov model, which, as seen earlier, evaluates the turbulent viscosity by

$$\nu_t = C_\mu \frac{k}{\omega}$$

The model is complete because the two scales, k and ω , are given by equations of state, which are the transport equations:

$$\frac{dk}{dt} = \dots \qquad \frac{d\omega}{dt} = \dots$$

⇒ This model does not require any a priori knowledge about the flow.

✓ There are only three possibilities for building a complete model:

↪ **Eddy-viscosity models**

1. Using the Boussinesq constitutive relation $\overline{u_i u_j} = -2\nu_t S_{ij} + \frac{2}{3}k\delta_{ij}$ and writing a transport equation for ν_t .
2. Using any constitutive relation (including the Boussinesq relation) and evaluating ν_t from two turbulence scales (k - ε , k - ω , *etc.*) for which transport equations are solved.

↪ **Reynolds-stress models (second-moment closures)**

3. For Reynolds-stress models, k can be evaluated from the Reynolds stress. It is only necessary to solve a transport equation for a second scale (ε , ω , *etc.*).

4.4. Objectivity

- ✓ Reminder about continuum mechanics:

Objective quantity = it does not depend on the observer

(i.e., if it is independent on the reference frame in which it is measured)

- ✓ More formally:

change of reference frame = translation + rotation, both functions of time:

$$x_i(t) = \mathcal{R}_{ij}(t)x'_j(t) + \mathcal{T}_i(t)$$

$x_i(t)$: coordinate vector of a point M in the frame of the first observer

$x'_i(t)$: coordinate vector of the same point in the frame of the second observer

- ✓ A quantity is objective if and only if it satisfies the following relations:

$$\leadsto \text{For a scalar } s : \quad s = s'$$

$$\leadsto \text{For a vector } \mathbf{V} : \quad V_i = \mathcal{R}_{ij}V'_j$$

$$\leadsto \text{For a tensor } \mathbf{M} : \quad M_{ij} = \mathcal{R}_{ik}M'_{kl}\mathcal{R}_{lj}^{-1}$$

✓ Examples:

↪ The mean velocity \mathbf{U} is not objective: for a second observer in translation compared to the first observer (no rotation), the velocity is

$$U_i = \frac{dx_i}{dt} = \frac{dx'_i}{dt} + \frac{dT}{dt} = U'_i + \frac{dT}{dt}$$

↪ The mean-velocity-gradient tensor $\nabla \mathbf{U}$ is not objective.

↪ The strain rate tensor $\mathbf{S} = \frac{1}{2} (\nabla \mathbf{U} + {}^t\nabla \mathbf{U})$ is objective.

↪ The rotation rate tensor $\mathbf{\Omega} = \frac{1}{2} (\nabla \mathbf{U} - {}^t\nabla \mathbf{U})$ is not objective.

↪ The fluctuating velocities are objective: $u_i = u_i^* - U_i$.

↪ As a consequence, the Reynolds-stress tensor $\mathbf{R} = \overline{u_i u_j}$ is objective.

✓ Consequence for modeling: the model representing an objective quantity must be objective.

✓ Example:

We have used in section 4.2, about Wang's constitutive relation, the fact that the mean velocity vector \mathbf{U} cannot enter the constitutive relation.

Indeed, since \mathbf{U} is not objective, the Reynolds stress tensor would not be objective if \mathbf{U} was involved.

✓ On the contrary, the symmetric part \mathbf{S} of the velocity gradient can be involved, since it is objective.

✓ A relation of the form $\mathbf{R} = f(\mathbf{S}, \boldsymbol{\Omega}, \boldsymbol{\omega})$ can be objective.

($\boldsymbol{\omega}$ is the system rotation vector with respect to a Galilean reference frame)

$$\nabla \mathbf{U} = \mathbf{S} + \boldsymbol{\Omega}$$

Symmetric part (objective): $\mathbf{S} = \frac{1}{2} (\nabla \mathbf{U} + {}^t \nabla \mathbf{U})$ $S_{ij} = \frac{1}{2} \left(\frac{\partial U_i}{\partial x_j} + \frac{\partial U_j}{\partial x_i} \right)$

Skew-symmetric part (non-objective):

$$\boldsymbol{\Omega} = \frac{1}{2} (\nabla \mathbf{U} - {}^t \nabla \mathbf{U}) \quad \Omega_{ij} = \frac{1}{2} \left(\frac{\partial U_i}{\partial x_j} - \frac{\partial U_j}{\partial x_i} \right)$$

✓ The relation is objective if and only if it can be written in the form

$$\mathbf{R} = f(\mathbf{S}, \mathbf{W})$$

where $W_{ij} = \Omega_{ij} + \varepsilon_{mji} \omega_m$ is called the *absolute rotation rate tensor*

✓ Conclusion: modelers work with \mathbf{S} and \mathbf{W} .

4.5. Realizability

The moments of order 1 (e.g., U_i), of order 2 (e.g., $\overline{u_i u_j}$), *etc.*, have some mathematical properties.

- ✓ We have considered that the instantaneous velocities are random variables.
- ✓ We are trying to model the moments of these random variables (particularly of order 2, the Reynolds stress tensor $\mathbf{R} = \overline{u_i u_j}$).
- ✓ A model can represent these moments of a random variable (i.e., is realizable) only if satisfies some properties.

4.5.1. Use in modeling

The Reynolds stress tensor being composed of the second moment of the velocity u_i^* , it satisfies the realizability conditions.

In particular, the Reynolds stress tensor is a positive-semidefinite tensor.

$$\forall n_i, \quad \begin{bmatrix} n_1 & n_2 & n_3 \end{bmatrix} \begin{bmatrix} \overline{u^2} & \overline{uv} & \overline{uw} \\ \overline{uv} & \overline{v^2} & \overline{vw} \\ \overline{uw} & \overline{vw} & \overline{w^2} \end{bmatrix} \begin{bmatrix} n_1 \\ n_2 \\ n_3 \end{bmatrix} = R_{ij}n_i n_j \geq 0$$

It is often what is meant by realizability in turbulence modeling.

However, numerous properties can be applied to higher moments.

4.5.2. Realizability of the Reynolds stress tensor

The positivity of the Reynolds stress tensor can be written in different ways:

- ✓ The 3 eigenvalues are positive.
- ✓ The principal minors are positive:

$$\overline{u^2} \geq 0$$

$$\overline{uv}^2 \leq \overline{u^2} \overline{v^2}$$

$$\overline{u^2} \overline{v^2} \overline{w^2} + 2 \overline{uv} \overline{uw} \overline{vw} - \overline{v^2} \overline{uw}^2 - \overline{w^2} \overline{uv}^2 - \overline{u^2} \overline{vw}^2 \geq 0$$

(others can be obtained by permutation: $\overline{v^2} \geq 0$, $\overline{w^2} \geq 0$, $\overline{uw}^2 \leq \overline{u^2} \overline{w^2}$, $\overline{vw}^2 \leq \overline{v^2} \overline{w^2}$)

- ✓ The 3 principal invariants are positive (coefficients of the characteristic equation):

$$I_1^R \geq 0 \quad ; \quad I_2^R \geq 0 \quad ; \quad I_3^R \geq 0$$

4.5.3. Conclusions about realizability

- ✓ Realizability gives a number of relations that guide the modeling.
- ✓ However, ensuring realizability under all circumstances is very difficult.
- ✓ As a practical matter, it can only be imposed in some simplified situation: for instance, by considering homogeneous turbulence, the transport terms vanish and one can focus on the terms ϕ_{ij}^* and ε_{ij} .

4.6. Universality

Ultimate aim of modeling: write a model that can be applied to all flows (or at least to a wide class of flows: for instance, single-phase flows of incompressible fluids).

✓ Utopian: Reynolds averaging = huge loss of information

~> trying to build a model able to correctly reproduce *all the phenomena*, in *all situations*, is hopeless.

~> Consensus today: modeling will never reach this goal.

~> What can be expected from a model?

- ▷ Being applicable to all the situations.
- ▷ Giving results that are satisfactory for the widest possible range of flows.
- ▷ Having limitations that are known.

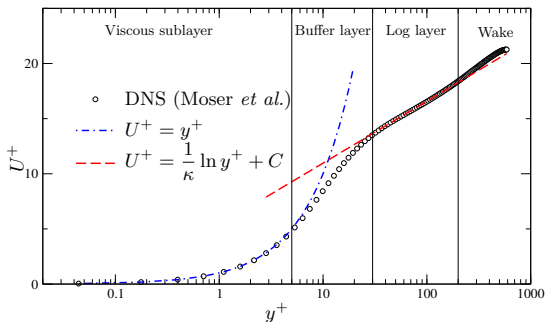
- ✓ Many research programs, in particular European programs, aim at applying numerous models to numerous test cases, in order to evaluate their potential, their shortcomings and possibly evaluate uncertainties.
- ✓ A recent and fashionable research topic is UQ (Uncertainty Quantification): the uncertainty due to the model is difficult to evaluate.
- ✓ However, a *weak form of universality* must be satisfied:
 - ↪ The coefficients of a model must be fixed after a calibration procedure in simple flows and must thereafter be considered as “universal”, i.e., applicable to any flow.
 - ↪ Modifying the coefficients of a model to *fit* experimental results is not admissible. This kind of *tuning* has been very detrimental to the reputation of CFD.

4.7. Consistency with the boundary layer theory

- ✓ A model is expected to reproduce at least a standard boundary layer.

Historically, first models (mixing length models) were developed *for boundary layer flows*, in particular for aeronautics.

- ✓ New models are expected to provide boundary layer results at least as good as those provided by existing models.



DNS of Moser *et al.* at $Re_\tau = 590$.

- ✓ There are not many theories in turbulence: the log layer theory (Prandtl, Izakson, Millikan) is one the most solid of them.
- ✓ It will be seen that the Karman constant κ (slope in the lin-log plot) is one of the constraints used to calibrate the models.

4.8. Numerical robustness

- ✓ As introduced at the beginning of this course, the turbulence models are not supposed to stay confined in the research field, but to be used in industry.
- ✓ The numerical difficulties faced when trying to integrate the system of equations (difficulties to converge, and even divergence) are often underestimated by modelers.
- ✓ A compromise must be found between two very important constraints:
 - ↪ a good representation of the physics;
 - ↪ numerical robustness and ease of use.

- ✓ Example: in order to model the slow part of the redistribution term, Speziale, Sarkar and Gatski (1991) propose the following nonlinear form:

$$\phi_{ij}^l = - \left(C_1 + C_1^* \frac{P}{\varepsilon} \right) \varepsilon b_{ij} + C_2 \varepsilon \left(b_{ik} b_{kj} - \frac{1}{3} b_{kl} b_{kl} \delta_{ij} \right)$$

- ↪ The nonlinear term is necessary to correctly reproduce the return to isotropy experiment of an initially anisotropic, homogeneous turbulence (for both strong and weak anisotropies).
- ↪ This term has however a limited influence in most of the common flows.
- ↪ Since it introduces a nonlinearity which make the numerical resolution more difficult, this term is often dropped.
- ✓ Warning: It is often possible to improve the numerical scheme
⇒ simplifying the models is not always the better solution.

5. Eddy-viscosity models (first moment closures)

5.1. Constitutive relation

- ✓ Reynolds-averaged Navier–Stokes equations

$$\frac{\partial U_i}{\partial x_i} = 0 \quad ; \quad \frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial P}{\partial x_i} + \nu \frac{\partial^2 U_i}{\partial x_j \partial x_j} - \frac{\partial \overline{u_i u_j}}{\partial x_j}$$

- ✓ A first moment closure = constitutive relation (model) which expresses the second moments ($\overline{u_i u_j}$) as a function of the first moments.
- ✓ We are seeking a relation in the form: $\mathbf{R} = f(\nu, \rho, P, \mathbf{U}, \beta_n)$
where f is any functional, which can involve variables at different points and different times.
- ✓ The scalar quantities β_n are state variables which globally characterize the state of the turbulence (k, ε, \dots).

5.1.1. General assumptions

Different remarks and assumptions make simplifications possible:

- ✓ The mean pressure does not appear in the exact Reynolds-stress equations
⇒ there is no reason to take it into account in the model.
- ✓ One can assume that f only involve the present instant t (assumption of instantaneity):
⇒ arbitrarily, the possibility of a *memory effect* of turbulence (which exists), is excluded.
- ✓ At high Reynolds number, the effects of viscosity are negligible: the global diffusive properties of turbulence are linked to the larger eddies of the flow (energetic structures) and, when the Reynolds number ν_t/ν is sufficiently high, are independent of the molecular viscosity.
⇒ ν can then be removed from the relation for the moment.
One must reconsider this issue for a low-Reynolds number turbulence.

✓ As usual in continuum mechanics, assumption of *locality*:

↪ The relation only involves the neighborhood of point \mathbf{x} .

↪ For instance, in the constitutive relation, the dependance on a the scalar *field*

$$\beta_i(\mathbf{x}'), \quad \forall \mathbf{x}' \in \Omega$$

is replaced by a dependance on the derivatives of the variable evaluated at point \mathbf{x}

$$\frac{\partial^k \beta_n}{\partial x_k^n}(\mathbf{x}), \quad k \geq 0$$

✓ Most of the time, derivatives of order ≥ 2 are neglected (standard assumption of a *materially simple medium*)

⇒ one considers that the variables and their first derivatives are sufficient to describe the properties of turbulence.

✓ In particular: the mean velocity gradient $\frac{\partial U_i}{\partial x_j}$ must be retained in the models.

↪ It appears in the exact equation of the Reynolds stress tensor (production term).

↪ It is crucial for describing the dynamics of turbulence (actually the most important parameter).

- ✓ Most of the time, gradients of scalar quantities $\beta_n(k, \varepsilon, \dots)$ are also neglected.
- ✓ Objectivity constraint (the modeled \mathbf{R} must be objective) $\Rightarrow \mathbf{U}$ cannot enter the relation.
- ✓ As explained before: objectivity + sensitivity to system rotation
 - \leadsto the constitutive relation cannot be a function of $\partial U_i / \partial x_j$ (not objective)
 - \leadsto rather a function of:

▷ the strain rate tensor:

$$S_{ij} = \frac{1}{2} \left(\frac{\partial U_i}{\partial x_j} + \frac{\partial U_j}{\partial x_i} \right)$$

▷ the absolute rotation rate tensor \mathbf{W} :

$$W_{ij} = \frac{1}{2} \left(\frac{\partial U_i}{\partial x_j} - \frac{\partial U_j}{\partial x_i} \right) + \varepsilon_{mji} \omega_m$$

This constraints and assumptions lead to the following formulation:

$$\mathbf{R}(\mathbf{x}, t) = f\left(\rho, \mathbf{S}(\mathbf{x}, t), \mathbf{W}(\mathbf{x}, t), \beta_n(\mathbf{x}, t)\right)$$

5.1.2. Dimensional analysis

- ✓ We know that the diffusive properties of turbulence must be described by two characteristic scales

$$\nu_t = \ell u = \ell^2 \tau^{-1}$$

- ✓ ℓ and τ are the integral scale of turbulence and the turnover time of the large eddies.
- ✓ Consequently, in order to correctly represent the diffusive effect of turbulence, ℓ and τ must be introduced as state variables.

$$\mathbf{R} = f(\rho, \mathbf{S}, \mathbf{W}, \ell, \tau)$$

✓ Dimensional analysis imposes that this relation must be valid whatever the system of units.

✓ Let us arbitrarily choose the following system of units:

$$L = \ell \quad ; \quad T = \tau \quad ; \quad M = \rho \ell^3$$

✓ In this new system of units, the relation is

$$\mathbf{R}^* = f(1, \mathbf{S}^*, \mathbf{W}^*, 1, 1)$$

where $\mathbf{R}^* = \frac{\mathbf{R}}{\ell^2 \tau^{-2}}$, $\mathbf{S}^* = \tau \mathbf{S}$ and $\mathbf{W}^* = \tau \mathbf{W}$.

✓ Defining $g(x, y) = f(1, x, y, 1, 1)$ this relation reduces to

$$\mathbf{R}^* = g(\mathbf{S}^*, \mathbf{W}^*) \tag{16}$$

5.1.3. General solution

- ✓ Theory of invariants (cf. Deville and Gatski, 2012): a symmetric tensorial function of two tensors (one symmetric, one skew-symmetric), is a polynomial of the form

$$\begin{aligned} \mathbf{R}^* = & g_0 \mathbf{I} + g_1 \mathbf{S}^* + g_2 (\mathbf{S}^* \mathbf{W}^* - \mathbf{W}^* \mathbf{S}^*) + g_3 \left(\mathbf{S}^{*2} - \frac{1}{3} \{ \mathbf{S}^{*2} \} \mathbf{I} \right) + \\ & g_4 \left(\mathbf{W}^{*2} - \frac{1}{3} \{ \mathbf{W}^{*2} \} \mathbf{I} \right) + g_5 (\mathbf{W}^* \mathbf{S}^{*2} - \mathbf{S}^{*2} \mathbf{W}^*) + \\ & g_6 \left(\mathbf{W}^{*2} \mathbf{S}^* + \mathbf{S}^* \mathbf{W}^{*2} - \frac{2}{3} \{ \mathbf{S}^* \mathbf{W}^{*2} \} \mathbf{I} \right) + g_7 (\mathbf{W}^* \mathbf{S}^* \mathbf{W}^{*2} - \mathbf{W}^{*2} \mathbf{S}^* \mathbf{W}^*) + \\ & g_8 (\mathbf{S}^* \mathbf{W}^* \mathbf{S}^{*2} - \mathbf{S}^{*2} \mathbf{W}^* \mathbf{S}^*) + g_9 \left(\mathbf{W}^{*2} \mathbf{S}^{*2} + \mathbf{S}^{*2} \mathbf{W}^{*2} - \frac{2}{3} \{ \mathbf{S}^{*2} \mathbf{W}^{*2} \} \mathbf{I} \right) + \\ & g_{10} (\mathbf{W}^* \mathbf{S}^{*2} \mathbf{W}^{*2} - \mathbf{W}^{*2} \mathbf{S}^{*2} \mathbf{W}^*) \end{aligned}$$

- ✓ The coefficients g_i are not constant, but rather functions that can only depend on the six independent invariants formed from the tensors \mathbf{S}^* and \mathbf{W}^* : $\{ \mathbf{S}^{*2} \}$, $\{ \mathbf{W}^{*2} \}$, $\{ \mathbf{S}^{*3} \}$, $\{ \mathbf{W}^{*2} \mathbf{S}^* \}$, $\{ \mathbf{S}^{*2} \mathbf{W}^{*2} \}$, $\{ \mathbf{S}^* \mathbf{W}^* \mathbf{S}^{*2} \mathbf{W}^{*2} \}$, where $\{.\}$ denotes the trace.
- ✓ Despite the numerous (very restrictive) hypotheses (instantaneity, locality, *etc.*) the relation is still very complex, involving fifth order nonlinear terms.
- ✓ Fortunately, the Cayley–Hamilton theorem limits the degree of each tensors in this relation to 2.

✓ Such a complex relation is almost inapplicable:

~> The model would be very difficult to numerically integrate.

~> The coefficients g_i are not provided by the theory. Even though they are often considered constant, the calibration process would be very difficult: how to find values of the 10 coefficients from simple flows?

⇒ the nonlinear terms of highest degree must be “sacrificed”.

✓ However, there is a very recent renewal of nonlinear models

↪ The variable coefficients can be obtained by machine learning

↪ A neural network can give the values of the coefficients as a function of the invariants. In this case, the relation is not known explicitly

↪ Another approach consists in finding an explicit relation (function) using a genetic algorithm (Gene Expression Programming, GEP)

↪ In all cases, a large amount of data is required for the learning phase

↪ It is a very active research topic. It is not a mature technique

↪ It does not solve the numerical issues

5.1.4. Reduced models

Cubic models

- ✓ Craft, Launder and Suga (1996): necessary to retain the cubic terms to reproduce the *curvature effects* (sensitivity of the Reynolds stresses to the curvature of the streamlines).
- ✓ Therefore, in some commercial codes, cubic models are proposed.
- ✓ The constitutive relation is then truncated by taking g_7 to g_{10} equal to zero.

Quadratic models

- ✓ In a statistically 2D flow (averaged quantities independent of z , for instance), the last six term (g_5 - g_{10}) can be shown dependent on the first five terms (g_0 - g_4).

- ✓ The relation thus simplifies as

$$\mathbf{R}^* = g_0 \mathbf{I} + g_1 \mathbf{S}^* + g_2 (\mathbf{S}^* \mathbf{W}^* - \mathbf{W}^* \mathbf{S}^*) + g_3 \left(\mathbf{S}^{*2} - \frac{1}{3} \{ \mathbf{S}^{*2} \} \mathbf{I} \right)$$

- ✓ This relation is the exact solution for Eq. (16) in 2D flows.

- ✓ It is also used in 3D flows, as simplified models.

- ✓ We have

$$\overline{u_i u_j} = g_0 \frac{\ell^2}{\tau^2} \delta_{ij} + g_1 \frac{\ell^2}{\tau} S_{ij} + g_2 \ell^2 (S_{ik} W_{kj} - W_{ik} S_{kj}) + g_3 \ell^2 \left(S_{ik} S_{kj} - \frac{1}{3} S_{kl} S_{kl} \delta_{ij} \right)$$

- ✓ This type of models is also included in some commercial codes.

Linear models

- ✓ Finally, one can decide, for simplicity and robustness reasons, to only keep the linear terms, which leads to

$$\overline{u_i u_j} = g_0 \frac{\ell^2}{\tau^2} \delta_{ij} + g_1 \frac{\ell^2}{\tau} S_{ij}$$

- ✓ The trace is $2k = 3g_0 \frac{\ell^2}{\tau^2}$ since $k = \frac{1}{2} \overline{u_i u_i}$, $\delta_{ii} = 3$ and $S_{ii} = 0$
such that $g_0 = \frac{2}{3}$ and $\frac{\ell^2}{\tau^2} = k$ are used

- ✓ Remember that $\nu_t \propto \ell u = \frac{\ell^2}{\tau}$

It is usual to note the proportionality coefficient C_μ with $C_\mu = -\frac{g_1}{2}$
such that we have

$$\overline{u_i u_j} = \frac{2}{3} k \delta_{ij} - 2\nu_t S_{ij}$$

- ✓ This is the Boussinesq relation.

- ✓ It is seen that this relation (which was initially introduced by a simple phenomenological reasoning) is the consequence of *a long series of simplifying assumptions*:
 - ↪ instantaneity: no influence of the history of the strain and the turbulence;
 - ↪ locality: the turbulence is only influenced by its vicinity;
 - ↪ materially simple medium;
 - ↪ linearity of the constitutive relation.
- ✓ It is thus clear that linear eddy-viscosity models cannot perform well in all the situations.

5.2. One-equation models

- ✓ The constitutive relations that have been derived, either linear or nonlinear, involve two independent turbulent scales, characterizing the energetic eddies.
- ✓ The particular choice of the scales does not matter a priori:

↪ The integral length scale ℓ

↪ The eddy-turnover time τ

↪ The turbulent energy $k = \frac{\ell^2}{\tau^2}$

↪ The dissipation rate $\varepsilon = \frac{\ell^2}{\tau^3}$

↪ The eddy-viscosity $\nu_t = \frac{\ell^2}{\tau}$

↪ The specific dissipation $\omega = \frac{\varepsilon}{k} = \frac{1}{\tau}$

↪ Any other combination $\varphi = \ell^a \tau^b$

- ✓ Reminder: a model is complete if no a priori knowledge about the flow is necessary \Rightarrow these scales must be obtained as solution of transport equations, *not prescribed by the user*.
- ✓ Particular case: linear constitutive relation (Boussinesq relation)

$$\mathbf{R} = -2\nu_t \mathbf{S} + \frac{2}{3}k \mathbf{I}$$

\leadsto The Reynolds-averaged Navier–Stokes equations become

$$\frac{\partial U_i}{\partial x_i} = 0$$

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial P^*}{\partial x_i} + \frac{\partial}{\partial x_j} \left[\left(\nu + \boxed{\nu_t} \right) \left(\frac{\partial U_i}{\partial x_j} + \frac{\partial U_j}{\partial x_i} \right) \right]$$

where $P^* = P + \frac{2}{3}\rho \boxed{k}$.

\leadsto Solving this system of equations gives U_i and P^* .

\Rightarrow k is not needed to obtain the mean velocity.

\Rightarrow A complete model can be obtained by writing a transport equation for ν_t .

\leadsto P^* is NOT the pressure, but since $k = 0$ at solid boundaries, $P^* = P$ at the surface of the obstacles \Rightarrow the forces (drag, lift) can be computed.

- ✓ The only model using a transport equation for ν_t used in industrial applications is the Spalart-Allmaras model (1992).

\leadsto ν_t is given by $\nu_t = f_{v1}\tilde{\nu}$, and $\tilde{\nu}$ is obtained through a transport equation:

$$\frac{\partial \tilde{\nu}}{\partial t} + U_j \frac{\partial \tilde{\nu}}{\partial x_j} = c_{b1} \tilde{S} \tilde{\nu} - c_{w1} f_w \left(\frac{\tilde{\nu}}{d} \right)^2 + \frac{1}{\sigma} \frac{\partial}{\partial x_k} \left[(\nu + \tilde{\nu}) \frac{\partial \tilde{\nu}}{\partial x_k} \right] + \frac{c_{b2}}{\sigma} \frac{\partial \tilde{\nu}}{\partial x_k} \frac{\partial \tilde{\nu}}{\partial x_k}$$

\leadsto The details about the coefficients c_i and the functions f_i can be found in the original article or, for instance, in the book of Wilcox (1998).

\leadsto The functions f_i are in particular introduced such that the model is valid in the near-wall region.

~> The model is used a lot in aeronautics, for different reasons:

- ▷ It is easy to integrate numerically.
- ▷ For attached flows, it produces results as good as those given by zero equations models (Cebeci–Smith, Baldwin–Lomax).
- ▷ For detached flows, it gives a much better description of the flow field than zero equation models.

~> However, the model is too simple (only one equation) to be valid in a wide range of flows.

~> For instance, it give wrong results in jet flows (see Wilcox, 1998), overestimating the expansion rate by more than 40%.

5.3. k - ε models

✓ Two reasons for using two scales and not only one:

↪ One wants to get rid of the limitations of the Spalart–Allmaras model.

↪ One uses a nonlinear constitutive relation:

▷ For instance, the quadratic constitutive relation is:

$$\overline{u_i u_j} = \frac{2}{3} \frac{\ell^2}{\tau^2} \delta_{ij} + g_1 \frac{\ell^2}{\tau} S_{ij} + g_2 \ell^2 (S_{ik} W_{kj} - W_{ik} S_{kj}) + g_3 \ell^2 \left(S_{ik} S_{kj} - \frac{1}{3} S_{kl} S_{kl} \delta_{ij} \right)$$

▷ The two scales ℓ and τ cannot be combined in only one scale.

✓ As said earlier, various scales can be used instead of ℓ and τ .

- ✓ The most standard choice nowadays is to determine ℓ and τ from the turbulent energy k and its dissipation rate ε (k - ε models):

$$\ell = \frac{k^{3/2}}{\varepsilon} \quad ; \quad \tau = \frac{k}{\varepsilon}$$

- ✓ Another rather common choice: turbulent energy k and specific dissipation rate $\omega = \varepsilon/k$ (k - ω models):

$$\ell = \frac{k^{1/2}}{\omega} \quad ; \quad \tau = \frac{1}{\omega}$$

5.3.1. Transport equations

✓ The exact transport equation for k is

$$\frac{\partial k}{\partial t} + U_k \frac{\partial k}{\partial x_k} = -\overline{u_i u_k} \frac{\partial U_i}{\partial x_k} - \varepsilon + \frac{\partial}{\partial x_k} \left(\nu \frac{\partial k}{\partial x_k} \right) + \frac{\partial}{\partial x_k} \left(\frac{1}{\rho} \overline{p u_k} - \frac{1}{2} \overline{u_i u_i u_k} \right)$$

✓ The right hand side involves:

↪ sources terms (production and dissipation, the latter being actually always a sink)

↪ transport terms: viscous diffusion, pressure “diffusion” and turbulent transport.

Source terms and transport terms

- ✓ The transport equations are all in the symbolic form

$$\frac{\partial f}{\partial t} + \frac{\partial U_k f}{\partial x_k} = \frac{\partial}{\partial x_k} F_k + s$$

where f can be a scalar, a vector or a second order tensor.

- ✓ The terms to be modeled are all on the right hand side.

✓ Integrating this equation over a control volume yields

$$\frac{\partial}{\partial t} \int_{\Omega} f \, dV = - \int_{\partial\Omega} U_k f \, n_k \, dS + \int_{\partial\Omega} F_k \, n_k \, dS + \int_{\Omega} s \, dV$$

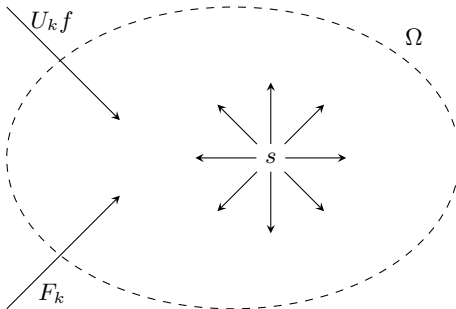
⇒ The total variation of f over Ω is due to two types of terms:

↪ Transport terms through the surface $\partial\Omega$ by the fluxes $U_k f$ and F_k :

These fluxes are of convective or diffusive type:

▷ they do not create nor destroy the quantity f inside the volume Ω , contrary to source terms

▷ but rather *transport* f through the surface



∼ Source (or sink) terms s : generation (or destruction) terms for the quantity f .

Examples:

- ▷ a chemical reaction can produce carbon oxide in the volume \Rightarrow source term in the transport equation for carbon oxide.
- ▷ a volume force (due to gravity, Coriolis, magnetic field, *etc.*) creates momentum (acceleration) \Rightarrow source terms in the transport equation for momentum.
- ▷ Turbulent production P is the generation term (or sometimes the destruction term) of turbulent energy $k \Rightarrow$ source (or sink) term in the k -equation.
- ▷ Dissipation transforms turbulent kinetic energy into heat \Rightarrow sink term in the k -equation and source term in transport equation for heat.

Modeling of the source terms of k

$$\frac{\partial k}{\partial t} + U_k \frac{\partial k}{\partial x_k} = -\overline{u_i u_k} \frac{\partial U_i}{\partial x_k} - \varepsilon + \frac{\partial}{\partial x_k} \left(\nu \frac{\partial k}{\partial x_k} \right) + \frac{\partial}{\partial x_k} \left(\frac{1}{\rho} \overline{p u_k} - \frac{1}{2} \overline{u_i u_i u_k} \right)$$

✓ Production term $P = -\overline{u_i u_k} \frac{\partial U_i}{\partial x_k}$:

↪ The model for P is simply obtained by introducing the constitutive relation (model for $\overline{u_i u_j}$).

↪ For instance, linear constitutive relation (Boussinesq relation)

$$\overline{u_i u_j} = -2\nu_t S_{ij} + \frac{2}{3} k \delta_{ij} \Rightarrow P = 2\nu_t S_{ij} S_{ij}$$

✓ Two terms require modeling:

↪ $\frac{\partial}{\partial x_k} \left(\frac{1}{\rho} \overline{p u_k} - \frac{1}{2} \overline{u_i u_i u_k} \right)$ (so-called turbulent diffusion)

↪ ε (dissipation)

Modeling of the transport terms

✓ The two turbulent terms $\overline{pu_k}$ and $-\overline{u_i u_i u_k}$ are generally modeled as a whole.

✓ It is assumed that the effect of these terms is diffusive

$$\Rightarrow \text{similar to the viscous term: } \frac{1}{\rho} \overline{pu_k} - \overline{u_i u_i u_k} = \frac{\nu_t}{\sigma_k} \frac{\partial k}{\partial x_k}$$

where σ_k is a Prandtl number.

This model is called *the Simple Gradient Diffusion Hypothesis* (SGDH).

✓ This type of model would be perfectly justified:

↪ if the turbulent agitation had a behavior of “pure agitation” as the molecular agitation

↪ if this agitation were at a scale very small compared to the mean flow scale

(in other words, if turbulence had characteristics similar to those of the molecular agitation).

✓ Actually, this hypothesis can be criticized:

↪ There is no scale separation in turbulence

↪ There are coherent structures (not a pure agitation)

↪ Pressure “diffusion” does not always behave the same way as the turbulent diffusion:

▷ In some cases pressure diffusion transport energy in the opposite direction

⇒ It can be anti-diffusive

▷ Some authors have proposed models specific to this term.

✓ Using the SGDH model, the modeled k equation is

$$\frac{\partial k}{\partial t} + U_k \frac{\partial k}{\partial x_k} = P - \varepsilon + \frac{\partial}{\partial x_k} \left[\left(\nu + \frac{\nu_t}{\sigma_k} \right) \frac{\partial k}{\partial x_k} \right]$$

The ε -equation

- ✓ ε is defined as

$$\varepsilon = \nu \overline{\frac{\partial u_i}{\partial x_k} \frac{\partial u_i}{\partial x_k}}$$

- ✓ The exact transport equation for ε is very complex.
- ✓ Individual modeling of the different terms has never been successful.
- ✓ A phenomenological approach is retained:

The ε -equation must involve the same terms as the k -equation: convection, source term (production), sink term (dissipation), viscous diffusion, turbulent diffusion:

$$\frac{\partial \varepsilon}{\partial t} + U_k \frac{\partial \varepsilon}{\partial x_k} = P_\varepsilon - \epsilon_\varepsilon + \frac{\partial}{\partial x_k} \left[\left(\nu + \frac{\nu_t}{\sigma_\varepsilon} \right) \frac{\partial \varepsilon}{\partial x_k} \right]$$

✓ The source/sink terms P_ε and ϵ_ε must be modeled.

↪ It is proposed to related them to the corresponding terms in the k -equation, i.e., P and ε .

↪ A simple dimensional analysis provides a relation between the production and dissipation terms of ε to those of k :

$$P_\varepsilon = C_{\varepsilon 1} \frac{\varepsilon}{k} P \quad ; \quad \epsilon_\varepsilon = C_{\varepsilon 2} \frac{\varepsilon}{k} \varepsilon$$

- ✓ The modeled equation for ε is finally written as

$$\frac{\partial \varepsilon}{\partial t} + U_k \frac{\partial \varepsilon}{\partial x_k} = C_{\varepsilon 1} \frac{\varepsilon}{k} P - C_{\varepsilon 2} \frac{\varepsilon^2}{k} + \frac{\partial}{\partial x_k} \left[\left(\nu + \frac{\nu_t}{\sigma_\varepsilon} \right) \frac{\partial \varepsilon}{\partial x_k} \right]$$

- ✓ Developing this phenomenological approach, Harlow & Nakayama (1968) and Hanjalić (1970) have been very daring!
- ✓ Although this equation is very often criticized, because of its phenomenological character, it is still very widely used nowadays, for eddy-viscosity models as well as second moment closures.
- ✓ The equation is certainly not perfect, but nobody has really been able to propose better for the moment.

5.3.2. The standard k - ε model

✓ Launder and Spalding (1974) have used all these ideas:

$$\overline{u_i u_j} = -2\nu_t S_{ij} + \frac{2}{3}k\delta_{ij}$$

$$\nu_t = C_\mu \frac{k^2}{\varepsilon} \quad ; \quad P = 2\nu_t S_{ij} S_{ij}$$

$$\frac{\partial k}{\partial t} + U_k \frac{\partial k}{\partial x_k} = P - \varepsilon + \frac{\partial}{\partial x_k} \left[\left(\nu + \frac{\nu_t}{\sigma_k} \right) \frac{\partial k}{\partial x_k} \right]$$

$$\frac{\partial \varepsilon}{\partial t} + U_k \frac{\partial \varepsilon}{\partial x_k} = C_{\varepsilon 1} \frac{\varepsilon}{k} P - C_{\varepsilon 2} \frac{\varepsilon^2}{k} + \frac{\partial}{\partial x_k} \left[\left(\nu + \frac{\nu_t}{\sigma_\varepsilon} \right) \frac{\partial \varepsilon}{\partial x_k} \right]$$

and calibrated the coefficients:

$$C_\mu = 0.09 \quad ; \quad \sigma_k = 1 \quad ; \quad C_{\varepsilon 1} = 1.44 \quad ; \quad C_{\varepsilon 2} = 1.92 \quad ; \quad \sigma_\varepsilon = 1.3$$

✓ These equations + these coefficients = the standard k - ε model.

5.4. k - ω model

- ✓ Wilcox (1988): back to the old Kolmogorov idea of using the characteristic frequency of the large eddies (also called *specific dissipation*) $\omega = \frac{\varepsilon}{C_\mu k}$ as the second scale
- ✓ Transport equation for ω in the same form as the ε equation
- ✓ Linear constitutive relation

$$\overline{u_i u_j} = -2 \nu_t S_{ij} + \frac{2}{3} k \delta_{ij}$$

$$\nu_t = \frac{k}{\omega} \quad ; \quad P = 2 \nu_t S_{ij} S_{ij}$$

$$\frac{\partial k}{\partial t} + U_k \frac{\partial k}{\partial x_k} = P - \beta' k \omega + \frac{\partial}{\partial x_k} \left[\left(\nu + \frac{\nu_t}{\sigma_k} \right) \frac{\partial k}{\partial x_k} \right]$$

$$\frac{\partial \omega}{\partial t} + U_k \frac{\partial \omega}{\partial x_k} = \alpha \frac{\omega}{k} P - \beta \omega^2 + \frac{\partial}{\partial x_k} \left[\left(\nu + \frac{\nu_t}{\sigma_\omega} \right) \frac{\partial \omega}{\partial x_k} \right]$$

$$\beta' = 0.09 \quad ; \quad \sigma_k = 2 \quad ; \quad \alpha = 5/9 \quad ; \quad \beta = 3/40 \quad ; \quad \sigma_\omega = 2$$

✓ This model has some advantages compared to the standard k - ε model, but also major drawbacks.

✓ **Advantages:**

↪ The model is integrable down to the wall. This is a major advantage.

↪ It gives better results than the standard k - ε model in adverse pressure gradients situations \Rightarrow it predicts better the separation location.

↪ The ω equation is actually independent on k , since its production term reduces to

$$\alpha \frac{\omega}{k} P = 2\alpha S_{ij} S_{ij}$$

This feature is very interesting, since it uncouples the ω equation from the k equation, which is a factor of numerical robustness.

↪ It is possible to show that only the powers of ω satisfy this feature. The choice of ω is then optimal from a numerical point of view.

✓ Drawbacks:

→ The ω variable tends to infinity at the wall

$$\omega(y) \underset{y \rightarrow 0}{\sim} \frac{6\nu}{\beta y^2}$$

which is the analytical solution of the ω equation in the near-wall region.

→ As a practical matter, Menter (1994) proposed the Dirichlet boundary condition

$$\omega = 10 \frac{6\nu}{\beta y_1^2}$$

where y_1 is the distance between the center of the cell adjacent to the wall and the wall.

→ Remark: this condition is not *consistent* (the discrete solution does not converge towards the exact solution when the grid is refined)!

→ It also implies that the correct near-wall behavior $k \underset{y \rightarrow 0}{\sim} y^2$ cannot be reproduced

- ~> In external aerodynamics (e.g., a wing), the model predicts a boundary layer development sensitive to the turbulence level present outside of the boundary layer, which is usually fixed arbitrarily by the user \Rightarrow this model is not applicable to external aerodynamics.
- \Rightarrow This model is seldom applied: however, it is used as a near-wall model associated to the $k-\varepsilon$ model far from the wall ($k-\omega$ SST model, described later).

5.5. Advantages and limitations of eddy-viscosity models

Three parts:

- ✓ General indications concerning all the eddy-viscosity models.
- ✓ Focus on the standard k - ε model.
- ✓ Finally, on nonlinear eddy-viscosity models.

5.5.1. Eddy-viscosity models in general

- ✓ Advantages: can be described in one sentence:

They contain less differential equations than Reynolds stress models.

↪ This means that they are less expensive in terms of CPU time

↪ However, the difference is not always large (an implicit CFD code spends most of the time solving the velocity–pressure system).

↪ They are not necessarily more numerically stable than Reynolds stress models: stability depends on the linearity or nonlinearity of the model.

✓ Limitations:

- ~> Eddy-viscosity models assume a relation between the Reynolds stress and the mean field (through \mathbf{S} and \mathbf{W}): this is not true in general
 - \Rightarrow the prediction of complex flows, in particular 3D flows, can be problematic.
- ~> This relation is assumed instantaneous \Rightarrow the response to a change in the strain is instantaneous: this is wrong.
Actually, turbulence is sensitive to the *history* of the strain (memory effect).
- ~> The transport equation of the second scale (often ε) is very empirical: many people brought it into question, but very few have proposed other approaches.

~> The models predict in general an expansion rate of axisymmetric jets larger than the expansion rate of plane jets, while experiments evidence exactly the opposite:

▷ this shortcoming is known as the *round jet/plane jet anomaly*.

▷ Corrections have been proposed (e.g., Pope, 1978), but are not widely used.

~> Models based on the second scale ε have difficulties in reproducing the effects due to adverse pressure gradients.

In particular, this implies that separation points are not always well predicted.

5.5.2. The standard k - ε model

- ✓ This model is based on a linear constitutive relation, which has some advantages, and many limitations.
- ✓ Advantages:
 - ↪ The model is easy to implement and numerically very stable.
 - ↪ It correctly reproduces the shear stress in free shear flows (for instance \overline{uv} produced by the shear $\partial U / \partial y$).
 - ↪ Its behavior is somehow predictable: the model makes always the same type of errors
 - ⇒ without being able to give error bars, such systematic errors can be taken into account for the interpretation of the results.
 - ↪ It is, despite its simplicity, applicable to much wider range of situation than the Spalart–Allmaras model.

✓ Limitations:

- ↪ Linearity of the model \Rightarrow virtually no chance to correctly reproduce complex flows, in particular 3D flows (see, for instance, the wake/mixing layer interaction).
- ↪ Proportionality between anisotropy and strain \Rightarrow the production is always predicted positive ($P = 2\nu_t S_{ij} S_{ij}$).
- ↪ *Not sensitive at all* to the effects due to the streamline curvatures and a solid body rotation: :
 - ▷ It gives exactly the same solution in a channel flow with and without rotation.
- \Rightarrow It is completely irrelevant to turbomachinery problems, for instance.
 - ▷ Rotation/curvature correction are often introduced.

↪ In a simple shear flow: predicts equal diagonal components of the Reynolds stress $\overline{u^2} = \overline{v^2} = \overline{w^2}$

⇒ it is not able to reproduce the secondary flow due to anisotropy.

For instance, in a square sectioned duct, the analysis of the transport equation of the mean vorticity $\Omega_x = \partial W / \partial y - \partial V / \partial z$ shows that the secondary eddies are produce by the term $\frac{\partial^2 (\overline{v^2} - \overline{w^2})}{\partial y \partial z}$. The linear model gives $\Omega_x = 0$.

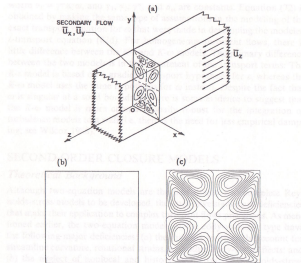


Figure 3 Turbulent secondary flow in a rectangular duct: (a) experiments; (b) standard $K-\epsilon$ model; (c) nonlinear $K-\epsilon$ model of Speziale (1987b).

After Speziale (1991).

↪ In plane strain situation (for instance, in a distorted duct, or close to a stagnation point):

▷ Exact production is in the form $P = (\overline{w^2} - \overline{v^2})D$

▷ the Boussinesq relation leads to $P = 4\nu_t D^2$

▷ For strong strains ($4\nu_t D$ is higher than $\overline{w^2} - \overline{v^2}$):

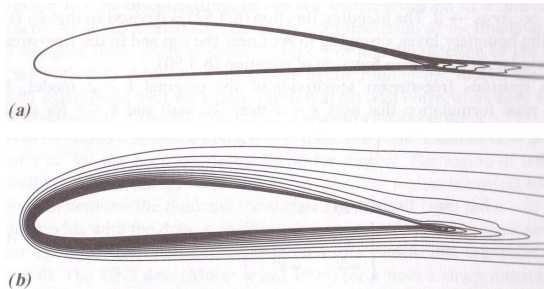
the Boussinesq relation leads to a strong overestimation of the production

⇒ This phenomenon is known as the *stagnation point anomaly*.

▷ This anomaly is critical:

- It leads to a strong overestimation of the turbulent energy in boundary layers downstream of stagnation points

⇒ It significantly delays separation.



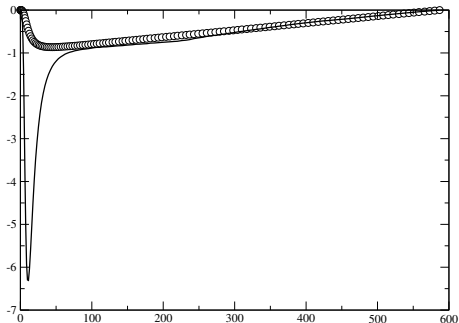
Turbulent energy contours.

(a) Without the stagnation point anomaly. (b) With the stagnation point anomaly.

From Durbin and Pettersson Reif (2001).

↪ Finally, the standard k - ε model is not valid in near-wall regions.

- ▷ The main problem is the estimation of the turbulent viscosity by the Prandtl-Kolmogorov relation $\nu_t = C_\mu k^2 / \varepsilon$:



A priori test: comparison of \overline{uv} and $-2\nu_t S_{12}$. DNS data of Moser *et al.* (1999).

- ▷ A specific treatment in the near-wall region is necessary.

5.5.3. Nonlinear models

Nonlinearities in the constitutive relation

⇒ present pros and cons compared to the standard k - ϵ model.

✓ Advantages:

- ↪ The constitutive relation is more subtle and can potentially reproduce complex flows.
- ↪ This relation can account for the effects the linear relation is not sensitive to: negative production, rotation effects, generation of secondary vortices, *etc.*
- ↪ The stagnation point anomaly can be avoided.
- ↪ Some nonlinear models are also valid in the near-wall region.

✓ Limitations:

~> Numerically less robust than linear models.

~> The calibration process is more difficult:

there are more coefficients \Rightarrow are difficult to isolate considering simple flows.

~> They have a behavior much less predictable than the standard $k-\epsilon$ model:

they can give very good results for some flows, and, for other flows, have a completely wrong behavior.

5.6. Correction for the stagnation point anomaly

Correcting this anomaly is crucial, such that numerous corrections were proposed.

✓ Durbin's realizability constraint

Durbin (1996) remarks that the stagnation point anomaly can be related to a realizability problem.

↪ Eigenaxes of the strain rate S_{ij} used as the coordinate axes.

The tensor S_{ij} is then diagonal:

$$\begin{pmatrix} \lambda_1 & 0 & 0 \\ 0 & \lambda_2 & 0 \\ 0 & 0 & \lambda_3 \end{pmatrix} \quad (\lambda_i\text{'s} = \text{eigenvalues})$$

In this coordinate frame: the three diagonal components of the Reynolds stress (*normal stresses*), given by the Boussinesq relation are

$$\forall \alpha, \quad \overline{u_\alpha^2} = -2C_\mu kT\lambda_\alpha + \frac{2}{3}k$$

where $T = k/\varepsilon$ has been defined.

It can be shown that if $T \leq \frac{1}{C_\mu \sqrt{6} S}$, i.e., if this bound is imposed on T :

$$T = \min \left(\frac{k}{\varepsilon} ; \frac{1}{C_\mu \sqrt{6} S} \right)$$

and the eddy viscosity is evaluated by $\nu_t = C_\mu kT$ instead of $\nu_t = C_\mu k \frac{k}{\varepsilon}$, then the realizability constraint $\overline{u_\alpha^2} \geq 0$ is satisfied.

↪ This upper limit imposed to the turbulent time scale corrects the stagnation point anomaly: indeed, production becomes

$$P = \min \left(2C_\mu \frac{k^2}{\varepsilon} S^2 ; \frac{2}{\sqrt{6}} C_\mu k S \right)$$

The second term is linear in S : the strong overestimation of production in the regions of strong plane strain is thus avoided (see the figure of § 5.5.2).

Imposing a realizability constraint can then have surprising effects!

✓ Shih *et al.*'s correction

- ↪ Shih *et al.* (1995): similar reasoning on the realizability of the Reynolds stress
- ↪ Variable C_μ coefficient, rather than an upper bound for the time scale (looking into details, this is more or less the same).
- ↪ The C_μ coefficient is a bit more complex, but as a summary, in a plane strain situation, it reduces to

$$C_\mu = \frac{1}{A_0 + A_S \frac{k}{\varepsilon} S}$$

- ↪ When the strain is strong, the second term of the denominator is dominant, and production becomes

$$P = \frac{2\sqrt{2}}{3} k S$$

which is in the same form as in the case of Durbin's correction.

- ↪ This model has become very popular since it was implemented in the commercial code Fluent, under the name *Realizable k - ε model*.
- ↪ It is now available in other codes as well.

✓ The RNG $k-\varepsilon$ model

- ~> Yakhot and Orszag, 1986: applying the theory of the renormalization group (from theoretical physics) to turbulence \Rightarrow one can theoretically derive the equations of the $k-\varepsilon$ model.
- ~> Method based on a spectral partitioning of turbulence + simplifying assumptions concerning the interactions between the scale.
- ~> By a recursive procedure, it is possible to tend to the Reynolds decomposition $\Rightarrow k-\varepsilon$ equations.
- ~> Exactly the equations of the standard $k-\varepsilon$ model, but:
 - ▷ the coefficients are obtained directly from the theory.
 - ▷ Unfortunately, it appeared that with that set of coefficients the model does not give very good results in the log layer.

↪ Modification by Yakhot *et al.* (1992):

▷ Different coefficients

▷ Additional term $(-R)$ in the ε equation:

$$R = \frac{C_\mu \eta^3 (1 - \eta/\eta_0)}{1 + \beta \eta^3} \frac{\varepsilon^2}{k}$$

with $\eta = \sqrt{2S} k / \varepsilon$.

↪ This is the so-called *RNG model*, for *Re-Normalization Group*.

↪ Adding that term is strictly equivalent to introducing a variable $C_{\varepsilon 1}$ coefficient in the form:

$$C_{\varepsilon 1}^* = C_{\varepsilon 1} - \frac{\eta(1 - \eta/\eta_0)}{1 + \beta \eta^3}$$

~> The other coefficient are

$$C_\mu = 0.085 \quad ; \quad C_{\varepsilon 1} = 1.42 \quad ; \quad C_{\varepsilon 2} = 1.68 \quad ; \quad \sigma_k = 0.72 \quad ; \quad \sigma_\varepsilon = 0.72$$

$$\eta_0 = 4.38 \quad ; \quad \beta = 0.012$$

~> The model is often proposed in commercial code and is rather popular in the industry, in particular because it corrects the stagnation point anomaly.

~> However, as shown by Laurence (2002), the improvement of the predictions in stagnation regions is due to the variable $C_{\varepsilon 1}$ coefficient, which strongly increases the dissipation, instead of diminishing the production \Rightarrow the model improves the predictions because of a compensation of errors.

6. Reynolds-stress models (second moment closures)

Numerous limitations of the eddy-viscosity models \Rightarrow going to Reynolds-stress models have decisive advantages:

- ✓ Equations for the Reynolds stresses are solved:

- \leadsto the models reproduce the memory effect of turbulence

- \leadsto for instance, in the case of a sudden change in the mean strain, turbulence respond with a time delay.

- ✓ By increasing the modeling order, much is gained in the representation of the physics:

- \leadsto instead of assuming that the Reynolds stresses have a given behavior (determined by the constitutive relation), their transport equations are solved

- \leadsto these equations contain the main physical mechanisms that drive turbulence: production, redistribution, turbulent transport, viscous diffusion, dissipation.

- ✓ In particular, the production terms, that are sufficient to explain many phenomena, do not require modeling!

\Rightarrow Reynolds-stress models represent a major breakthrough compared to eddy-viscosity models.

6.1. Reynolds-stress transport equation

$$\begin{aligned}
 \frac{\partial \overline{u_i u_j}}{\partial t} + \underbrace{U_k \frac{\partial \overline{u_i u_j}}{\partial x_k}}_{\mathcal{C}_{ij}} = & \underbrace{-\overline{u_i u_k} \frac{\partial U_j}{\partial x_k} - \overline{u_j u_k} \frac{\partial U_i}{\partial x_k}}_{P_{ij}} + \underbrace{\frac{1}{\rho} p \left(\frac{\partial u_i}{\partial x_j} + \frac{\partial u_j}{\partial x_i} \right)}_{\phi_{ij}} \\
 & - \underbrace{\frac{1}{\rho} \frac{\partial}{\partial x_k} (\overline{u_i p} \delta_{jk} + \overline{u_j p} \delta_{ik})}_{D_{ij}^p} - \underbrace{2\nu \frac{\partial \overline{u_i}}{\partial x_k} \frac{\partial \overline{u_j}}{\partial x_k}}_{\varepsilon_{ij}} + \underbrace{\nu \frac{\partial^2 \overline{u_i u_j}}{\partial x_k \partial x_k}}_{D_{ij}^\nu} - \underbrace{\frac{\partial \overline{u_i u_j u_k}}{\partial x_k}}_{D_{ij}^T}
 \end{aligned} \tag{17}$$

\mathcal{C}_{ij} Convection (by the mean flow)

P_{ij} Production

✓ ϕ_{ij} Pressure-strain or redistribution term

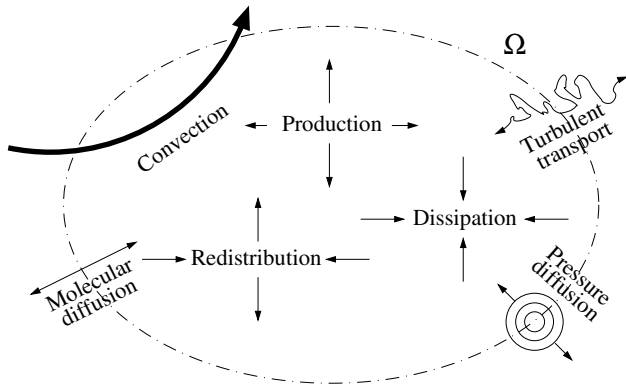
✓ D_{ij}^p Pressure diffusion term

✓ ε_{ij} Dissipation

D_{ij}^ν Molecular or viscous diffusion

✓ D_{ij}^T Turbulent transport or turbulent diffusion

Only the checked terms require modeling.



Warning: this sketch must not hide the fact that:

- ~ production terms are not always positive;
- ~ the different components of production and dissipation are not equal;
- ~ the different transport terms through the control surface can transport energy in both directions, depending on the situation.

- ✓ The terms D_{ij}^T and D_{ij}^p are most of the time called “turbulent diffusion” and “pressure diffusion”, respectively, because they are transport terms leading to effects analog to that of the molecular diffusion D_{ij}^ν .
- ✓ But they actually result from completely different mechanisms:
 - $\leadsto D_{ij}^T$ results from the mean effect of the convection of energy by the fluctuating velocity.
 - $\leadsto D_{ij}^p$ results from the mean effect of the fluctuating pressure forces.
- ✓ The name “diffusion” can hide the fact that these terms have no reason to be purely diffusive, even if, in many cases, it is what is observed.

6.2. The redistribution term

- ✓ Term that plays the most important role after production.
- ✓ Since the production does not require any modeling, it is essentially the redistribution term that characterizes the models.

↪ Chou's theory (1945) shows that there are two mechanisms in the redistribution term:

▷ A slow term, independent of the mean velocity: $\phi_{ij}^1(\mathbf{x}) = B_{ij}(\mathbf{x})$

▷ A rapid term, function of the mean velocity:

$$\phi_{ij}^2(\mathbf{x}) = [A_{ijk l}(\mathbf{x}) + A_{jik l}(\mathbf{x})] \frac{\partial U_l}{\partial x_k}(\mathbf{x})$$

↪ Writing the rapid term as the product of the mean velocity gradient and a tensor only depending on turbulence (not on the mean flow) is based on the so-called *quasi-homogeneity* hypothesis (the mean velocity gradient does not change too much over a distance corresponding to the integral scale of turbulence).

↪ The locality hypothesis consists in assuming that A_{ijml} and B_{ij} can be modeled by an algebraic relation which only depends on *quantities evaluated at point \mathbf{x}* .

▷ According to their exact equations, these terms are actually non-local.

▷ This hypothesis has to be brought into question close to the wall, in particular.

6.2.1. Modeling of the slow term

- ✓ The slow term $\Phi^1 = [\phi_{ij}^1]$, according to its equation, only depends on the state of turbulence
 \Rightarrow the mean field and density have no reason to enter the model.
- ✓ Experiments show that the role of the slow term is essentially to make the turbulence tend to an isotropic state by redistributing the energy between the components
(experiment: homogeneous, initially anisotropic, decaying turbulence)
- ✓ Indeed, this term being traceless, it does not play any role in the budget of turbulent energy k .

✓ We thus seek a model in the form

$$\Phi^1 = f(\mathbf{R}, \beta_i)$$

✓ Similar to the case of eddy-viscosity models, we introduce 2 turbulent scales, ℓ and τ

$$\Phi^1 = f(\mathbf{R}, \ell, \tau)$$

✓ Dimensional analysis:

↪ This relation must be valid whatever the system of units.

↪ Let us chose

$$\ell = \frac{k^{3/2}}{\varepsilon} \quad ; \quad \tau = \frac{k}{\varepsilon}$$

↪ We get

$$\frac{\Phi^1}{\varepsilon} = f\left(\frac{\mathbf{R}}{k}, 1, 1\right)$$

↪ This relation simplifies as

$$\frac{\Phi^1}{\varepsilon} = g\left(\frac{\mathbf{R}}{k}\right)$$

by introducing

$$g(\mathbf{X}) = f(\mathbf{X}, 1, 1)$$

- ✓ The theory of invariants (see for instance Lumley, 1978) shows that the most general relation writes

$$\frac{\Phi^1}{\varepsilon} = \alpha \mathbf{I} + \beta \mathbf{b} + \gamma \mathbf{b}^2$$

where $\mathbf{b} = \frac{\mathbf{R}}{2k} - \frac{1}{3}\delta_{ij}$ is the anisotropy tensor (isotropic turbulence corresponds to $\mathbf{b} = 0$).

- ✓ α , β and γ are functions of the principal invariants of \mathbf{b}

$$I = \{\mathbf{b}\} = 0 \quad ; \quad II = -\frac{\{\mathbf{b}^2\}}{2} \quad ; \quad III = \frac{\{\mathbf{b}^3\}}{3}$$

- ✓ The use of the Cayley-Hamilton theorem appears clearly here: all the powers of \mathbf{b} higher than 2 can be reduced.

✓ trace of Φ^1 is zero $\Rightarrow \{\Phi^1\} = 3\alpha - 2\gamma II = 0$

✓ The most general relation then is

$$\Phi^1 = \beta \varepsilon \mathbf{b} + \gamma \varepsilon \left(\mathbf{b}^2 + \frac{2}{3} II \mathbf{I} \right) \quad (18)$$

✓ It is relatively simple.

✓ But it can still be simplified:

\leadsto Assuming $\gamma = 0$ and $\beta = cste = -C_1 \Rightarrow$ *Rotta's model*.

This is the only *linear model*.

\leadsto Assuming $\gamma = 0$ and a variable β (function of the invariants): so-called
quasi-linear model

(linear tensorial form, but nonlinear coefficients)

\leadsto The nonlinear term can also be used and the coefficient considered constant:
these models are simply called *non-linear models* with constant coefficients.

6.2.2. Modeling of the rapid term

$$\phi_{ij}^2(\mathbf{x}) = [A_{ijml}(\mathbf{x}) + A_{jiml}(\mathbf{x})] \frac{\partial U_l}{\partial x_m}(\mathbf{x}) \quad (19)$$

- ✓ The fourth order tensor $A_{ijml}(\mathbf{x})$ is a function of velocity fluctuations \Rightarrow the Reynolds stress \mathbf{R} and turbulent scales only (e.g., k , ε) enter the model.
- ✓ Φ^2 is linear in the velocity gradient $\nabla \mathbf{U} = \mathbf{S} + \mathbf{W} \Rightarrow$ the model must be a function of \mathbf{R} , k , ε , \mathbf{S} and \mathbf{W} , linear in \mathbf{S} and \mathbf{W} .
- ✓ Theory of invariants (see Speziale *et al.*, 1991) shows that the general form is

$$\begin{aligned} \Phi^2 = & \beta_1 k \mathbf{S} + \beta_2 k \left(\mathbf{bS} + \mathbf{Sb} - \frac{2}{3} \{ \mathbf{bS} \} \mathbf{I} \right) + \beta_3 k \left(\mathbf{b}^2 \mathbf{S} + \mathbf{Sb}^2 - \frac{2}{3} \{ \mathbf{b}^2 \mathbf{S} \} \mathbf{I} \right) \\ & + \beta_4 k (\mathbf{bW} + \mathbf{Wb}) + \beta_5 k (\mathbf{b}^2 \mathbf{W} + \mathbf{Wb}^2) \end{aligned} \quad (20)$$

✓ Model of Speziale, Sarkar and Gatski (SSG)

↪ Speziale, Sarkar and Gatski (1991) studied solutions for short times (rapid distortion theory) and equilibrium solutions for long times ($t \rightarrow \infty$) in homogeneous turbulent flows subjected to plane strains

↪ They showed these solutions can be reproduced using the quasi-linear formulation

$$\Phi^2 = -g_1^* P \mathbf{b} + (g_3 - g_3^* II^{1/2}) k \mathbf{S}$$

$$+ g_4 k \left(\mathbf{bS} + \mathbf{Sb} - \frac{2}{3} \{ \mathbf{bS} \} \mathbf{I} \right) + g_5 k (\mathbf{bW} + \mathbf{Wb})$$

↪ They associate this model with the slow term described in 6.2.2., which is

$$\Phi^1 = -g_1 \varepsilon \mathbf{b} + g_2 \varepsilon \left(\mathbf{b}^2 - \frac{1}{3} \{ \mathbf{b}^2 \} \mathbf{I} \right)$$

↪ Coefficients:

$$g_1 = 3.4 ; g_2 = 4.2 ; g_1^* = 1.8 ; g_3 = 0.8 ; g_3^* = 1.3 ; g_4 = 1.25 ; g_5 = 0.4$$

~> Some comments about the SSG model:

- ▷ It has been applied to numerous case, with some success.
- ▷ In particular, it is well adapted to rotating flows.
- ▷ It is not valid in the near-wall region \Rightarrow it has to be used with wall functions.
- ▷ It is one of the most widely used Reynolds-stress model (with the Rotta+IP model).
- ▷ It is available in some industrial CFD codes.

~> Remark: the nonlinearity of the slow term is often removed by taking $g_2 = 0$.

✓ Rotta+IP or LRR model

↪ Launder, Reece and Rodi (1975): associated the following model for the rapid term

$$\phi_{ij}^2 = -C_2 \left(P_{ij} - \frac{2}{3} P \delta_{ij} \right)$$

to Rotta's slow term

$$\phi_{ij}^1 = -C_1 \varepsilon a_{ij}.$$

↪ P_{ij} is the production tensor and $P = \frac{1}{2} P_{ii}$ is the production rate.

↪ The model is called the *Rotta+IP model* or *LRR model*.

↪ The model for the rapid term ϕ_{ij}^2 (*IP model*, for *Isotropization of Production*):

▷ has a simple phenomenological interpretation.

▷ it can also be seen as a simplification of the SSG model: indeed, taking $g_1 = 3.6$; $g_2 = 0$;

$g_1^* = 0$; $g_3 = 0.8$; $g_3^* = 0$; $g_4 = 1.2$; $g_5 = 1.2$ gives the Rotta+IP model.

▷ Crow's constraint (rapid distortion theory) $\Rightarrow C_2 = \frac{3}{5}$.

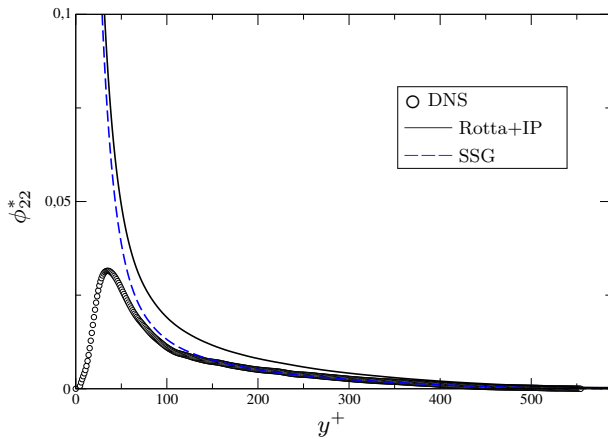
▷ $C_1 = 1.8$ (calibration against the return-to-isotropy experiment).

✓ Other models

- ~> More complex models exist.
- ~> In particular, they are built such a way to satisfy the realizability constraints (the Rotta+IP and SSG models are not realizable).
- ~> It would be too long to present them here. Some examples are the models derived by
 - ▷ Shih and Lumley (1985);
 - ▷ Launder and Tselepidakis (1991);
 - ▷ Craft and Launder (1996), TCL model, for *Two-component limit* (see its slow term in section 6.2.2.).
- ~> These models are very nonlinear.
- ~> They have remained confined in the research field and are not implemented in industrial codes.
- ~> Some of them, in particular the TCL model, have been successfully tested against a wide range of experimental data.
- ~> A very good review of the existing Reynolds-stress models is given by Hanjalić and Launder (2011).

6.2.3. Problems in the near-wall region

- ✓ The SSG and Rotta+IP models are not valid in the near-wall region.
- ✓ The problem can be traced to the hypotheses used in the derivation of the model (locality, quasi-homogeneity) which must be brought in question in the near-wall region.
- ✓ The SSG model gives good results in the log layer, contrary to the Rotta+IP model.
- ✓ Correction terms, the so-called *wall-echo* terms, are generally introduced in CFD codes to correct the Rotta+IP model.



A priori test of the Rotta+IP and SSG models. DNS data from Moser *et al.* (1999).

6.3. Modeling of the turbulent transport terms

- ✓ The two transport terms due to turbulent fluctuations are:

↪ the triple correlation term (third order moments), also called turbulent diffusion

$$D_{ij}^T = -\frac{\partial \overline{u_i u_j u_k}}{\partial x_k}$$

↪ and the pressure diffusion term

$$D_{ij}^p = -\frac{1}{\rho} \frac{\partial}{\partial x_k} (\overline{u_i p} \delta_{jk} + \overline{u_j p} \delta_{ik})$$

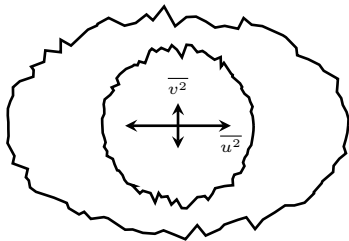
- ✓ Similar to the terms in the k -equation: it is standard to consider that these two terms have similar effects, of diffusive type, and to model them together, despite the remarks made above.
- ✓ The term to be modeled is then: $\overline{u_i u_j u_k} + \frac{1}{\rho} (\overline{u_i p} \delta_{jk} + \overline{u_j p} \delta_{ik})$
- ✓ Historically, it has been considered that pressure diffusion is negligible, and the modeling has been based on measurements of $\overline{u_i u_j u_k}$ and its exact equation.

6.3.1. Shir's model

- ✓ Simplest model (Shir, 1973): considers the triple correlation as a simple diffusion of the Reynolds stresses by a turbulent viscosity (similar to the k or the ε equation), i.e., in using a simple gradient diffusion hypothesis (SGDH):

$$\overline{u_i u_j u_k} = -C_s \frac{k^2}{\varepsilon} \frac{\partial \overline{u_i u_j}}{\partial x_k}$$

- ✓ This model is too isotropic: when turbulence is anisotropic, the turbulent transport is enhanced in the direction in which the fluctuation are stronger.



- ✓ The model is used because it is numerically very stable.

6.3.2. The Daly and Harlow model

- ✓ Daly and Harlow (1970) introduced a generalized gradient diffusion hypothesis (GGDH):

$$\overline{u_i u_j u_k} = -C_s \frac{k}{\varepsilon} \overline{u_k u_l} \frac{\partial \overline{u_i u_j}}{\partial x_l}$$

- ✓ Model is much more popular because it is anisotropic.
- ✓ The coefficient proposed by Daly and Harlow is $C_s = 0.22$, but values between 0.20 and 0.25 are used.

6.4. Dissipation tensor modeling

- ✓ The dissipation of turbulent energy is due to the small scales.
- ✓ At high Reynolds number, the turbulent cascade theory shows that the small scales of the flow are rather isotropic.
- ✓ This remark suggests writing the dissipation tensor in an isotropic form, known as the *Kolmogorov model*:

$$\varepsilon_{ij} = \frac{2}{3}\varepsilon\delta_{ij}$$

where the $2/3$ coefficient is introduced because $\varepsilon = \frac{1}{2}\varepsilon_{ii}$ and $\delta_{ii} = 3$

- ✓ It is only necessary to solve a transport equation for ε .
- ✓ The same equation as the standard k - ε model is used, except as regards the turbulent transport terms, in which a generalized gradient diffusion hypothesis is used:

$$\frac{\partial \varepsilon}{\partial t} + U_k \frac{\partial \varepsilon}{\partial x_k} = C_{\varepsilon 1} \frac{\varepsilon}{k} P - C_{\varepsilon 2} \frac{\varepsilon^2}{k} + \nu \frac{\partial^2 \varepsilon}{\partial x_k \partial x_k} + \frac{\partial}{\partial x_k} \left(C_{\varepsilon} \frac{k}{\varepsilon} \overline{u_k u_l} \frac{\partial \varepsilon}{\partial x_l} \right)$$

- ✓ The isotropy of the small scales is not a valid hypothesis anymore when the turbulent Reynolds number $Re_t = \nu_t/\nu$ is weak, in particular in the near wall region.
- ✓ Remark: similarly to the eddy-viscosity models, a transport equation for another quantity than ε can be used (in particular ω is sometimes used).

6.5. Conclusions about the Reynolds-stress models

- ✓ It is very difficult to give as precise indications as for eddy-viscosity models about the relative performance of the Reynolds-stress models.
- ✓ Indeed, the Reynolds-stress models have been much less used, in particular in industrial application: their implementation in commercial codes is recent and they are not used much.
- ✓ As a summary, only the two models have spread outside of the research domain: Rotta+IP model and SSG model.
- ✓ Both models are based on the Daly and Harlow model for the turbulent transport term and the Kolmogorov model for dissipation \Rightarrow they differ by their redistribution term.
- ✓ For complex flows, they have proved superior to eddy-viscosity models.
- ✓ However, commercial code developers still have to work to make these models as numerically stable as the eddy-viscosity models.

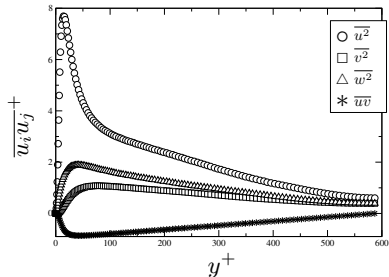
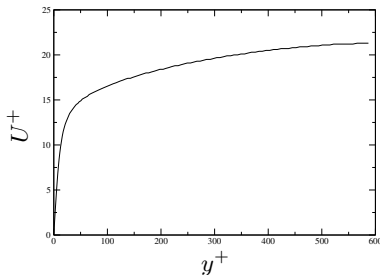
7. The near-wall region

- ✓ We have considered so far that molecular viscosity effects are negligible compared to turbulent effects since $Re_t = \frac{\nu_t}{\nu} \gg 1$
 - ✓ The near-wall region is a region where this is not true anymore
 - ✓ A large majority of the model presented so far is not valid in the near wall region.
 - ✓ In this region, some physical phenomena take place such that the hypotheses upon which the models are based are not valid.
- ⇒ A particular treatment in the near-wall region is necessary.

7.1. Physical phenomena specific to the near-wall region

7.1.1. Viscous effects

- ✓ strong mean velocity gradients (non-slip condition) \Rightarrow a turbulence production peak.
- ✓ a damping of all the components of the fluctuating velocity.
- ✓ a narrowing of the turbulence spectrum (vanishing of the inertial zone).

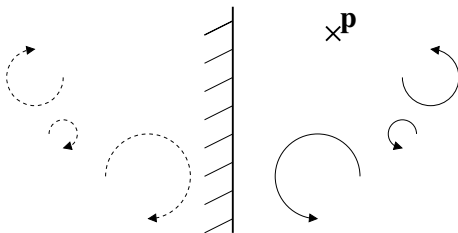


DNS data in channel flow at $Re_\tau = 590$ (Moser *et al.*, 1999)

7.1.2. Non-viscous effects

✓ Wall echo:

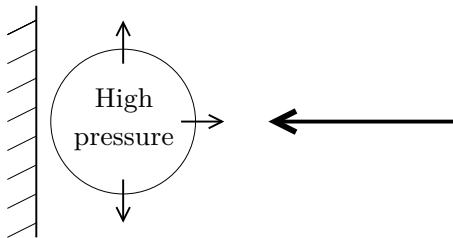
- ~ It can be mathematically shown that the effect of the reflexion of the pressure fluctuation on the wall can be represented considering an image flow.
- ~ The image flow generates pressure fluctuations which are added to the ones generated by the real flow.



✓ Blocking effect:

~> Any fluctuation in the direction of the wall generates a zone of high pressure that tends to slow down the flow in the wall-normal direction and to deviate the velocity towards the directions parallel to the wall.

~> This effect leads to a selective damping of the wall-normal fluctuation.



- ✓ These two non-viscous effects are non-local (the wall is felt at distance by turbulence).
- ✓ All these effects in the vicinity of the wall bring into question:
 - ↪ The quasi-homogeneity: very strong variations of the mean velocity gradient.
 - ↪ The locality: the non-viscous effects are non-local.
 - ↪ The high-Reynolds-number hypothesis: the lack of separation between the large and the small scales implies that viscosity is not negligible.

7.2. Consequences for modeling

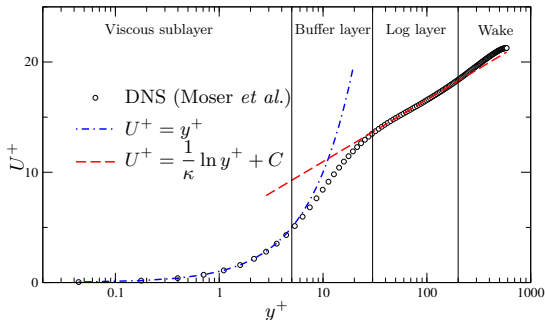
- ✓ Turbulence models are based on hypotheses which are not applicable in the near-wall region.
- ✓ There are several possibilities to circumvent this problem:
 - ~ using wall function which avoid the resolution of the near-wall region;
 - ~ introducing functions depending on the distance to the wall or the turbulent Reynolds number Re_t in order to “force” the model to behave correctly \Rightarrow low-Reynolds number models;
 - ~ prescribing a turbulent scale in the near-wall region \Rightarrow two-layer models;
 - ~ reconsidering the hypotheses used in the derivation of the models, in particular the locality \Rightarrow elliptic relaxation;
 - ~ using models that work in the near-wall region $\Rightarrow k-\omega$ model, TCL model.

7.3. Warning: “high-Reynolds number” and “low-Reynolds number” models

- ✓ It is very important to understand that the notions of “high-Reynolds number” and “low-Reynolds number” do not refer to the *global* Reynolds number of the flow $\left(Re = \frac{U_{\text{ref}} L_{\text{ref}}}{\nu}\right)$ but to the *turbulent* Reynolds number $\left(Re_t = \frac{\nu_t}{\nu}\right)$ which is *local*: it varies inside the flow and, in particular, goes to 0 at the wall.
- ✓ It must be kept in mind that
 - ↪ close to the wall, there is a *low-Reynolds number region*;
 - ↪ there can be low-Reynolds number region elsewhere.
- ✓ What is called a *high-Reynolds number model* is a model which is *not integrable* down to the wall \Rightarrow wall functions must be used.
- ✓ Oftentimes, what is called a *low-Reynolds number model* is simply a model able to reproduce the wall region (i.e., integrable down to the wall).
- ✓ But *low-Reynolds number model* can also specifically refer to models integrable down to the wall using damping functions.

7.4. Wall functions

7.4.1. “Universal” behavior in the near-wall region



✓ In the viscous sub-layer: $\frac{U}{u_\tau} = \frac{y u_\tau}{\nu} \quad (U^+ = y^+)$

✓ In the log layer: $\frac{U}{u_\tau} = \frac{1}{\kappa} \ln \frac{y u_\tau}{\nu} + C \quad (U^+ = \frac{1}{\kappa} \ln y^+ + C)$

$$\varepsilon = \frac{u_\tau^3}{\kappa y} \quad ; \quad \overline{uv} = -u_\tau^2 \quad ; \quad \overline{u_i u_j} = \text{cste}$$

- ✓ These laws have been obtained based on very strict hypotheses:
 - ↪ flow above a plane plate (channel flow or boundary layer);
 - ↪ very high Reynolds number (limit of infinite Re);
 - ↪ no pressure gradient.

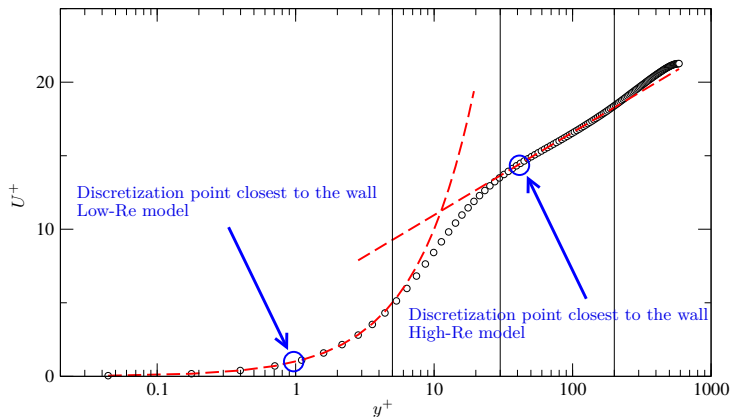
- ✓ These hypotheses are very strict, they actually *never* apply.

- ✓ However, the conclusions are fundamental: the velocity profile, the Reynolds stresses and the dissipation, made non-dimensional by the friction velocity and the viscosity, have a behavior which does not depend on the Reynolds number
 \Rightarrow they are somehow “universal”.

- ✓ In real flows ($Re \neq \infty$, curved walls, pressure gradient, tridimensionality, separation, *etc.*) it has to be considered that
 - ↪ either the “universal” behavior is perturbed
 - ↪ or it is completely modified (at separation points, for instance)

7.4.2. Strategy

- ✓ The universal profile of the mean velocity in the near-wall region shows that, in order to solve the mean velocity down to the wall, the discretization point closest to the wall must be placed at $y^+ \simeq 1$, since $\frac{\partial U^+}{\partial y^+} = 1$.



✓ The condition is *very strict*: for instance, for a flow at a Reynolds number of 10^6 , with a characteristic length scale (geometry) of the order of one meter, the location $y^+ = 1$ corresponds approximately to $y = 20 \mu\text{m}$.

⇒ In the past (70's-80's), such fine meshes were not affordable.

✓ Nowadays, the computing power is sufficient, but in a complex geometry, it is extremely difficult, if not impossible at all, to satisfy the condition $y^+ \simeq 1$ everywhere (at best, generating the mesh is very long).

✓ Wall function strategy:

↪ Any high-Reynolds number model is used (standard $k-\varepsilon$, RNG $k-\varepsilon$, Rotta+IP, SSG, *etc.*).

↪ The discretization point closest to the wall is placed in the log layer.

↪ The exact boundary conditions ($U = 0$, $k = 0$, $\overline{u_i u_j} = 0$, ...) are not used, but rather wall functions, based on the “universal” laws obtained in the log layer.

↪ This artificial boundary conditions correspond to a slip wall with friction.

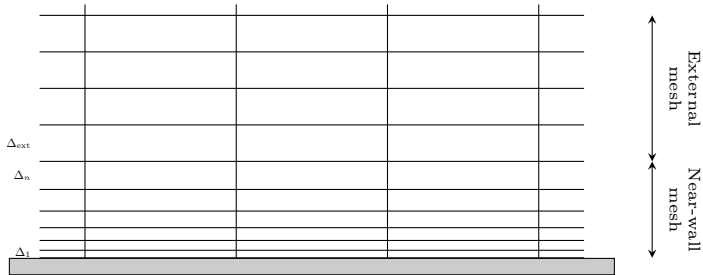
7.4.3. Advantages of the wall functions:

- ✓ They give good results in situations close to ideal conditions (very high Reynolds number, flow parallel to the wall, weak curvature of the wall, *etc.*).
- ✓ They substantially reduce of the number of grid points.
- ✓ They make the mesh generation process easier.
- ✓ They are quite robust: even though their range of applicability is in theory limited, they can be used all the time (this does not guarantee the quality of the results).

✓ It does not mean that the number of cells is reduced by a factor of 30.

~> Indeed, meshes are progressive in the near-wall region: usually, a geometric expansion of the layers of cells is used: $\Delta_i = r\Delta_{i-1}$, where r is the expansion ratio.

~> We want to mesh the near-wall region up to some distance d with a progressive mesh, and then the external region with a constant mesh.



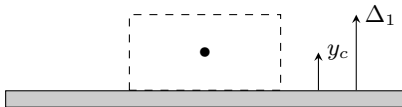
~> We want to choose the two thicknesses Δ_1 and Δ_{ext} . The expansion ratio r and the number of cell layers n are imposed by the two relations

$$\Delta_{\text{ext}} = \Delta_1 r^n \quad \text{and} \quad d = \Delta_1 \frac{r^n - 1}{r - 1}$$

where d is the distance covered by the progressive mesh.

~> Discretization point closest to the wall located at $y_c^+ = 1$ means

$$\Delta_1^+ = \frac{\Delta_1 u_\tau}{\nu} \simeq 2 \quad \text{which implies} \quad \Delta_1 \simeq 2 \frac{\nu}{u_\tau}$$



~> Using the *rule of thumb* $u_\tau \simeq 5\%U_\infty$, we have $\Delta_1 \simeq 2 \frac{\nu}{0.05U_\infty} \simeq \frac{d}{0.025Re}$

↪ We have

$$r = 1 + \frac{\Delta_{\text{ext}}}{d} - \frac{1}{0.025 Re} \xrightarrow{Re \rightarrow \infty} 1 + \frac{\Delta_{\text{ext}}}{d}$$

It can be seen that for large Reynolds numbers, a ratio r of 1.1 means that the thickness Δ_{ext} of the external layers is about 10% of d .

↪ Thus, for large Reynolds numbers,

$$n = \frac{\ln Re}{\ln r} + \frac{\ln(0.025(r-1))}{\ln r}$$

▷ The numbers of layers grows like $\ln Re$.

▷ The size of the cells in the directions parallel to the wall do not depend on the Reynolds number, but only of the geometry.

⇒ In total, the number of cells grows like $\ln Re$, to be compared to Re^3 for DNS!

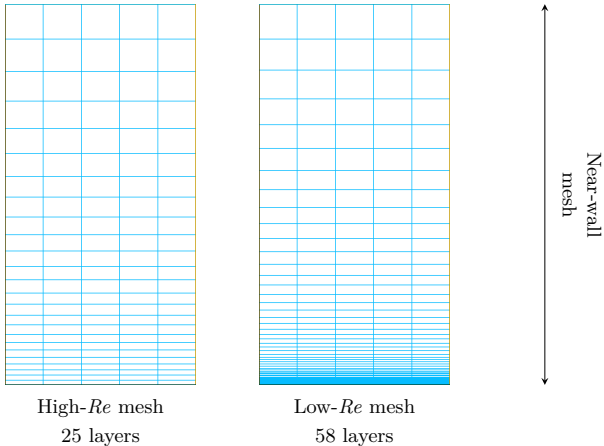
↪ With wall function, the thickness Δ_1 is larger by a factor of 30, such that

$$n = \frac{\ln Re}{\ln r} + \frac{\ln (0.025(r-1)/30)}{\ln r} = \frac{\ln Re}{\ln r} + \frac{\ln (0.025(r-1))}{\ln r} - \frac{\ln 30}{\ln r}$$

↪ Thus, for large Reynolds numbers, there is just a constant additional number of layers $\frac{\ln 30}{\ln r} \simeq \frac{3.4}{r-1} \simeq \frac{3.4}{\Delta_{\text{ext}}/d}$ to switch to the low-Reynolds number strategy.

↪ This number only depends on the ratio $\frac{\Delta_{\text{ext}}}{d}$. For instance, for $\Delta_{\text{ext}} = 10\% d$, there are about 34 additional layers, whatever the Reynolds number.

~ Example: near-wall meshes for $Re = 100\,000$, with $\Delta_{\text{ext}} = 10\%d$



~ For a large-scale industrial case at large Reynolds number, a reduction by a factor of 2 or 3 of the total number of cells can be expected, which is already significant!

7.4.4. Limitations of the wall functions

- ✓ The relations does not hold in theory: at stagnation, separation, reattachment points, in recirculation regions, when the boundary layer is subjected to a pressure gradient, when the wall is curved, when the flow is 3D, *etc.*
 - ✓ The location of the discretization point closest to the wall must be very well controlled: it must be located in the log layer, i.e., between $y^+ = 30$ and about one tenth of the boundary layer thickness
- ⇒ the adequate size of the grid cells can be evaluated a priori, but must always be checked a posteriori (by evaluating u_τ from the results).

- ✓ There are elaborate wall functions which
 - ↪ account for the influence of pressure gradients;
 - ↪ ensure a more or less correct behavior if the discretization point closest to the wall is below $y^+ = 30$ (*adaptive* wall functions)
- ✓ Remark: if this point is below the log layer (buffer layer, viscous sublayer), high-Reynolds number models are not valid \Rightarrow adaptive wall functions are necessarily used with low-Reynolds number models.
 - \Rightarrow Most of the CFD codes now propose such hybrid low- Re /high- Re strategies (adaptive wall functions (AWF), enhanced wall treatment, etc.): boundary conditions are used that blend wall functions and exact boundary conditions:
 - ▷ If the mesh is locally coarse, AWF tend to standard wall functions.
 - ▷ If the mesh is locally fine, AWF tend to exact boundary conditions.
 - ▷ If the mesh is in between, AWF empirically blend these two limiting cases.

7.5. Low-Reynolds number models using damping functions

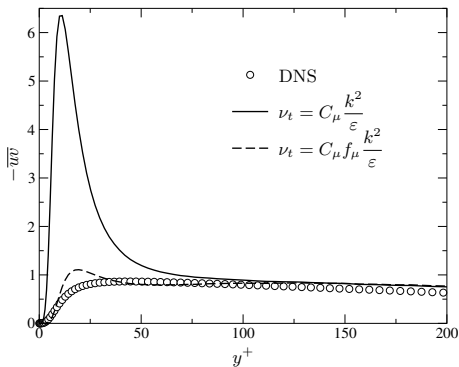
- ✓ In order to avoid the use of wall functions, the models must be integrated down to the wall.
- ✓ For this purpose, their incorrect behavior in the near-wall region must be corrected.
- ✓ The simplest way is to “force” the model to reproduce the data (experimental or from DNS) by introducing *damping functions*.
- ✓ For instance, in order to correct the standard k - ε model, a f_μ function is generally introduced in the turbulent viscosity:

$$\nu_t = C_\mu f_\mu \frac{k^2}{\varepsilon}$$

✓ Lam & Bremhorst (1981), for instance, propose:

$$f_{\mu} = \left[1 - \exp \left(-0,0165 \frac{\sqrt{k}y}{\nu} \right) \right]^2 \left(1 + \frac{20,5}{Re_t} \right)$$

which corrects the prediction of \overline{uv} in a channel flow.



- ✓ It can be seen that this function depends on both the distance to the wall and the turbulent Reynolds number.
- ✓ Some models only depends on one of them:
 - ↪ dependence on $Re_y = \sqrt{k}y/\nu \Rightarrow$ active only in the near-wall region.
 - ↪ dependence on $Re_t \Rightarrow$ active in all the low-Reynolds number regions, including the near-wall region.
- ✓ There is a large number of models of this type.
- ✓ Damping functions are also necessary in the ε equation (in front of of $C_{\varepsilon 1}$ and $C_{\varepsilon 2}$).
- ✓ The models using damping functions are simply (and ambiguously) called **low-Reynolds number models**.

- ✓ This approach avoids the use of wall functions.
 - ✓ It requires a discretization point closest to the wall located at $y^+ \simeq 1$.
 - ✓ The damping functions are nonlinear (exponential) \Rightarrow sometimes, numerical difficulties.
 - ✓ The damping functions are very empirical, chosen in order to match a particular type of flow (boundary layer).
- \Rightarrow lack of universality (but less than wall functions).
- ✓ This approach is also applicable to Reynolds-stress models.

7.6. Two-layer models

- ✓ So-called *two-layer models* divide the flow in 2 regions: the near-wall region and the rest of the domain.
- ✓ In the rest of the domain, the basic equations of the model are solved (for instance, the standard k - ε model).
- ✓ In the near-wall region, a one-equation model is used: the k equation is solved, but ε is obtained from an algebraic relation.
- ✓ Example: the Norris and Reynolds (1975) model.

$$\varepsilon = \frac{k^{3/2}}{\ell} \left(1 + \frac{5.3}{Re_y} \right) \quad \text{with } \ell \text{ given by} \quad \ell = \kappa C_\mu^{-3/4} y$$

- ✓ A damping function is also used for the turbulent viscosity:

$$f_\mu = 1 - \exp \left(-\frac{Re_y}{50.5} \right)$$

- ✓ A test is necessary to determine in which layer the point is: for instance the value $f_\mu = 0.95$ can be chosen as the limit of the application of the 1-equation model \Rightarrow the limit is in the log layer.
- ✓ These models require a discretization point closest to the wall located at $y^+ \simeq 1$.
- ✓ It appears that, in general, they are numerically more stable than low-Reynolds number models (with damping functions).
- ✓ Remark: the 1-equation model used in the near-wall region is not *complete*.
 - \leadsto It is however applicable because it is only used close to the wall where the value of ℓ can be prescribed.
 - \leadsto It is thus assumed that in complex configurations the variation of the length scale ℓ with the distance to the wall is the same as in a simple boundary layer.
- ✓ They are more empirical than low-Reynolds number models (prescription of ℓ).
- ✓ This approach is also applicable to Reynolds-stress models.

7.7. The k - ω SST Model

- ✓ To derive the SST model (Menter, 1994) remarks that:
 - \leadsto the standard k - ε model does not work in the near-wall region;
 - \leadsto the k - ω model works well in the near-wall region, but is sensitive to the external turbulence level;
 - \Rightarrow a “mixed” model would have all the advantages.
- ✓ The idea is then to write a set of equation that tends to the k - ω model in the near-wall region and to the k - ε model far from the wall.
- ✓ The transition between the two models is made by quite complex functions which depend on the distance to the wall \Rightarrow the principle is simple, but the details are complex.
- ✓ The model has been applied successfully to many configurations. This is a popular model in the aeronautics industry.
- ✓ However, this is still a linear eddy-viscosity model \Rightarrow it suffers from the limitations of these models, but it inherits the good properties of the k - ω model in the near-wall region.

- ✓ Using the change of variables $\omega = \epsilon/(C_\mu k)$, the k - ϵ model can be written under the form of a k - ω model:

$$\overline{u_i u_j} = -2\nu_t S_{ij} + \frac{2}{3}k\delta_{ij}$$

$$\frac{\partial k}{\partial t} + U_k \frac{\partial k}{\partial x_k} = P - \beta^* k\omega + \frac{\partial}{\partial x_k} \left[(\nu + \sigma_k \nu_t) \frac{\partial k}{\partial x_k} \right]$$

$$\frac{\partial \omega}{\partial t} + U_k \frac{\partial \omega}{\partial x_k} = \alpha \frac{P}{\nu_t} - \beta \omega^2 + \frac{\partial}{\partial x_k} \left[(\nu + \sigma_\omega \nu_t) \frac{\partial \omega}{\partial x_k} \right] + 2(1 - F_1) \sigma_{\omega 2} \frac{1}{\omega} \frac{\partial k}{\partial x_i} \frac{\partial \omega}{\partial x_i}$$

- ✓ These are not exactly the same equations as for the k - ω model: the coefficients in red are different; there is an additional cross-term $\frac{1}{\omega} \frac{\partial k}{\partial x_i} \frac{\partial \omega}{\partial x_i}$

- ✓ The model is driven by $F_1 = \tanh \left\{ \left\{ \min \left[\max \left(\frac{\sqrt{k}}{\beta^* \omega y}, \frac{500\nu}{y^2 \omega} \right), \frac{4\rho \sigma_{\omega 2} k}{C_{Dk\omega} y^2} \right] \right\}^4 \right\}$

↪ When $F_1 \rightarrow 1$ (near-wall region): the cross-term disappears and k - ω model is recovered.

↪ When $F_1 \rightarrow 0$ (far from the wall): the cross-term is active and the equations are exactly the standard k - ϵ equations with the change of variable $\omega = \epsilon/k$.

✓ Coefficients are a blend of k - ω and k - ε coefficients

$$C = C_1 F_1 + C_2 (1 - F_1)$$

\leadsto k - ω coefficients:

$$\alpha_1 = 5/9 \quad ; \quad \beta_1 = 3/40 \quad ; \quad \sigma_{k1} = 0.85 \quad ; \quad \sigma_{\omega1} = 0.5$$

\leadsto k - ε coefficients:

$$\alpha_2 = 0.44 \quad ; \quad \beta_2 = 0.0828 \quad ; \quad \sigma_{k2} = 1 \quad ; \quad \sigma_{\omega2} = 0.856$$

✓ To avoid an overestimation of the shear stress in adverse pressure gradient boundary layers (very important situation for aeronautics), a bound is introduced in the eddy-viscosity (hence the name SST for *Shear Stress Transport*)

$$\nu_t = \frac{a_1 k}{\max(a_1 \omega, S F_2)} \quad ; \quad S = \sqrt{2 S_{ij} S_{ij}}$$

✓ To avoid the stagnation point anomaly, a bound is introduced in the production

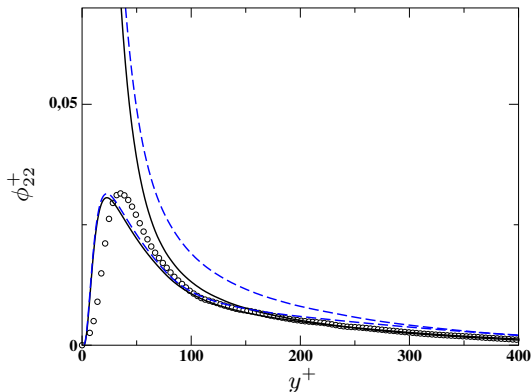
$$P = \min(\nu_t S^2, 10 \beta^* k \omega)$$

7.8. Elliptic relaxation

- ✓ From the observation that most of the problems of Reynolds-stress models in the near-wall region come from the hypotheses of Chou's theory (locality, quasi-homogeneity), Durbin, in 1991, proposed a different theory.
- ✓ This theory shows that the pressure-strain correlation is solution of the equation

$$\phi_{ij} - L^2 \nabla^2 \phi_{ij} = \phi_{ij}^h$$

- ✓ This equation is known as the *elliptic relaxation equation*.
- ✓ It is seen that
 - $\leadsto \phi_{ij}$ is not given by an algebraic relation, but rather by a differential equation (one per component).
 - \leadsto This approach can be used with any quasi-homogeneous model ϕ_{ij}^h (Rotta+IP, SSG, *etc.*).
 - \leadsto The locality and quasi-homogeneity hypotheses are not used, except for the right hand side ϕ_{ij}^h , which is active essentially far from the wall.



Effect of the elliptic relaxation on the models: — — — Rotta+IP ; — SSG. Comparison with DNS data in a channel flow at $Re_\tau = 590$ (Moser *et al.*, 1999).

\Rightarrow By reproducing accurately the redistribution term in the near-wall region, the model reproduces the blocking effect (nonlocal effect) \Rightarrow the two-component limit of turbulence is obtained.

✓ Drawbacks of the method

↪ There are 6 additional equation \Rightarrow this is 13 a equation model!

↪ The model is numerically unstable.

\Rightarrow this model has not spread outside of the research world.

✓ However, simplified models which can be used in industrial situations have been derived:

↪ Reynolds-stress model: Elliptic Blending Reynolds-Stress Model (EBRSM, Manceau and Hanjalić, 2002), using a single additional differential equation (8 equation model) and numerically stable:

$$\alpha - L^2 \nabla^2 \alpha = 1$$

$$\phi_{ij}^* = (1 - \alpha^3) \phi_{ij}^w + \alpha^3 \phi_{ij}^h$$

where ϕ_{ij}^w is an asymptotically correct near-wall model.

↪ This model is available in StarCCM+, in Code_Saturne and will be available soon in OpenFOAM.

→ Eddy-viscosity model: $\overline{v^2}$ - f model, proposed by Durbin in 1991, available in most of the commercial codes.

→ The $\overline{v^2}$ - f model is a drastic simplification of the Durbin's Reynolds-stress model using elliptic relaxation. It consists of replacing the eddy-viscosity

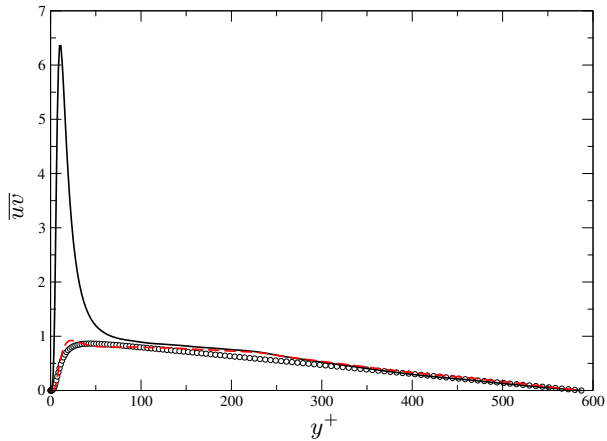
$$\nu_t = C_\mu \frac{k^2}{\varepsilon}$$

by

$$\nu_t = C_\mu \overline{v^2} \frac{k}{\varepsilon} \quad (21)$$

→ The $\overline{v^2}$ scale (the Reynolds-stress component normal to the wall) is actually the correct scale. For instance, writing the Daly and Harlow model in a channel flow shows that the only non-zero component of turbulent diffusion is

$$\frac{\partial \overline{u_i u_j v}}{\partial y} = - \frac{\partial}{\partial y} \left(\underbrace{C_s \overline{v^2} \frac{k}{\varepsilon}}_{\nu_t} \frac{\partial \overline{u_i u_j}}{\partial y} \right)$$



A priori test of \overline{uv} given by:

— the standard eddy-viscosity;

- - - the $\overline{v^2}$ - f eddy-viscosity.

DNS data from Moser *et al.* (1999) : channel flow at $Re_\tau = 590$

- ✓ The equations for $\overline{v^2}$ and its redistribution term f_{22} (simply called f) are identical to those obtained in a channel flow (i.e., only the wall-normal component is considered in the model).
- ✓ These equations, which are valid in a channel flow, are then considered valid in any configuration.
- ✓ This was a daring hypothesis, but it appeared that the model gives good results in a many situations.

$$\begin{aligned}
D_t k &= P - \varepsilon + \nabla \cdot ((\nu + \nu_t) \nabla k) \\
D_t \varepsilon &= \frac{C'_{\varepsilon 1} P - C_{\varepsilon 2} \varepsilon}{T} + \nabla \cdot \left(\left(\nu + \frac{\nu_t}{\sigma_\varepsilon} \right) \nabla \varepsilon \right) \\
D_t \overline{v^2} &= k f - \frac{\overline{v^2}}{k} \varepsilon + \nabla \cdot ((\nu + \nu_t) \nabla \overline{v^2}) \\
f - L^2 \nabla^2 f &= (C_1 - 1) \frac{(2/3 - \overline{v^2}/k)}{T} + C_2 \frac{P}{k} \\
\nu_t &= C_\mu \overline{v^2} T \quad ; \quad P = 2\nu_t S_{ij} S_{ij} \\
L &= C_L \max \left(\frac{k^{3/2}}{\varepsilon} ; C_\eta \left(\frac{\nu^3}{\varepsilon} \right)^{1/4} \right) \quad ; \quad T = \max \left(\frac{k}{\varepsilon} ; 6 \left(\frac{\nu}{\varepsilon} \right)^{1/2} \right)
\end{aligned}$$

8. Conclusion about RANS models

- ✓ RANS modelling is a complex topic, which requires a lot of physical understanding and a bit of maths
- ✓ There is no model that gives good results in all the situations
- ✓ Nor is one model better than the others
- ✓ This is the reason why there are many different models in the codes
- ✓ In each application field (aerodynamics of planes, of cars, turbomachinery, heat transfer, ...), industries choose one model and often, they keep it for a very long time

- ✓ Research on RANS models has been active for 60 years!
- ✓ There has been a big decline in research (not as concerns the use in industry) over the last 20 years: a lot of hope has been placed in unsteady approaches (LES, hybrid RANS/LES)
- ✓ There is a renewal based on new approaches. The *data-driven* approaches (AI, machine learning, data assimilation) make it possible:
 - ~ to refine the models by using variable coefficients
 - ~ to consider a mixture of experts: AI make a choice or blends the models in the different regions of the flow
 - ~ The GEKO model of Ansys Fluent is a kind of mixture of experts but *hand-made* (regions chosen by the user)

✓ Because of the cost and ecological impact of unsteady simulations, RANS will stay for a long time:

~> On its own as long as the results are satisfactory for a given application

~> In combination with LES otherwise (hybrid RANS/LES)

Heat transfer modeling

1. Instantaneous equations

Back to the Navier–Stokes equations, we have

✓ Mass conservation (continuity)

$$\frac{\partial \rho^*}{\partial t} + \frac{\partial \rho^* u_i^*}{\partial x_i} = 0 \quad (22)$$

✓ Momentum conservation

$$\frac{\partial \rho^* u_i^*}{\partial t} + \frac{\partial \rho^* u_i^* u_j^*}{\partial x_j} = -\frac{\partial p^*}{\partial x_i} + \frac{\partial}{\partial x_j} \left[\mu \left(\frac{\partial u_i^*}{\partial x_j} + \frac{\partial u_j^*}{\partial x_i} \right) \right] + \rho^* g_i \quad (23)$$

✓ Energy conservation

$$\rho^* C_v \frac{\partial T^*}{\partial t} + \rho^* C_v \frac{\partial u_i^* T^*}{\partial x_i} = \frac{\partial}{\partial x_i} \left(\lambda \frac{\partial T^*}{\partial x_i} \right) + 2\mu s_{ij}^* s_{ij}^* \quad (24)$$

The terms in red (ρ^* , μ , C_v , λ and μ) are functions of the temperature.

1.1. Usual approximations

✓ Variation of the physical properties :

Approximated dependence laws can be used for the physical properties of the fluid.

- ▷ For instance, Sutherland's law is often used to describe the evolution of the viscosity with temperature:

$$\frac{\mu(T)}{\mu_0} = \left(\frac{T}{T_0}\right)^{3/2} \frac{T_0 + S}{T + S} \quad (25)$$

- ▷ The heat capacity C_p and the Prandtl number Pr are often considered constant, such that

$$\lambda(T) = \frac{\mu(T)C_p}{Pr} = \frac{\mu_0 C_p}{Pr} \left(\frac{T}{T_0}\right)^{3/2} \frac{T_0 + S}{T + S} \quad (26)$$

✓ **Low-Mach-number approximation:**

↪ For low Mach numbers, it can be assumed that density does not depend on pressure, but only on temperature

$$\rho^* = f(\cancel{P^*}, T^*) \quad (27)$$

↪ For a perfect gas, we have

$$p^* = \rho^* r T^* \quad (28)$$

such that

$$d\rho^* = \frac{\partial \rho^*}{\cancel{\partial P^*}} \Big|_{T^*} dP^* + \frac{\partial \rho^*}{\partial T^*} \Big|_{P^*} dT^* = -\frac{\rho^*}{T^*} dT^* \quad (29)$$

and

$$\rho^* = \rho_0^* \frac{T_0^*}{T^*} \quad (30)$$

↪ The flow is thus considered incompressible, but density varies as a function of the inverse of the temperature: the fluid is dilatable.

↪ The equations of motion can be derived using asymptotic expansions at the limit of small Mach numbers.

1.2. Boussinesq approximation

✓ Assuming that the temperature differences are not too large, the **Boussinesq approximation** can be applied:

↪ Density variations can be neglected ($\rho^* = \rho_0$)

↪ Except in the buoyancy term $\rho^* g_i$.

✓ Additional simplifications: in this case, it is standard to also assume that

↪ μ , k and C_v are independent of the temperature;

↪ Density variations in the buoyancy term are linear in the temperature:

$$\rho^* = \rho_0^* - \beta \rho_0^* (T^* - T_0^*) \quad \text{with} \quad \beta = -\frac{1}{\rho_0^*} \frac{\partial \rho^*}{\partial T} \quad (31)$$

such that

$$\rho^* g_i = \rho_0^* g_i - \rho_0^* g_i \beta (T^* - T_0) \quad (32)$$

✓ The following equations are obtained

$$\frac{\partial u_i^*}{\partial x_i} = 0 \quad (33)$$

$$\frac{\partial u_i^*}{\partial t} + \frac{\partial u_i^* u_j^*}{\partial x_j} = -\frac{1}{\rho_0} \frac{\partial p^{**}}{\partial x_i} + \nu \frac{\partial^2 u_i^*}{\partial x_j \partial x_j} - \beta g_i (T^* - T_0) \quad (34)$$

$$\frac{\partial T^*}{\partial t} + \frac{\partial u_i^* T^*}{\partial x_i} = \alpha \frac{\partial^2 T^*}{\partial x_i \partial x_i} \quad (35)$$

where $\alpha = \frac{\lambda}{\rho C_p}$ is the molecular diffusivity.

✓ Remarks:

↪ As usual, the hydrostatic pressure is contained in p^{**} :

$$p^{**} = p^* - \rho_0 g_j x_i \delta_{ij} \quad (36)$$

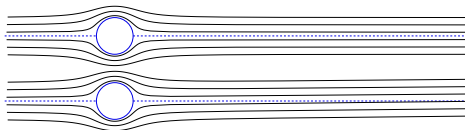
↪ For low Mach number flows, the heat source due to viscous dissipation

$$2\mu s_{ij}^* s_{ij}^*$$

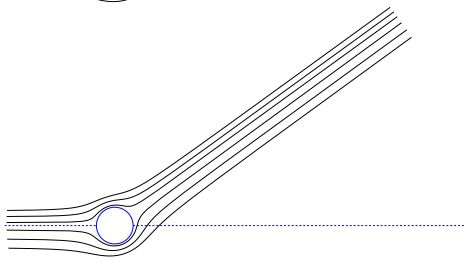
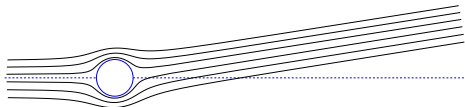
is neglected.

2. The different flow regimes

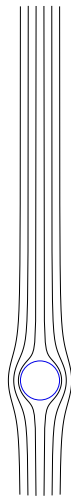
2.1. Example



Forced convection



Mixed convection



Natural convection

2.2. Richardson number

- ✓ Relative weight of the buoyant term compared to the convective term = Richardson number

$$\frac{\partial u_i^*}{\partial t} + \frac{\partial u_i^* u_j^*}{\partial x_j} = -\frac{1}{\rho_0} \frac{\partial p^*}{\partial x_i} + \nu \frac{\partial^2 u_i^*}{\partial x_j \partial x_j} - \beta g_i (T^* - T_0) \quad (37)$$

$$\frac{\partial u_i^* u_j^*}{\partial x_j} \simeq \frac{U_{\text{ref}}^2}{L_{\text{ref}}} \quad ; \quad \beta g_i (T^* - T_0) \simeq \beta g \Delta T \Rightarrow Ri = \frac{\beta g \Delta T L_{\text{ref}}}{U_{\text{ref}}^2}$$

- ✓ It can also be seen as a time-scale ratio

$$\leadsto \text{The time scale of convective phenomena: } \frac{\partial u_i^*}{\partial t} \simeq \frac{\partial u_i^* u_j^*}{\partial x_j} \Rightarrow \tau_{\text{conv}} = \frac{L_{\text{ref}}}{U_{\text{ref}}}$$

\leadsto Time scale of buoyancy phenomena

$$\Rightarrow Ri = \frac{\tau_{\text{conv}}}{\tau_{\text{buo}}} \quad (38)$$

- ✓ When the Richardson number is small compared to unity ($Ri \ll 1$), it is considered that the flow is in the forced convection regime.
- ✓ In this case, heat transfer has no influence on dynamics, the model for the Reynolds stress does not require modifications.

3. Reynolds decomposition

✓ Introducing the Reynolds decomposition, with $T^* = T + \theta$, leads to

$$\frac{\partial U_i}{\partial x_i} = 0 \quad (39)$$

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x_j} = -\frac{1}{\rho} \frac{\partial P}{\partial x_i} + \nu \frac{\partial^2 U_i}{\partial x_j \partial x_j} - \frac{\partial \overline{u_i u_j}}{\partial x_j} - \beta g_i (T - T_0) \quad (40)$$

$$\frac{\partial T}{\partial t} + U_i \frac{\partial T}{\partial x_i} = \alpha \frac{\partial^2 T}{\partial x_i \partial x_i} - \frac{\partial \overline{u_i \theta}}{\partial x_i} \quad (41)$$

✓ Compared to the isothermal case, three additional unknowns are involved, the turbulent heat fluxes $\overline{u_i \theta}$.

- ✓ The turbulent heat fluxes play the same role for the mean temperature as the Reynolds stresses for the mean velocity: they increase the transport of internal energy (rather than momentum for the Reynolds stresses). As for the $\overline{u_i u_j}$, first moment closures or second moment closures can be applied.
- ✓ Heat transfer modeling has received much less attention than Reynolds stress modeling. The main reason is that it is difficult to measure them. They have been investigated mainly since the 80's, using DNS. We will not go as much into details as for the Reynolds stresses. For a complete review, see for instance the article of Hanjalić [16].

4. Forced convection

- ✓ When the Richardson number is small compared to unity ($Ri \ll 1$), it is considered that the flow is in the forced convection regime.
- ✓ In this case, heat transfer has no influence on dynamics, the model for the Reynolds stress does not require modifications.
- ✓ Similar to the case of the Reynolds stresses, the turbulent heat transfer can be modeled using algebraic relations (first moment closure).

4.1. Simple Gradient Diffusion Hypothesis (SGDH)

- ✓ The turbulent heat fluxes are simply written as

$$\overline{u_i \theta} = -\alpha_t \frac{\partial T}{\partial x_i} \quad (42)$$

- ✓ The turbulent diffusivity (or eddy diffusivity) is linked to the turbulent viscosity (or eddy viscosity)

$$\alpha_t = \nu_t / \sigma_t. \quad (43)$$

- ✓ This is analogous to the Boussinesq relation for the Reynolds stresses (except that this is now a vector, rather than a tensor). It corresponds to the Fourier law for molecular diffusion.

- ✓ σ_t is the turbulent Prandtl number. It can be taken as constant (in general, $\sigma_t = 0,9$ or 1) or evaluated using an experimental correlation, as, for instance, the Kays and Crawford correlation

$$\sigma_t = \frac{1}{0,5882 + 0,228(\nu_t/\nu) - 0,0441(\nu_t/\nu)^2[1 - \exp(-5,165/(\nu_t/\nu))]} \quad (44)$$

- ✓ Such a relation assumes that there is a direct link between turbulent diffusivity and turbulent viscosity, i.e., a similarity between the thermal and dynamic length and time scales.
- ✓ Assuming that the turbulent Prandtl number is of the order of 1 means that turbulence transports heat as rapidly as momentum. This assumption is called *the Reynolds analogy*.

4.2. k - ε - $\overline{\theta^2}$ - ε_θ models

- ✓ In order to avoid using this restrictive hypothesis, the thermal-to-mechanical time-scale ratio $R = \tau_\theta/\tau$ can be defined, where $\tau = k/\varepsilon$ and $\tau_\theta = k_\theta/\varepsilon_\theta$.
- ✓ Here, k_θ is, similarly to the dynamic case, half the temperature variance $k_\theta = \overline{\theta^2}/2$
- ✓ ε_θ is its dissipation rate.
- ✓ τ_θ can be directly used to evaluate the turbulent diffusivity as

$$\alpha_t = k\tau_\theta \tag{45}$$

- ✓ However many authors suggest the use of a mixed time scale τ_m , function of both τ_θ and τ .

- ✓ For instance, Nagano, Tagawa, Tsuji (1991) suggest the relation

$$\tau_m = \frac{k}{\varepsilon} \left[(2R)^2 + 3,4(2R)^{1/2} \left(\frac{\nu_t}{\nu} \right)^{-3/4} \right] \quad (46)$$

- ✓ It is seen that to evaluate the thermal-to-mechanical time-scale ratio R , both k_θ and ε_θ are required.
- ✓ Transport equations must be solved for these two variables, which leads to a 4 equation model: $k-\varepsilon-k_\theta-\varepsilon_\theta$, or, equivalently, $k-\varepsilon-\overline{\theta^2}-\varepsilon_{\theta^2}$.

✓ The exact transport equation for $\overline{\theta^2}$ reads

$$\frac{\partial \overline{\theta^2}}{\partial t} + U_k \frac{\partial \overline{\theta^2}}{\partial x_k} = \underbrace{-2\overline{\theta u_k} \frac{\partial T}{\partial x_k}}_{P_\theta} - \underbrace{2\alpha \overline{\frac{\partial \theta}{\partial x_k} \frac{\partial \theta}{\partial x_k}}}_{\varepsilon_\theta} + \underbrace{\frac{\partial}{\partial x_k} \left(\alpha \frac{\partial \overline{\theta^2}}{\partial x_k} \right)}_{D_\theta^\alpha} + \underbrace{\frac{\partial}{\partial x_k} \left(-\overline{u_k \theta^2} \right)}_{D_\theta^t} \quad (47)$$

✓ As usual, a production term P_θ , a dissipation term ε_θ , a molecular diffusion term D_θ^α and a turbulent transport term D_θ^t . are recognized.

✓ For instance, D_θ^t can be modeled using a turbulent diffusivity

$$D_\theta^t = \frac{\partial}{\partial x_k} \left(\alpha_t \frac{\partial \overline{\theta^2}}{\partial x_k} \right) \quad (48)$$

✓ For ε_θ , a transport equation is solved analogous to that used for ε , but the analysis of its exact equation suggest to include both dynamical and thermal production and dissipation terms

$$\frac{D\varepsilon_\theta}{Dt} = C_{P1} \frac{\varepsilon_\theta}{\overline{\theta^2}} P_\theta + C_{P2} \frac{\varepsilon_\theta}{k} P - C_{D1} \frac{\varepsilon_\theta^2}{\overline{\theta^2}} - C_{D2} \frac{\varepsilon_\theta}{k} \varepsilon + \frac{\partial}{\partial x_j} \left[\left(\alpha + \frac{\alpha_t}{\sigma_\theta} \right) \frac{\partial \varepsilon_\theta}{\partial x_j} \right] \quad (49)$$

4.3. Generalized Gradient Diffusion Hypothesis (GGDH)

- ✓ The SGDH model is an isotropic eddy-diffusivity model.
- ✓ Analogous to the Daly–Harlow model for the triple correlations $\overline{u_i u_j u_k}$, an anisotropic formulation can be used,

$$\overline{u_i \theta} = -C_\theta \frac{k}{\varepsilon} \overline{u_i u_j} \frac{\partial T}{\partial x_j} \quad (50)$$

- ✓ In this model, the $\overline{u_i u_j}$'s are required, such that it is typically associated with a Reynolds stress model.
- ✓ It is also possible, as for the SGDH model, to use, rather than the dynamic time scale, k/ε , the thermal time scale τ_θ or a mixed time scale, by solving additional transport equations for $\overline{\theta^2}$ and ε_θ .

4.4. Second moment closure

- ✓ A second moment closure can also be used, solving transport equations for $\overline{u_i \theta}$.
- ✓ These models are detailed below, for the case of the mixed and natural convection regimes.

5. Mixed and natural convection

- ✓ When the Richardson number is not small compared to unity, the influence of buoyancy on the dynamics cannot be neglected
- ✓ When the Richardson number is large compared to unity, the flow is considered in the natural convection regime (the flow is produced by the thermal field)
- ✓ In the case intermediate between forced and natural convection, the regime is called mixed convection

5.1. Influence on the dynamics

- ✓ In order to understand the influence of the thermal field on the dynamics, it is necessary to write the exact transport equations for the Reynolds stresses

$$\frac{D\overline{u_i u_j}}{Dt} = P_{ij} + D_{ij}^\nu + D_{ij}^T + \phi_{ij} + D_{ij}^p - \varepsilon_{ij} + G_{ij} \quad (51)$$

5.1.1. Modification of production

- ✓ In comparison with the isothermal case, a buoyancy term appears

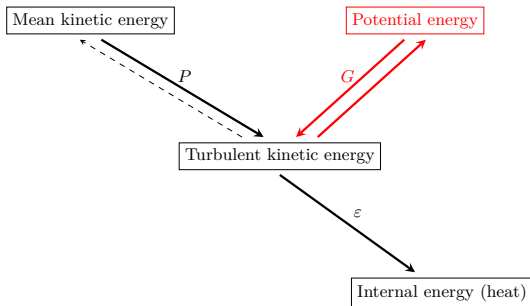
$$G_{ij} = -\beta \left(\overline{u_i \theta} g_j + \overline{u_j \theta} g_i \right) \quad (52)$$

✓ Transport equation for k

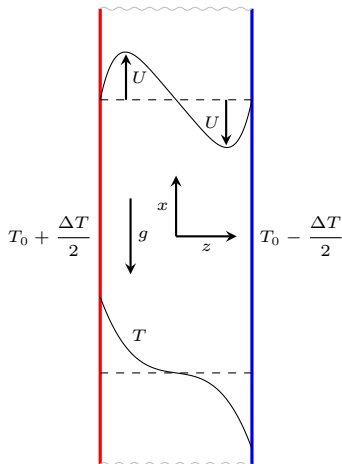
$$\frac{Dk}{Dt} = P + G - \varepsilon + D^\nu + D^T + D^p \quad (53)$$

Buoyancy production: $G = \frac{1}{2}G_{ii} = -\beta g_i \overline{u_i \theta}$

✓ Energy cascade:



✓ Example: differentially heated vertical channel



= natural convection

Flow between two infinite plates, separated by a distance h , generated by the temperature difference between the two walls: $\partial/\partial t = 0$; $\partial/\partial x = 0$; $\partial/\partial y = 0$ $V = 0$; $W = 0$.

- ✓ This case is the most simple case of a flow completely driven by the temperature field.
- ✓ The mean velocity and mean temperature equations reduce to

$$0 = \nu \frac{\partial^2 U}{\partial z^2} - \frac{\partial \overline{uw}}{\partial z} + \beta g(T - T_0) \quad (54)$$

$$0 = \frac{\nu}{Pr} \frac{\partial^2 T}{\partial z^2} - \frac{\partial \overline{w\theta}}{\partial z} \quad (55)$$

✓ It can be seen that

- ↪ As for the case of an isothermal channel flow (generated by a pressure gradient), only the Reynolds shear stress has an influence on the mean velocity (\overline{uv} for a standard channel flow).
- ↪ The buoyancy term plays in this flow the role of the driving force that was played by the pressure gradient in the standard channel flow case.
- ↪ Only the component $\overline{w\theta}$ has an influence on the mean temperature profile.

✓ The production terms of the Reynolds stresses reduce to

Component	Production	Buoyancy production
$\overline{u^2}$	$-2\overline{uw}\frac{\partial U}{\partial z}$	$+2\beta g\overline{u\theta}$
$\overline{v^2}$	0	0
$\overline{w^2}$	0	0
\overline{uw}	$-\overline{w^2}\frac{\partial U}{\partial z}$	$+\beta g\overline{w\theta}$
k	$-\overline{uw}\frac{\partial U}{\partial z}$	$+\beta g\overline{u\theta}$

- ✓ It can be seen that the temperature fluctuations interact with the velocity fluctuations to create (or suppress) turbulence (potential energy \rightleftharpoons turbulent energy).
- ✓ The vertical direction is a privileged direction.
- ✓ But it will be seen that the sign of the turbulent heat fluxes $\overline{u_i \theta}$ strongly depends on the thermal stratification, i.e., on the sign of $\partial T / \partial x$, the vertical mean temperature gradient (zero here)
- ✓ Turbulence generates thermal agitation, which interacts with turbulence, by modifying the distribution of turbulent energy among the components. Such a complex effect typically requires a second moment closure for a correct reproduction.

5.1.2. Influence on redistribution

- ✓ However, buoyancy also has an indirect influence on some terms, which must be taken into account for their modeling

↪ Redistribution ϕ_{ij} is modified: indeed, Chou's analysis is modified by the fact that the transport equation for the instantaneous velocity involves the buoyancy

$$\phi_{ij} = \phi_{ij}^1 + \phi_{ij}^2 + \phi_{ij}^3 \quad (56)$$

through a term denoted by ϕ_{ij}^3

▷ The term is absolutely similar to the rapid term.

▷ Rapid term ϕ_{ij}^2 (reminder)

ϕ_{ij}^2 is a function of the mean velocity gradient and the dynamical turbulence

$$\text{Production: } P_{ij} = -\overline{u_i u_k} \frac{\partial U_j}{\partial x_k} \quad \Rightarrow \quad \text{IP model: } \phi_{ij}^2 = -C_2 \left(P_{ij} - \frac{2}{3} P \delta_{ij} \right)$$

▷ Buoyant term ϕ_{ij}^3 : similar

ϕ_{ij}^3 is a function of $\beta \mathbf{g}$ and the coupling dynamical/thermal turbulence:

$$\text{Production: } G_{ij} = -\beta g_i \overline{u_j \theta} - \beta g_j \overline{u_i \theta} \Rightarrow \text{IP model: } \phi_{ij}^3 = -C_3 \left(G_{ij} - \frac{2}{3} G \delta_{ij} \right)$$

5.1.3. Influence of buoyancy on dissipation

✓ The dissipation term ε_{ij} is also modified.

↪ This can be seen by writing the exact equation for dissipation that also involves buoyancy terms.

↪ Most of the time, the model for ε_{ij} is not modified, but only the transport equation for ε , in which a buoyancy production term appears

$$G_\varepsilon = -2\beta g_i \nu \overline{\frac{\partial u_i}{\partial x_k} \frac{\partial \theta}{\partial x_k}} \quad (57)$$

✓ A buoyant term must be added to the equation solved for ε :

$$\frac{d\varepsilon}{dt} = C_{\varepsilon 1} \frac{\varepsilon}{k} P_k + G_\varepsilon - C_{\varepsilon 2} \frac{\varepsilon^2}{k} + D_\varepsilon \quad (58)$$

✓ There is no consensus in the literature: does buoyancy significantly influence the energy cascade \Rightarrow the dissipation? How?

\leadsto Negligible influence:

$$G_\varepsilon = 0$$

\leadsto Same influence as P_k :

$$G_\varepsilon = C_{\varepsilon 3} \frac{\varepsilon}{k} G_k$$

with $C_{\varepsilon 3} = C_{\varepsilon 1}$ or $C_{\varepsilon 3} < C_{\varepsilon 1}$

\leadsto Unstable stratification: influence; Stable stratification: negligible influence:

$$G_\varepsilon = C_{\varepsilon 3} \frac{\varepsilon}{k} \max(G_k; 0)$$

\leadsto Influence function of the *flux Richardson number*:

$$C_{\varepsilon 1} \frac{\varepsilon}{k} (P_k + G_k) (1 + C_{\varepsilon 3} Ri_f)$$

$$\text{with } Ri_f = -\frac{G_k}{P_k + G_k}$$

5.2. Transport equation for the turbulent heat fluxes

- ✓ Similar to the Reynolds stresses, the exact transport equation for the turbulent heat fluxes can be written

$$\begin{aligned}
 \frac{\partial \overline{u_i \theta}}{\partial t} + U_k \frac{\partial \overline{u_i \theta}}{\partial x_k} &= \underbrace{\frac{\partial}{\partial x_k} (-\overline{u_i u_k \theta})}_{D_{\theta i}^t} + \underbrace{\frac{\partial}{\partial x_k} \left(\frac{1}{\rho} \overline{p \theta} \delta_{ik} \right)}_{D_{\theta i}^p} \\
 &+ \underbrace{\frac{\partial}{\partial x_k} \left(\nu \theta \frac{\partial \overline{u_i}}{\partial x_k} + \alpha \overline{u_i} \frac{\partial \theta}{\partial x_k} \right)}_{D_{\theta i}^\nu} + \underbrace{\frac{1}{\rho} \overline{p} \frac{\partial \theta}{\partial x_i}}_{\phi_{\theta i}} \\
 &\underbrace{-\beta g_i \overline{\theta^2}}_{G_{\theta i}} - \underbrace{\left(\overline{u_i u_k} \frac{\partial T}{\partial x_k} + \overline{u_k \theta} \frac{\partial U_i}{\partial x_k} \right)}_{P_{\theta i}} - \underbrace{(\nu + \alpha) \overline{\frac{\partial u_i}{\partial x_k} \frac{\partial \theta}{\partial x_k}}}_{\varepsilon_{\theta i}} \quad (59)
 \end{aligned}$$

✓ All the terms are already familiar, since they are similar to those appearing in the transport equations for the Reynolds stresses:

\leadsto turbulent transport $D_{\theta i}^t$

\leadsto pressure diffusion $D_{\theta i}^p$

\leadsto molecular diffusion $D_{\theta i}^\nu$

\leadsto pressure-temperature gradient correlation, called *scrambling term* $\phi_{\theta i}$

\leadsto production $P_{\theta i}$ and $G_{\theta i}$

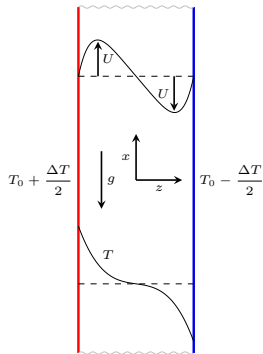
\leadsto dissipation $\varepsilon_{\theta i}$

✓ There are however important differences:

- ↪ Production $P_{\theta i}$ does not arise from the *turbulence/mean velocity gradient* interaction but rather from both the *turbulence/mean temperature gradient* and *turbulent heat fluxes/mean velocity gradient* interactions.
- ↪ The buoyancy production term $G_{\theta i}$ involves the temperature variance $\overline{\theta^2}$.
- ↪ Molecular diffusion must be modeled, except for the case $\nu = \alpha$, i.e., when $Pr = 1$, since in this case $\frac{\partial}{\partial x_k} \left(\nu \frac{\partial \overline{u_i \theta}}{\partial x_k} \right)$.

5.3. Illustration of the role of the production terms

5.3.1. Differentially heated vertical channel



✓ In this case,

the production of the turbulent heat fluxes writes

Comp.	U grad. Prod.	T grad. Prod.	Buoyancy Prod.
$\overline{u\theta}$	$-\overline{w\theta}\frac{\partial U}{\partial z}$	$-\overline{uw}\frac{\partial T}{\partial z}$	$\beta g\overline{\theta^2}$
$\overline{v\theta}$	0	0	0
$\overline{w\theta}$	0	$-\overline{w^2}\frac{\partial T}{\partial z}$	0

✓ It can be seen that

- ↪ The vertical direction is a privileged direction for the production of turbulent heat fluxes.
- ↪ $\overline{w\theta}$, which is the only component that has an influence on the mean temperature (see above), is production by the turbulence/temperature gradient interaction.
- ↪ This remark justifies that gradient models (SGDH and GGDH) can work, despite the fact that they are oversimplified: the flux of energy from the hot wall to the cold wall follows the direction opposed to the temperature gradient and its intensity is driven by the turbulence intensity.
- ↪ There is no production of $\overline{v\theta}$: this component is only fed by the scrambling term.

5.3.2. Unstably stratified channel

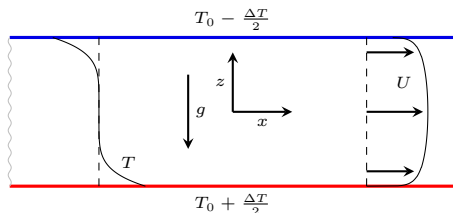
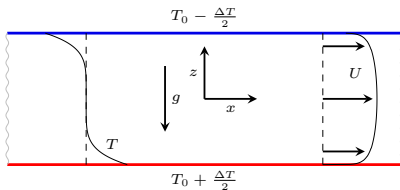


Figure: Flow between two infinite plates, separated by a distance h , generated by a pressure gradient $\partial P/\partial x$, with a temperature difference between the two walls: $\partial/\partial t = 0$; $\partial/\partial x = 0$; $\partial/\partial y = 0$; $V = 0$; $W = 0$.



Comp.	Prod.	Buo. prod.
$\overline{u^2}$	$-2\overline{uw}\frac{\partial U}{\partial z}$	0
$\overline{v^2}$	0	0
$\overline{w^2}$	0	$2\beta g\overline{w\theta}$
k	$-\overline{uw}\frac{\partial U}{\partial z}$	$\beta g\overline{w\theta}$

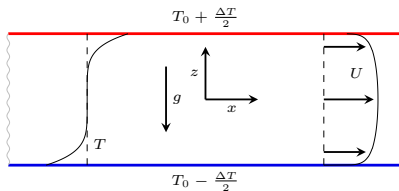
Comp.	U grad. prod.	T grad. prod.	Buo. prod.
$\overline{u\theta}$	$-\overline{w\theta}\frac{\partial U}{\partial z}$	$-\overline{uw}\frac{\partial T}{\partial z}$	0
$\overline{v\theta}$	0	0	0
$\overline{w\theta}$	0	$-\overline{w^2}\frac{\partial T}{\partial z}$	$\beta g\overline{\theta^2}$

✓ The negative temperature gradient produces a positive $\overline{w\theta}$.

✓ This $\overline{w\theta}$ in turn enters the production of $\overline{w^2}$.

⇒ An unstable stratification thus intensifies the vertical fluctuations.

5.3.3. Stably stratified channel



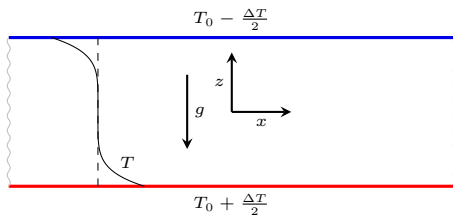
Comp.	U grad. T grad.	Buo. prod.
$\overline{u^2}$	$-2\overline{uw}\frac{\partial U}{\partial z}$	0
$\overline{v^2}$	0	0
$\overline{w^2}$	0	$2\beta g\overline{w\theta}$
k	$-\overline{uw}\frac{\partial U}{\partial z}$	$\beta g\overline{w\theta}$

Comp.	U grad. prod.	T grad. prod.	Buo. prod.
$\overline{u\theta}$	$-\overline{w\theta}\frac{\partial U}{\partial z}$	$-\overline{uw}\frac{\partial T}{\partial z}$	0
$\overline{v\theta}$	0	0	0
$\overline{w\theta}$	0	$-\overline{w^2}\frac{\partial T}{\partial z}$	$\beta g\overline{\theta^2}$

- ✓ A positive temperature gradient produces a negative $\overline{w\theta}$ négatif
 - ✓ that in turn implies a negative production of $\overline{w^2}$.
 - ✓ A stable stratification thus damps the vertical fluctuations.
 - ✓ Can lead to 2-component turbulence (in the atmosphere or the ocean, for instance).
- ⇒ This analysis of a simple case shows that thermal stratification influences the turbulence anisotropy.

5.3.4. Rayleigh-Bénard flow

- ✓ Particular case of the flow without a pressure gradient ($\partial P / \partial x = 0$) with an unstable stratification .



Temperature profile in Rayleigh-Bénard convection. After

Comp.	Prod.	Buo. prod.
$\overline{u^2}$	$-2\overline{uw}\frac{\partial U}{\partial z}$	0
$\overline{v^2}$	0	0
$\overline{w^2}$	0	$2\beta g\overline{w\theta}$
k	$-\overline{uw}\frac{\partial U}{\partial z}$	$\beta g\overline{w\theta}$

Comp.	U grad. prod.	T grad. prod.	Buo. prod.
$\overline{u\theta}$	$-\overline{w\theta}\frac{\partial U}{\partial z}$	$-\overline{uw}\frac{\partial T}{\partial z}$	0
$\overline{v\theta}$	0	0	0
$\overline{w\theta}$	0	$-\overline{w^2}\frac{\partial T}{\partial z}$	$\beta g\overline{\theta^2}$

- ✓ It can be seen (Figure) that the temperature gradient is zero at the center.
- ✓ So, in the central region, $\partial T/\partial z = 0$, and turbulence is produced by the buoyancy term $\beta g\overline{\theta^2}$ only.
- ✓ If $\beta g\overline{\theta^2}$ is neglected (SGDH or GGDH) \Rightarrow laminar (linear) temperature profile.

5.4. Modeling of the unknown terms

5.4.1. Scrambling term

- ✓ Similar to the redistribution term for the dynamics, this term is the most important one in the modeling process.
- ✓ Only the simplest model will be presented below, but more sophisticated models can be used. In particular, the method based on the general formulation of tensorial relations can be applied.
- ✓ An analysis similar to that of Chou for the redistribution term shows that the term $\phi_{\theta i}$ can also be separated in three terms:

↪ The slow term: similar to Rotta's model, Monin (1965) proposed

$$\phi_{\theta i}^1 = -C_{\theta 1} \frac{1}{\tau} \overline{u_i \theta} \quad (60)$$

where τ is the dynamic time-scale k/ε .

↪ But a mixed time-scale can also be used (cf. previous discussion about the SGDH model).

- ✓ Rapid term: it can be analytically shown (using a quasi-homogeneous assumption) that this term can be deduced from the model for ϕ_{ij}^3 . If the model $\phi_{ij}^3 = -C_3 \left(G_{ij} - \frac{2}{3} G \delta_{ij} \right)$ is used, the model of Owen (1973) is obtained for $\phi_{\theta i}^2$

$$\phi_{\theta i}^2 = C_{\theta 2} \overline{u_k \theta} \frac{\partial U_i}{\partial x_k} \quad (61)$$

- ✓ Buoyancy term: as for the dynamic buoyancy term, an isotropization of buoyancy production is used (Owen, 1973):

$$\phi_{\theta i}^3 = C_{\theta 3} \beta g_i \overline{\theta^2} \quad (62)$$

5.4.2. Turbulent transport term

- ✓ The adopted model the model of Daly–Harlow, already used for the dynamics
- ✓ The model represents both the triple correlations and the pressure diffusion by

$$D_{\theta i}^t + D_{\theta i}^p = \frac{\partial}{\partial x_k} \left(C_\theta \frac{1}{\tau} \overline{u_k u_l} \frac{\partial \overline{u_i \theta}}{\partial x_l} \right) \quad (63)$$

5.4.3. Molecular diffusion terms

- ✓ The following model is often used

$$D_{\theta i}^\nu = \frac{\partial}{\partial x_k} \left(\frac{\nu + \alpha}{2} \frac{\partial \overline{\theta u_i}}{\partial x_k} \right) \quad (64)$$

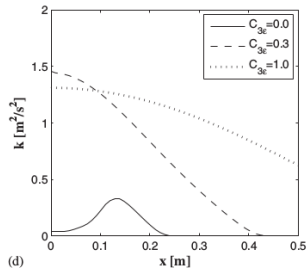
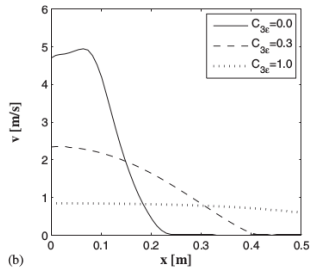
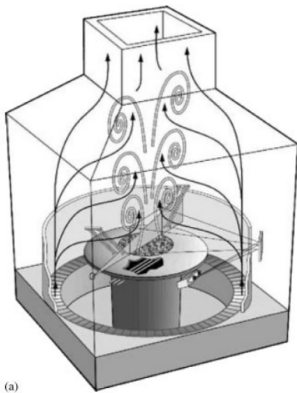
- ✓ It has at least the merit of giving the exact term when $\nu = \alpha$.

5.4.4. Dissipation vector

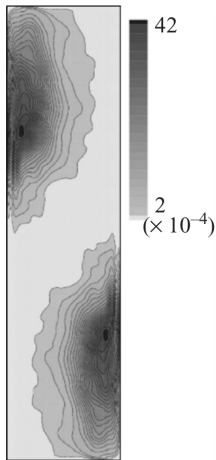
- ✓ For the most common fluids, the Prandtl number is neither very small nor very large compared to unity
- ✓ Most of the time, similar to the case of the Reynolds stress dissipation tensor, ε_{ij} , isotropy is assumed for the dissipation vector of the turbulent heat fluxes $\varepsilon_{\theta i}$
- ✓ This assumption implies $\varepsilon_{\theta i} = 0$ (isotropic vector=0)
- ✓ Warning: for other fluids (liquid metals, for instance), this assumption is not valid.

Example: turbulent buoyant plume of Chung & Devaud, 2008 [5]

$$\text{Here: } G_\varepsilon = C_{\varepsilon 1}(1 - C_{\varepsilon 3})\frac{\varepsilon}{k}G_k$$



5.5. Transition and relaminarization



Turbulent kinetic energy
DNS of a differentially heated
cavity at $Ra = 10^{10}$
From Trias *et al.*, 2007 [44]

- ✓ Buoyancy effects can lead to co-existing laminar and turbulent regions
- ✓ RANS model are not designed to represent such phenomena
- ✓ The location of transition and relaminarization depends on (uncontrolled) modelling subtleties

5.6. Eddy-viscosity models

Reminder:

Boussinesq relation:

SGDH

$$\overline{u_i u_j} = -2\nu_t S_{ij} + \frac{2}{3}k\delta_{ij}$$
$$\overline{u_i \theta} = -\frac{\nu_t}{Pr_t} \frac{\partial T}{\partial x_i}$$

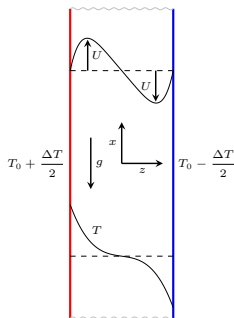
5.6.1. Inconsistency of the model

✓ Production term: $G_k = -\beta g_i \overline{u_i \theta} = \beta g_i \frac{\nu_t}{Pr_t} \frac{\partial T}{\partial x_i}$

✓ Unstratified (or weakly stratified) flows:

$$\leadsto g_i \frac{\partial T}{\partial x_i} = 0 \text{ (orthogonal)} \Rightarrow G_k = 0$$

$\Rightarrow G_\varepsilon$ as well



5.6.2. Buoyancy-extended Eddy-Viscosity Models

Reminder: $\frac{D\overline{u_i u_j}}{Dt} = P_{ij} + G_{ij} + \phi_{ij} - \varepsilon_{ij} + D_{ij}^\nu + D_{ij}^T + D_{ij}^p$

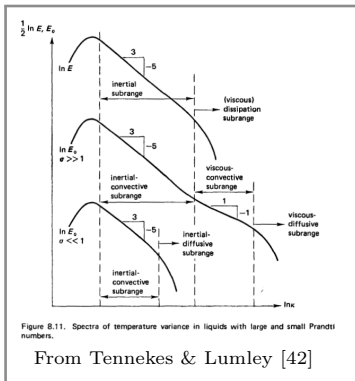
✓ Weak equilibrium + Equilibrium ($P_k + G_k = \varepsilon$):

$$\overline{u_i u_j} = \underbrace{\frac{2}{3}k\delta_{ij} + \frac{k}{\varepsilon} \frac{1 - C_2}{C_1} \left(P_{ij} - \frac{2}{3}P_k\delta_{ij} \right)}_{\text{Boussinesq part}} + \underbrace{\frac{k}{\varepsilon} \frac{1 - C_3}{C_1} \left(G_{ij} - \frac{2}{3}G_k\delta_{ij} \right)}_{\text{Buoyancy extension}} \quad (65)$$

\Rightarrow Buoyancy extended model:

$$\overline{u_i u_j} = \underbrace{\frac{2}{3}k\delta_{ij} - 2\nu_t S_{ij}}_{\overline{u_i u_j}_{\text{Bouss}}} + \underbrace{C_\theta^* \tau \left(G_{ij} - \frac{2}{3}G_k\delta_{ij} \right)}_{\overline{u_i u_j}_{\text{Buo}}} \quad (\text{Davidson, 1990 [9]}) \quad (66)$$

5.6.3. Variable turbulent Prandtl number?



$$\text{SGDH: } \overline{u_i \theta} = -\frac{\nu_t}{Pr_t} \frac{\partial T}{\partial x_i} \text{ with variable } Pr_t ?$$

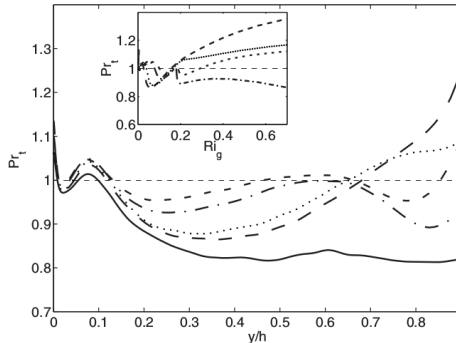
- ✓ Diffusion is due to mixing by large scales
- ✓ These are the same scales for mechanical and thermal turbulence

$$\Rightarrow Pr_t = \frac{\nu_t}{\kappa_t} \text{ must be close to unity}$$

- ✓ Modifying Pr_t for buoyant flows is a common practice (atmosphere/ocean)
- ✓ Why?

$$\triangleright \text{Should have a buoyancy extension: } \overline{v \theta} = -\frac{\nu_t}{Pr_t} \frac{\partial T}{\partial y} + \overline{v \theta}_{\text{Buoy}}$$

$$\triangleright \text{To compensate: } -\frac{\nu_t}{Pr_t^*} \frac{\partial T}{\partial y} = -\frac{\nu_t}{Pr_t} \frac{\partial T}{\partial y} + \overline{v \theta}_{\text{Buoy}} \Rightarrow Pr_t^* = \left(\frac{1}{Pr_t} + \frac{1}{\nu_t} \frac{\overline{v \theta}_{\text{Buoy}}}{\partial T / \partial y} \right)^{-1}$$



Pr_t extracted from a DNS of stably-stratified channel flow

From Garcia-Villalba and del Alamo, 2011 [14]

- ✓ Modifying Pr_t is a patch
- ✓ At least, it should not be a constant but a function of the Richardson number
- ✓ Variations must be modest

5.7. Algebraic modelling

- ✓ Using a certain number of hypotheses, which we will not detail herein, about the equilibrium properties of turbulence, the following algebraic model can be obtained

$$\overline{u_i \theta} = -C_\theta \frac{k}{\varepsilon} \left[\overline{u_i u_k} \frac{\partial T}{\partial x_k} + (1 - C_{\theta 2}) \overline{u_k \theta} \frac{\partial U_i}{\partial x_k} + (1 - C_{\theta 3}) \beta g_i \overline{\theta^2} \right] \quad (67)$$

- ✓ This is the AFM (algebraic flux model) (Hanjalić, Kenjeres, Durst, 1996 [17])

✓ It can be noted that

- ↪ this is an *implicit* algebraic model ($\overline{u_i\theta}$'s appear on both sides of the relation);
- ↪ this model accounts for the influence of the three production terms;
- ↪ the scrambling term is accounted for;
- ↪ if only the first term is considered, the GGDH model is recovered.

5.8. Mixed and natural convection: conclusions

- ✓ Buoyancy strongly couples fluctuating velocity and temperature fields.
- ✓ The vertical direction is obviously a privileged direction.
- ✓ **Potential energy** \leftrightarrow **turbulent energy** transfer depends on stratification (stable/unstable/neutral).
- ✓ Buoyancy production generates anisotropic turbulence (Reynolds stress and turbulent heat flux): damps or promotes vertical fluctuations.
- ✓ Buoyancy also affects redistribution, dissipation, scrambling.

⇒ Second moment closure is the natural level to account for these phenomena:
Reynolds stress model (RSM) + Differential flux model (DSM).

✓ Simplified models are also used:

↪ RSM+Algebraic Flux Model (AFM)

↪ RSM+gradient models (SGDH, GGDH)

↪ Very often: Eddy-viscosity models (EVM), with very limited success

6. Final remarks: the near-wall region

- ✓ The same difficulties as for the modeling of the dynamics are faced for the modeling of the heat transfer.
- ✓ We will not address this issue in this course, however, it is to be noted that, to obtain good predictions in the near-wall region, and in particular the heat transfer between the fluid and the wall, it is necessary (and sometimes sufficient) to reproduce correctly the near-wall dynamics, in particular the two-component limit of turbulence (the wall-normal fluctuations ensure the transport of energy).

- ✓ In particular, the use of wall function is not very relevant. Thermal wall functions must also be written (for the mean temperature, and, possibly, for $\overline{\theta^2}$, ε_θ , $\overline{u_i \theta}$).
- ✓ Moreover, the *universality* of the dynamic wall functions is called into question when the thermal field has an influence. Despite many research efforts, none of the available wall functions are satisfactory in heat transfer.
- ✓ The extension to heat transfer of near-wall models is an active research topic. One can go as far as for the dynamics as concerns the modeling sophistication (TCL model, elliptic relaxation, *etc.*).

Appendix

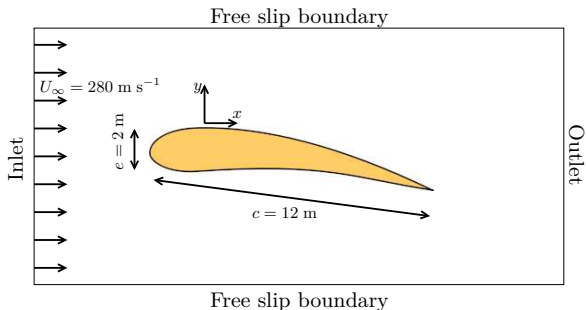
How to perform a good computation

- ✓ Contrary to a widespread belief, solving an engineering problem using a CFD code does not consists in safely clicking some buttons in the interface.
- ✓ Many parameters chosen by the user (model, mesh, boundary conditions, discretization schemes, convergence criteria, etc.) have an influence on the results \Rightarrow results can be very user-dependent if the code is not correctly used.
- ✓ An educated use of CFD is necessary.
- ✓ The turbulence model is a major component, that have been extensively discussed in this course.
- ✓ This appendix provides an overview of the other important parameters and the way to chose them.

✓ We consider the geometrically simple case of an airfoil in a uniform flow at $U_\infty = 280 \text{ m s}^{-1}$.

↪ Turbulent boundary layers develop on both sides.

↪ For the sake of simplicity, it is assumed that the boundary layers are turbulent from the leading edge.



↪ The geometry is assumed invariant by translation in the spanwise (z) direction.

✓ For the code to give the correct results, here is a series of points that deserve attention:

↪ The equations.

↪ The near-wall mesh.

↪ The computational domain.

↪ The boundary conditions.

↪ The numerical schemes.

1. Choice of the equations and of the near-wall mesh

✓ As explained above, it is not possible to perform a DNS of this high-Reynolds number flow. The flow is computed in RANS.

✓ In the present 2D geometry, the mean variables, such as mean pressure, mean velocity, turbulent energy, dissipation, etc., are independent of z and t :

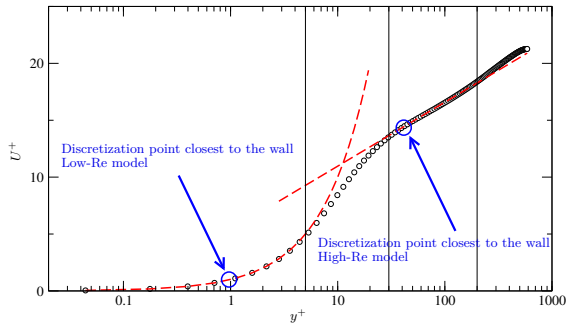
$$\overline{\phi}(x, y, z, t) = \overline{\phi}(x, y)$$

✓ In such a 2D, stationary case:

↪ The mesh can be 2D. In industrial codes, it is usually a 3D mesh with only 1 cell in span with periodic or free-slip boundary conditions (symmetries).

↪ A steady-state algorithm is possible.

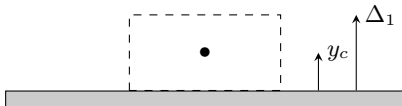
- ✓ For a RANS computation, we have to choose between “low-Reynolds number” and “high-Reynolds number” strategies.
- ✓ As explained above, this choice has nothing to do with the value of the Reynolds number $Re = U_\infty c / \nu$.
- ↪ We are talking about the resolution or not of the near-wall region ($y^+ < 30$).
- ↪ This is the region where the turbulent Reynolds number ν_t / ν is low, i.e., not $\gg 1$.



✓ The two methods have completely different requirements in terms of meshing of the near-wall region:

↪ Low-Reynolds number: fine mesh ($y_c^+ \simeq 1$) where y_c is the distance to the wall of the discretization point closest to the wall (center of the control volume adjacent to the wall in the finite volume method).

↪ High-Reynolds number: coarse mesh ($y_c^+ > 30$).



✓ Since $y_c^+ = \frac{y_c u_\tau}{\nu}$, we need an estimate of u_τ to evaluate the physical distance y_c , in order to decide what is possible or not, and to build the corresponding mesh.

- ✓ An elaborate estimation is possible in the present particular case of a boundary layer, but not in general, complex configurations. Using the rule of thumb $u_\tau = 5\% U_\infty$ is sufficient in general.
- ✓ Here, $u_\tau = 0.05 U_\infty = 14 \text{ m s}^{-1} \Rightarrow y_c = \frac{\nu}{u_\tau} \simeq 1.1 \mu\text{m}$
- ✓ For a low-Reynolds number strategy, the size Δ_1 of the cell adjacent to the wall must be as small as $\Delta_1 = 2y_c \simeq 2 \mu\text{m}$

✓ Using such small cells everywhere in the domain would lead to a huge number of cells.

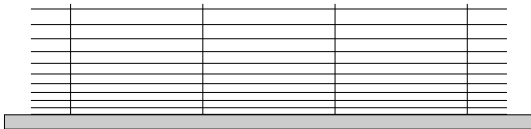
↪ Let us just evaluate the number of cells necessary to cover the boundary layer (the computational domain is much larger).

↪ The boundary layer thickness δ is about 10 cm at the end of the wing (trailing edge).

↪ The number of layers of cells of constant size necessary to cover the boundary layer is $n = \frac{\delta}{\Delta_1} \simeq 50000$

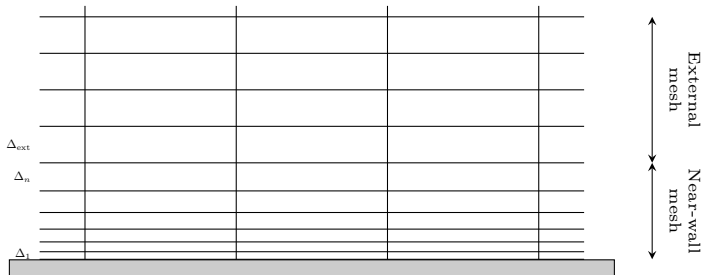
✓ Therefore, a geometric expansion of the grid step is used

$$\Delta_i = r\Delta_{i-1} \quad (68)$$



As already seen in the RANS chapter, section 7.4.3:

- ✓ One wants to match a progressive near-wall mesh ($\Delta_1 \rightarrow \Delta_n$) and the external mesh (Δ_{ext}).



- ✓ If one fixes the two thicknesses Δ_1 and Δ_{ext} , the expansion ratio r and the number of cell layers n are imposed by

$$\Delta_{\text{ext}} = \Delta_1 r^n \quad \text{and} \quad d = \Delta_1 \frac{r^n - 1}{r - 1}$$

where d is the distance covered by the progressive mesh.

- ✓ Thus we have

$$r = 1 + \frac{\Delta_{\text{ext}} - \Delta_1}{d} \quad ; \quad n = \frac{\ln \Delta_{\text{ext}} / \Delta_1}{\ln r} \quad \text{or} \quad n = \frac{\ln \left(1 + (r - 1) \frac{d}{\Delta_1} \right)}{\ln r}$$

- ✓ For large Reynolds numbers, $\Delta_1 \ll \Delta_{\text{ext}}$, such that

$$r \simeq 1 + \frac{\Delta_{\text{ext}}}{d}$$

- ✓ The number of layers necessary to cover the entire boundary layer δ (i.e., we use $d = \delta$ here) can then be easily evaluated:

$$r = 1.01 \quad \rightarrow \quad n = 625$$

$$r = 1.05 \quad \rightarrow \quad n = 160$$

$$r = 1.1 \quad \rightarrow \quad n = 89$$

$$r = 1.2 \quad \rightarrow \quad n = 51$$

- ✓ A reasonable number of layers is obtained using such meshes.

- ✓ Now, for a high-Reynolds number strategy, the size Δ_1 of the cell adjacent to the wall must be larger than $\Delta_1 = 30 \times 2y_c \simeq 60 \mu\text{m}$

\leadsto Layers of constant size necessary to cover the boundary layer:

$$n = \frac{10 \times 10^{-2}}{\Delta_1} \simeq 1666$$

- ✓ Using a geometric expansion of the layers

$$n = \frac{\ln \left(1 + (r - 1) \frac{d}{\Delta_1} \right)}{\ln r}$$

$$r = 1.01 \quad \rightarrow \quad n = 289$$

$$r = 1.05 \quad \rightarrow \quad n = 91$$

$$r = 1.1 \quad \rightarrow \quad n = 54$$

$$r = 1.2 \quad \rightarrow \quad n = 32$$

✓ The number of layers is lower than for the low-Reynolds number strategy, but not much lower.

↪ The factor is only about 2.

↪ This is to cover the boundary layer. For the rest of the domain, the two strategies use the same meshes.

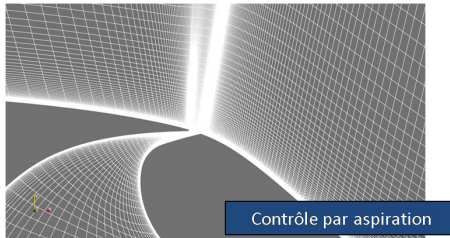
⇒ Using the low-Reynolds number strategy is possible nowadays.

↪ The problem is not the computing power, but rather the difficulty to mesh a complex domain with $y_c^+ \simeq 1$ ensured everywhere.

↪ That is why most of the CFD codes propose so-called adaptive wall functions (AWF) or enhanced wall treatment (see the RANS chapter, section 7.4.4).

✓ Remark:

- ~> the grid step in the directions parallel to the wall is not dependent on the Reynolds number.
- ~> It is to be adapted to the geometry (local curvature of the wall, geometrical details, etc.) and the expected flow structure (sudden variations of the velocity field).



Control by suction

2. Domain

2.1. Height of the domain

- ✓ A free slip condition is generally applied at the top and bottom boundaries: there is no mass flux through these boundaries.

- ✓ This is artificial: the flow domain is, in the real world, unbounded.

- ✓ Mass conservation implies

$$U_b^{\text{inlet}} S = U_b s$$

where S is the inlet section and s the section at some location. U_b is the bulk velocity (velocity averaged over the section).

⇒ there is an acceleration due to the restriction of the section (blockage effect).

- ✓ This artificial acceleration can affect the results.

- ✓ In order to minimize this effect, the acceleration must be as small as possible.
- ✓ The acceleration is linked to the blockage ratio

$$B = \frac{S - s}{S} = 1 - \frac{s}{S}$$

In the present 2D case, the blockage ratio reduces to

$$B = \frac{e}{H}$$

where H is the height of the domain.

- ✓ The acceptable blockage ratio depends on the error we can accept, and the price we are ready to pay (in terms of number of cells).
- ~> The same problem is faced in experiments: wind tunnels are of limited size. In order to work with *not too small* scale models, it is common to accept the compromise $B \simeq 5\%$
- ~> In CFD, it is much easier and cheaper to increase the size of the domain: cells far from the walls are large \Rightarrow we can choose the constraint $B < 1\%$
 - ▷ The minimal value for the domain height H in the present case is thus
$$H > 100e = 200 \text{ m!}$$

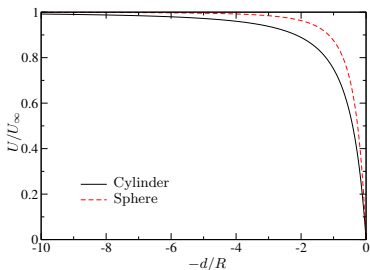
2.2. Location of the inlet boundary

- ✓ A uniform inlet profile $U = U_{\text{inlet}}$ is usually applied at the inlet.
- ✓ Due to the presence of the obstacle, the streamlines are deviated upstream of the leading edge.
- ✓ Imposing a uniform inlet profile thus introduces an error.
- ✓ If the inlet is too close from the leading edge, this error can be significant.

- ✓ It can be shown for an incompressible flow of a perfect fluid that the velocity on the stagnation streamline decreases as a function of the distance d to the stagnation point

$$\frac{U}{U_{\infty}} = 1 - \frac{1}{\left(1 + \frac{d}{R}\right)^2} \quad \text{for a circular cylinder (2D) of radius } R$$

$$\frac{U}{U_{\infty}} = 1 - \frac{1}{\left(1 + \frac{d}{R}\right)^3} \quad \text{for a sphere (3D) of radius } R$$



→ Assuming that we can, for the purpose of this evaluation, approximate the airfoil by a cylinder of diameter e , the minimal distance between the inlet plane and the leading edge such that

$U_\infty - U < 1\%U_\infty$ is

$$\left(1 + \frac{d}{e/2}\right)^{-2} < 0.01 \Rightarrow d > 4.5e$$

→ This is only an evaluation. This must be checked *a posteriori*.

→ It is convenient to check the pressure coefficient given by the computation:

$$C_p = \frac{P - P_\infty}{\frac{1}{2}\rho U_\infty^2}$$

▷ Bernoulli's theorem is roughly applicable upstream of the obstacle (the flow is not turbulent, the dissipation is negligible):

$$P + \rho \frac{U^2}{2} = P_\infty + \rho \frac{U_\infty^2}{2} \quad \Rightarrow \quad C_p = \frac{\rho \frac{U_\infty^2}{2} - \rho \frac{U^2}{2}}{\frac{1}{2}\rho U_\infty^2} = 1 - \frac{U^2}{U_\infty^2}$$

such that C_p varies between 0 (far upstream) and 1 (stagnation point).

▷ $P \neq P_\infty$ at the inlet due to the presence of the obstacle $\Rightarrow C_p \neq 0$.

▷ Since $\delta C_p \simeq -2 \frac{\delta U}{U}$, the value of C_p provides an estimate of the relative error on the velocity.

→ It is wise to recompute the flow with a different inlet location.

2.3. Location of the outlet boundary

- ✓ Outlet boundary condition also impose an artificial constraint of the flow computed.
- ✓ Oftentimes, the outlet conditions are just homogeneous Neuman conditions

$$\frac{\partial \phi}{\partial x_n} = 0$$

- ✓ Here, there is no help from the theory. The only available method is to compute the flow for several locations of the outlet and compare the results.

3. Inlet conditions

- ✓ Boundary conditions for turbulent variable (say k and ε) can have a strong influence.
- ✓ Sometimes, they can be rigorously obtained from an preliminary computation (e.g., fully developed pipe flow), but usually they must be inferred from available information.
- ✓ Experimental data sometimes provide k , but not ε , that cannot be measured.
- ✓ Many codes propose entering the turbulence intensity I and the integral length scale ℓ instead of k and ε , using the relations

$$k = \frac{3}{2}U_{\infty}^2 I^2 \quad \text{and} \quad \varepsilon = C_{\mu}^{3/4} \frac{k^{3/2}}{\ell} \quad (69)$$

- ✓ This is very useful since ℓ is sometimes available or can be inferred from the geometry.

- ✓ For instance, in fully developed flows in ducts (i.e., sufficiently far from the inlet of the duct), the turbulence intensity and integral length scale can be roughly estimated as

$$I \simeq 7\% \quad ; \quad \ell = 5\% D_H$$

where D_H is the hydraulic diameter

$$D_H = 4 \frac{A}{\zeta}$$

i.e., the ratio of the section A of the duct to the perimeter ζ of the duct (so-called wetted perimeter).

- ✓ Many codes automatically prescribe inlet conditions for turbulent variables if the user does not prescribe them.

↪ This is very dangerous! The code basically uses the values for a developed duct flow: they are only valid for duct flows, and only if the correct length scale is chosen.

↪ An educated user (i.e., a user who has read the present chapter) should never let the code choose the inlet boundary.

- ✓ In the present case (airfoil), the incoming flow is laminar
- ✓ Boundary condition $k = 0$ and $\varepsilon = 0$ cannot be used, since they lead to the solution $k = 0$ and $\varepsilon = 0$ everywhere in the domain.
- ✓ The standard method consists in imposing a low turbulence intensity (e.g., $I \simeq 0.1\%$) and a low turbulent Reynolds number (e.g., $R_t \simeq 10$).
- ✓ The length scale is thus obtained using

$$R_t = \frac{\nu_t}{\nu} = \frac{C_\mu}{\nu} \frac{k^2}{\varepsilon}$$

and relations (69), yielding

$$\ell = \frac{\nu R_t}{\sqrt{\frac{3}{2}} C_\mu^{1/4} U_\infty I}$$

4. Initialization

- ✓ In the case of a steady-state computation, the initial conditions are not supposed to have an influence on the results.
- ✓ However:
 - ~> Some initial conditions can lead to a divergence of the computation
 - ~> The number of iterations necessary to reach convergence can significantly depend on the initial conditions
 - ~> Sometimes, there are several solutions to the system of equations. The solution obtained at the end of the computation can depend on the initial conditions [33].
- ✓ The initialization with $U_i = 0$, $k = 0$, $\varepsilon = 0$ is not always the best choice. Some constant, but non-zero values are sometimes preferable (same evaluations as for the boundary conditions).
- ✓ Some codes have automatic initialization procedures.

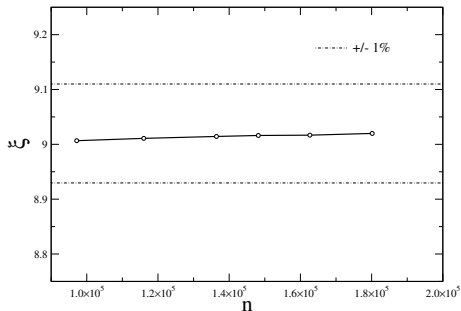
5. Numerics

5.1. Grid refinement studies

- ✓ There is no objective way to directly build a sufficiently fine mesh. The error due to spatial discretization cannot be estimated.
- ✓ The only safe procedure is to perform a grid refinement study: several computations using different meshes.
- ✓ The solutions using different meshes must be compared: as many quantities as possible must be plotted (global quantities, velocity profiles, turbulent energy profiles, streamlines, etc.).
- ✓ The *quantities of interest* must be plotted as a function of the number of cells n : for instance, if the main quantity of interest is C_x (drag coefficient), the curve $C_x = f(n)$ must be plotted (with at least 3 points).

✓ Example: flow through a diaphragm

↪ Here, the quantity of interest is the head loss coefficient $\xi = \frac{\Delta P}{\frac{1}{2}\rho U^2}$



↪ If our criterion is to keep below 1% the error on ξ due to discretization errors, it can be seen on the plot that the mesh is sufficiently fine.

✓ How to refine the mesh

↪ After a first computation, we want to refine the grid to investigate grid convergence. How can we proceed?

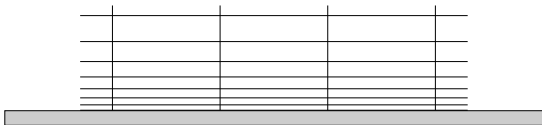
↪ The basic method consists in increasing the number of cells in each direction. However:

▷ For Low-Reynolds number models: we have chosen the thickness of the wall-adjacent layer such that $y_c^+ \simeq 1$; it is useless to refine the mesh, except for some models that can require slightly finer meshes.

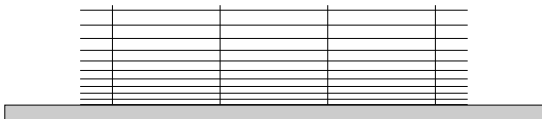
▷ For High-Reynolds number models: the wall-adjacent layer *cannot* be refined, since $y_c^+ > 30$ is necessary.

⇒ In the wall-normal direction, it is better to keep the thickness Δ_1 of the first layer constant, and to reduce r .

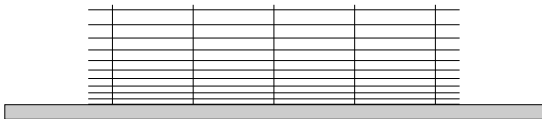
▷ Initial mesh



▷ Refinement in the wall-normal direction without modifying the thickness of the first layer ($r \searrow$)



▷ Refinement in the streamwise direction



5.2. Convergence error

- ✓ The resolution uses iterative methods (iterative steady-state algorithm or time-marching scheme).
- ✓ Computation stops when the relative residuals are below a prescribed value.
- ✓ Be careful! Default values are often too large ($\simeq 10^{-4}$). This is a major source of numerical error.
- ✓ Actually it is better to wait for the residual to reach some plateau: very often, residuals cannot be further reduced.
- ✓ Monitoring the evolution of quantities of interest (e.g., the drag or the lift for an airfoil) is necessary to avoid unpleasant surprises.

5.3. Convection scheme

- ✓ The most critical term in the equations is the nonlinear term (convection).
- ✓ Codes usually propose to choose the discretization scheme for this term among:
 - ↪ The basic upwind differencing (UD) scheme: first-order accurate.
 - ↪ A variety of second-order or third-order accurate schemes.
- ✓ The UD scheme is less accurate, but much more stable (thanks to numerical diffusion), in particular at the beginning of the computation (artificial initial conditions):
 - ↪ It is usual to start the computation with UD until convergence and restart with a more accurate scheme.
 - ↪ Sometimes, it is impossible to obtain a steady-state solution with second/third order accurate schemes (oscillations).

- ✓ Contrary to a common belief, it is fully acceptable to work with the UD scheme:
 - ↪ The low order of accuracy just means that, *for a given mesh*, the solution is less accurate.
 - ↪ A finer mesh must be used with UD to reduce the numerical error to the same level as with higher-order schemes.
 - ↪ It means that the CPU time will be larger to obtain the same level of numerical error.

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